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## INTRODUCTORY NOTE

This manual contains in consolidated form (1) Civil Air Regulations Part 4a, Airplane Airworthiness, dated April 7, 1950, and Amendment 4a-1; and (2) the rules, policies, and interpretations issued by the Administrator of the Federal Aviation Agency in application to the various sections of the regulations.

FAA *rules* are supplementary regulations issued pursuant to authority expressly conferred on the Administrator in the Civil Air Regulations. Such rules are mandatory and must be complied with.

FAA *policies* provide detailed technical information on recommended methods of complying with the Civil Air Regulations. Such policies are for the guidance of the public and are not mandatory in nature.

FAA *interpretations* define or explain words and phrases of the Civil Air Regulations. Such interpretations are for the guidance of the public and will be followed by the Agency in determining compliance with the regulations.

This manual is arranged to give the number, title, and text of each section of the regulations followed by any rules, policies, or interpretations applicable to that section. These rules, policies, or interpretations of the Administrator are identified by consecutive dash numbers appended to the regulation section number.

As amendments and other pertinent materials pertaining to Part 4a are issued, they will be included in this manual.

# Contents

## Subpart A—Airworthiness Requirements

### General

	Section	Page
Scope.....	4a.1.....	1
Airplane categories.....	4a.2.....	1

### Airworthiness and Type Certificates

Requirements for issuance.....	4a.15.....	1
Data required for airworthiness certificate.....	4a.16.....	1
Data required for type certificate.....	4a.17.....	1
Inspection and tests.....	4a.18.....	2
Flight tests.....	4a.19.....	2
Procedure for type certification.....	4a.20.....	2

### Changes

Continued compliance.....	4a.25.....	2
Minor changes.....	4a.26.....	2
Major changes.....	4a.27.....	2
Changes required by the Administrator.....	4a.28.....	2

### Approval of Materials, Parts, Processes, and Appliances

Specifications.....	4a.31.....	2
Approval of aircraft component ( <i>FAA rules which apply to sec. 4a.31</i> ).....	4a.31-1.....	3

## Subpart B—Definitions

Weights.....	4a.37.....	3
Structural terms.....	4a.38.....	3
Air density, <i>p</i> .....	4a.39.....	3
Speed.....	4a.40.....	3
Design gust velocity, <i>U</i> .....	4a.41.....	4
Dynamic pressure, <i>q</i> .....	4a.42.....	4
Load factors.....	4a.43.....	4
Aerodynamic coefficients, <i>C<sub>L</sub></i> , <i>C<sub>M</sub></i> , <i>CP</i> , etc.....	4a.44.....	4
Standard atmosphere (standard air).....	4a.45.....	4
Primary structure.....	4a.46.....	4

## Subpart C—Structural Loading Conditions

### General Structural Requirements

Strength.....	4a.61.....	5
Deformations.....	4a.62.....	5
Stiffness.....	4a.63.....	5
Proof of strength and rigidity.....	4a.64.....	5
Materials, fabrication, protection, etc.....	4a.65.....	5

## Flight Loads

	Section	Page
General.....	4a.72.....	5

### Air Speeds

Air speeds.....	4a.73.....	5
-----------------	------------	---

### Load Factors

General.....	4a.74.....	5
Maneuvering load factors.....	4a.75.....	5
Gust load factors.....	4a.76.....	6
Factors of safety.....	4a.77.....	6

### Symmetrical Flight Conditions (Flaps Retracted)

General.....	4a.78.....	6
Condition I (positive high angle of attack).....	4a.79.....	6
Condition I <sub>1</sub> (positive high angle of attack modified).....	4a.80.....	6
Condition II (negative high angle of attack).....	4a.81.....	6
Condition III.....	4a.82.....	6
Condition IV (negative low angle of attack).....	4a.83.....	7
Condition V (inverted flight).....	4a.84.....	7
Condition VI (gliding).....	4a.85.....	7

### Symmetrical Flight Conditions (Flaps or Auxiliary Devices in Operation)

General.....	4a.86.....	7
Condition VII (positive gust, flaps deflected).....	4a.87.....	7
Condition VIII (negative gust, flaps deflected).....	4a.88.....	7
Condition IX (dive, flaps deflected).....	4a.89.....	7

### Unsymmetrical Flight Conditions

General.....	4a.90.....	8
Condition I <sub>u</sub> .....	4a.91.....	8
Condition III <sub>u</sub> .....	4a.92.....	8
Condition V <sub>u</sub> .....	4a.93.....	8

### Special Flight Conditions

Gust at reduced weight.....	4a.94.....	8
Lift-wire-cut.....	4a.95.....	8
Drag-wire-cut.....	4a.96.....	8
Unsymmetrical propeller thrust.....	4a.97.....	8
Wing tanks empty.....	4a.98.....	8

### Wing Load Distribution

Wing load distribution.....	4a.99.....	8
-----------------------------	------------	---

### Control Surface Loads

General.....	4a.115.....	8
--------------	-------------	---

### Horizontal Tail Surfaces

Balancing.....	4a.116.....	9
Maneuvering (horizontal surfaces).....	4a.117.....	9
Damping (horizontal surfaces).....	4a.118.....	9
Tab effects (horizontal surfaces).....	4a.119.....	9

## Vertical Tail Surfaces

	Section	Page
Maneuvering.....	4a.120	9
Damping (vertical surfaces).....	4a.121	9
Gusts (vertical surfaces).....	4a.122	9
Tab effects (vertical surfaces).....	4a.123	10
Special cases (vertical surfaces).....	4a.124	10

## Ailerons

Maneuvering.....	4a.125	10
Tab effects (ailerons).....	4a.126	10
Flying conditions (ailerons).....	4a.127	10

## Wing Flaps and Tabs

Wing flaps.....	4a.128	10
Tabs.....	4a.129	10

## Special Devices

Special devices.....	4a.130	11
----------------------	--------	----

## Control System Loads

General.....	4a.137	11
Control wires or push rods.....	4a.138	11
Elevator systems.....	4a.139	11
Rudder systems.....	4a.140	11
Aileron systems.....	4a.141	11
Flap and tab control systems.....	4a.142	11

## Ground Loads

General.....	4a.147	11
Level landing.....	4a.148	12
Three-point landing.....	4a.151	12
Energy absorption.....	4a.152	12
Side load.....	4a.153	12
One-wheel landing.....	4a.154	12
Braked landing.....	4a.155	13
Side loads on tail wheel or skid.....	4a.156	13

## Water Loads

General.....	4a.161	13
--------------	--------	----

## Float Seaplanes

Landing with inclined reactions (float seaplanes).....	4a.162	13
Float attachment members.....	4a.163	13
Landing with vertical reactions (float seaplanes).....	4a.164	13
Safety factors.....	4a.165	13
Landing with side load (float seaplanes).....	4a.166	13

## Boat Seaplanes

Local bottom pressures.....	4a.167	14
Distributed bottom pressures.....	4a.168	14
Step loading condition.....	4a.169	14
Bow loading condition.....	4a.170	14
Stern loading condition.....	4a.171	15
Side loading condition.....	4a.172	15

### Seaplane Float Loads

	Section	Page
Seaplane float loads.....	4a.173	15
Seaplane float bottom loads.....	4a.174	15

### Wing-Tip Float Loads

Wing-tip float loads.....	4a.175	16
Wing structure.....	4a.176	16

### Miscellaneous Water Loads

Sea wing loads.....	4a.177	16
---------------------	--------	----

### Special Loading Conditions

Engine torque.....	4a.187	16
High angle of attack and torque.....	4a.188	16
Engine mounts, nacelles, etc.....	4a.189	16
Side load on engine mount.....	4a.190	16
Up load on engine mount.....	4a.191	16
Passenger loads.....	4a.192	16
Safety belt loads.....	4a.193	17
Local loads.....	4a.194	17
Rigging loads.....	4a.195	17
Air loads on struts.....	4a.196	17

### Multiplying Factors of Safety

General.....	4a.207	17
Fittings.....	4a.208	17
Castings.....	4a.209	17
Parallel double wires.....	4a.210	17
Wires at small angles.....	4a.211	17
Double drag trusses.....	4a.212	18
Torque tubes used as hinges.....	4a.213	18
Control surface hinges and control system joints.....	4a.214	18
Wire sizes.....	4a.215	18
Wing lift truss system.....	4a.216	18

### Subpart D—Proof of Structure

General.....	4a.227	18
Proof of structural analysis.....	4a.228	18
Combined structural analysis and tests.....	4a.229	18
Load tests.....	4a.230	19
Flight load tests.....	4a.231	19
Load tests required.....	4a.232	19

### Wings

Proof of wings.....	4a.237	19
Redundancies.....	4a.238	19

### Beams

Beams.....	4a.239	19
Secondary bending.....	4a.240	19
Lateral buckling.....	4a.241	19
Webs.....	4a.242	19
Axial load.....	4a.243	19
Joint slippage in wood beams.....	4a.244	19
Bolt holes.....	4a.245	20
Box beams.....	4a.246	20

**Drag Trusses**

	Section	Page
Drag trusses.....	4a.247	20

**Ribs**

Ribs.....	4a.248	20
Load distribution.....	4a.249	20
Ailerons and high-lift devices.....	4a.250	20
Rib tests.....	4a.251	20

**Covering**

Covering.....	4a.252	20
---------------	--------	----

**Nonparallel Wires**

Nonparallel wires.....	4a.253	20
------------------------	--------	----

**Tail and Control Surfaces**

Proof of tail and control surfaces.....	4a.263	20
Vibration tests.....	4a.264	21

**Control Systems**

Proof of control systems.....	4a.269	21
Control system tests.....	4a.270	21
Operation test.....	4a.271	21
Control system joints.....	4a.272	21

**Landing Gear**

Proof of landing gear.....	4a.277	21
Energy absorption tests.....	4a.278	21

**Hulls and Floats**

Proof of hulls and floats.....	4a.283	21
--------------------------------	--------	----

**Fuselages and Engine Mounts**

Proof of fuselages and engine mounts.....	4a.289	21
Critical column loads.....	4a.290	22
Baggage compartments.....	4a.291	22

**Fittings and Parts**

Proof of fittings and parts.....	4a.297	22
Fittings and attaching members.....	4a.298	22
Bolts.....	4a.299	22

**SUBPART E—Detail Design and Construction**

General.....	4a.301	22
--------------	--------	----

**Materials, Workmanship, and Fabrication Methods**

Materials and workmanship.....	4a.302	22
Fabrication methods.....	4a.303	22
Gluing.....	4a.304	22
Torch welding.....	4a.305	22
Electric welding.....	4a.306	22
Brazing and soldering.....	4a.307	22
Protection.....	4a.308	23
Inspection.....	4a.309	23

### Joints, Fittings, and Connecting Parts

	Section	Page
Joints, fittings, and connecting parts.....	4a.312.....	23
Bolts, pins, and screws.....	4a.313.....	23
Wood screws.....	4a.314.....	23
Eyebolts.....	4a.315.....	23
Castings.....	4a.316.....	23

### Tie-Rods and Wires

Tie-rods and wires.....	4a.319.....	23
Wire terminals.....	4a.320.....	23
Wire anchorages.....	4a.321.....	23
Counter wire sizes.....	4a.322.....	23

### Flutter Prevention

General flutter prevention measures.....	4a.326.....	24
--	-------------	----

### Detail Design of Wings

External bracing.....	4a.329.....	24
Wire-braced monoplanes.....	4a.330.....	24
Lift trusses.....	4a.331.....	24
Jury struts.....	4a.332.....	24
Wing beams.....	4a.333.....	24
Wing beam joints.....	4a.334.....	24
Drag truss.....	4a.335.....	24
Aileron and flap attachments.....	4a.336.....	24
Internally braced biplanes.....	4a.337.....	24
Fabric covering.....	4a.338.....	24
Metal-covered wings.....	4a.339.....	24

### Detail Design of Tail and Control Surfaces

Installation.....	4a.445.....	24
Stops.....	4a.446.....	24
Elevator trailing edge tab systems.....	4a.447.....	25
Hinges.....	4a.448.....	25
Elevators.....	4a.449.....	25
Dynamic and static balance.....	4a.450.....	25
Wing flaps.....	4a.451.....	25
Tabs.....	4a.452.....	25

### Detail Design of Control Systems

Installation.....	4a.459.....	25
Stops.....	4a.460.....	25
Joints.....	4a.461.....	25
Welds.....	4a.462.....	25
Flap controls.....	4a.463.....	25
Flap controls.....	4a.464-T.....	25
Tab controls.....	4a.465.....	26
Spring devices.....	4a.466.....	26
Single-cable controls.....	4a.467.....	26
Control system locks.....	4a.468.....	26
Trim controls.....	4a.469-T.....	26



## Detail Design of Landing Gear

	Section	Page
Shock absorption.....	4a.475	26
Shock-absorbing systems.....	4a.476	26
Wheels.....	4a.477	26
Main landing gear wheels.....	4a.478	27
Tail and nose wheels.....	4a.479	27
Tires.....	4a.480	27
Tire markings.....	4a.481	27
Retracting mechanism.....	4a.482	27
Brakes.....	4a.483-T	27

### Hulls and Floats

Hulls and floats.....	4a.488	27
Buoyancy (main seaplane floats).....	4a.489	27
Buoyancy (boat seaplanes).....	4a.490	27
Water stability.....	4a.491	28
Float design.....	4a.492	28

### Fuselage and Cabins

Provision for turn-over.....	4a.497	28
External door.....	4a.498	28
Location of passenger door.....	4a.499	28
Exits.....	4a.500	28

### Pilot Compartment

Construction.....	4a.501	28
Location.....	4a.502	29
Operation information and limitations.....	4a.504	29
Windows and windshields.....	4a.505	29
Leakage.....	4a.506	29
Seats.....	4a.507	29
Navigation instruments.....	4a.508	29
Opening between pilot compartment and passengers' cabin.....	4a.509	29

### Passenger and Baggage Compartments

Passenger compartments.....	4a.510	29
Passenger chairs.....	4a.511	29
Baggage compartments.....	4a.512	29

### Reinforcement Near Propellers

Reinforcement near propellers.....	4a.513	30
------------------------------------	--------	----

## Subpart F—Equipment

General.....	4a.523	30
Requirements.....	4a.524	30
Approval of aircraft components ( <i>FAA rules which apply to sec. 4a.524</i> ).....	4a.524-1	30
Life preserver of flotation device.....	4a.525	30
Fire extinguishing apparatus.....	4a.526	30

### Non-Air Carrier (NAC) Airplanes

Non-air carrier (NAC) airplanes.....	4a.531	30
NAC landplanes; visual contact day flying (within 100 miles of a fixed base).....	4a.532	30
NAC landplanes; visual contact day flying (unlimited distance).....	4a.533	31
NAC landplanes; visual contact night flying.....	4a.534	31
NAC landplanes; instrument day flying.....	4a.535	31
NAC landplanes; instrument night flying.....	4a.536	32
NAC seaplanes and amphibians.....	4a.537	32

**Air-Carrier Airplanes; Passenger (ACP)**

	Section	Page
Air carrier airplanes; passengers (ACP).....	4a.547	32
ACP landplanes; visual contact day flying.....	4a.548	32
ACP landplanes; visual contact night flying.....	4a.549	32
ACP landplanes; instrument day flying.....	4a.550	33
ACP landplanes; instrument night flying.....	4a.551	33
ACP seaplanes and amphibians.....	4a.552	33

**Installation Requirements**

Installation requirements.....	4a.557	33
--------------------------------	--------	----

**Instrument Installation**

Instruments.....	4a.558	33
Air-speed indicator.....	4a.559	33
Power plant instruments and controls.....	4a.560	33
Fuel quantity gauge.....	4a.561	33
Magnetic compass.....	4a.562	33
Navigational instruments.....	4a.563	33
Gyroscopic instruments.....	4a.564	34

**Safety Equipment Installation**

Safety belts.....	4a.565	34
Fire extinguishers.....	4a.566	34
Safety belt signal.....	4a.567	34
Landing flares.....	4a.568	34
Deicers.....	4a.569	34

**Electrical Equipment Installation**

General.....	4a.570	34
Battery.....	4a.571	34
Fuses.....	4a.572	34
Generator.....	4a.573	34
Running load.....	4a.574	35
Anchor lights.....	4a.575	35
Landing lights.....	4a.576	35
Instrument lights.....	4a.577	35
Position lights.....	4a.578	35
Master switch.....	4a.579	35

**Miscellaneous Equipment Installation**

Seats.....	4a.580	35
Accessories.....	4a.581	35

**Subpart G—Power Plant Installation****Engines**

Engines.....	4a.591	35
--------------	--------	----

**Propellers**

Propellers.....	4a.597	36
Controllable pitch.....	4a.598	36
Propeller clearance.....	4a.599	36

## Fuel Systems

	Section	Page
Capacity and feed.....	4a.605	36
Tank installation.....	4a.606	36
Tank construction.....	4a.607	36
Tank strength.....	4a.608	37
Gauge.....	4a.609	37
Lines and fittings.....	4a.610	37
Strainers.....	4a.611	37
Valves.....	4a.612	37
Dump valves.....	4a.613	37
Drains.....	4a.614	37

### Miscellaneous Fuel System Requirements

Filler openings.....	4a.615	37
Prevention of ice formation.....	4a.616	37

## Lubrication Systems

General.....	4a.621	38
Tank installation.....	4a.622	38
Tank strength.....	4a.623	38
Gauge.....	4a.624	38
Piping.....	4a.625	38
Drains.....	4a.626	38
Oil temperature.....	4a.627	38
Filler openings.....	4a.628	38

## Cooling Systems

General.....	4a.633	38
Radiators.....	4a.634	38
Piping.....	4a.635	38
Drains.....	4a.636	38
Filler openings.....	4a.637	38

## Power Plant Instruments, Controls, Accessories

Instruments.....	4a.642	38
Controls.....	4a.643	39
Throttle controls.....	4a.644	39
Ignition switches.....	4a.645	39
Propeller pitch controls.....	4a.646	39
Accessories (air carrier planes).....	4a.647	39

## Manifolding, Cowling, and Fire Wall

General.....	4a.651	39
Manifolds.....	4a.652	39
Air intakes.....	4a.653	39
Engine cowling.....	4a.654	39
Fire wall.....	4a.655	39
Heating systems.....	4a.656	40

## Miscellaneous Powerplant Requirements

Materials.....	4a.661	40
----------------	--------	----

## Subpart H—Performance

	Section	Page
Performance requirements.....	4a.671.....	40
Landing speeds.....	4a.672.....	40
Take-off.....	4a.673.....	40
Climb.....	4a.674.....	40
Controllability and maneuverability.....	4a.675.....	40
Controllability at the stall.....	4a.676.....	40
Balance.....	4a.677.....	41
Stability.....	4a.678.....	41
Spinning.....	4a.679.....	41
Flutter and vibration.....	4a.680.....	41
Flutter and vibration.....	4a.681-T.....	41
Ground and water characteristics.....	4a.682.....	41

### Modified Performance Requirements

Modified performance requirements for multiengine airplanes not certified in the transport category.....	4a.687.....	42
--	-------------	----

### Performance Tests

General.....	4a.701.....	42
Flight test pilot.....	4a.702.....	42
Parachutes.....	4a.703.....	42
Reports.....	4a.704.....	42
Loading conditions.....	4a.705.....	43
Use of ballast.....	4a.706.....	43
Fuel to be carried.....	4a.707.....	43
Maximum air speed.....	4a.708.....	43
One-engine-inoperative performance.....	4a.709.....	43
Air-speed indicator calibration.....	4a.710.....	43
Check of fuel system.....	4a.711.....	43

### Air Carrier Aircraft Performance Characteristics

Performance characteristics of air carrier aircraft.....	4a.717.....	43
--	-------------	----

### Operation Limitations

Weight.....	4a.723.....	44
New production aircraft—empty weight and c.g. determination ( <i>FAA policies which apply to sec. 4a.723</i> ).....	4a.723-1.....	44
Provisional weight (air carrier airplanes).....	4a.724.....	44
Center of gravity limitations.....	4a.725.....	44
New production aircraft—empty weight and c.g. determination ( <i>FAA policies which apply to sec. 4a.725</i> ).....	4a.725-1.....	44
Air-speed limitations.....	4a.726.....	44
Power plant limitations.....	4a.727.....	45

### Transport Category Airplane Performance Requirements

Performance requirements for transport category airplanes.....	4a.737-T.....	45
Minimum requirements for certification.....	4a.738-T.....	45

### Definitions

Stalling speeds.....	4a.739-T.....	45
Flap positions.....	4a.740-T.....	45
Maximum one-engine-inoperative operating altitude.....	4a.741-T.....	45

**Weights**

	Section	Page
Weights.....	4a.742-T	46
Fuel dumping provisions.....	4a.743-T	46

**Performance Requirements and Determinations**

Required performance and performance determinations.....	4a.744-T	46
Stalling speed requirements.....	4a.745-T	46
Climb requirements.....	4a.746-T	46
Takeoff determination.....	4a.747-T	47
Speeds.....	4a.748-T	47
Take-off path.....	4a.749-T	47
Temperature accountability.....	4a.749a-T	48
Landing determination.....	4a.750-T	48

**Flight Characteristics**

Flight characteristics.....	4a.751-T	48
Controllability and maneuverability.....	4a.752-T	49
Trim.....	4a.753-T	49
Stability.....	4a.754-T	49
Static longitudinal stability.....	4a.755-T	49
Specific stability conditions.....	4a.756-T	50
Dynamic longitudinal stability.....	4a.757-T	50
Directional and lateral static stability.....	4a.758-T	50
Stalling.....	4a.759-T	50

**Operating Manual**

Airplane operating manual.....	4a.760-T	51
Airplane flight manuals ( <i>F A A policies which apply to sec. 4a.760-T</i> ).....	4a.760-T-1	51

**SUBPART I—Miscellaneous Requirements**

Identification plate.....	4a.770	51
Standard weights.....	4a.771	51
Leveling means.....	4a.772	51

**APPENDIXES**

Appendix A—Tables and figures.....	53
Appendix B—Special Civil Air Regulations which affect Part 4a.....	61
SR-399B. Provisional Maximum Certificated Weights for Certain Airplanes Operated by Alaskan Air Carriers, Alaskan Air Taxi Operators, and the Department of the Interior.....	63
SR-407. Basis for Approval of Modification of Airplane Types Douglas DC-3 and Lockheed L-18.....	65
SR-411A. Trial Operation of Transport Category Airplanes in Cargo Service at Increased Zero Fuel and Landing Weights.....	71
SR-426. Performance Credit for Transport Category Airplanes Equipped with Stand-by Power.....	75

# Airplane Airworthiness

## Subpart A—Airworthiness Requirements

*Note:* Current application for airworthiness and type certificates are not processed under this part, but are processed under other airworthiness parts. At present this part primarily governs modification of aircraft which were originally certificated under Part 4a. Unless otherwise provided, all references to other parts of this subchapter are those provisions in effect on September 29, 1947.

### General

4a.1 *Scope.* The airworthiness requirements set forth in this part shall be used as a basis for obtaining airworthiness or type certificates: *Provided, That:* (a) Deviations from the requirements of this part which, in the opinion of the Administrator, insure the equivalent condition for safe operation and, (b) equivalent requirements of the United States Army or Navy with respect to airworthiness may be accepted in lieu of the requirements set forth in this part. Unless otherwise specified an amendment to this part will apply only to airplanes for which applications for type certificates are received subsequent to the effective date of such amendment.

#### 4a.2 *Airplane categories.*

(a) At the election of the applicant, an airplane may be certificated under the requirements for a particular category according to the intended use of the airplane. Sections of this part which affect only one particular category are designated by a suffix added to the appropriate section numbers, as follows:

Normal category.....	Suffix N
Transport category.....	Suffix T
Acrobatic category.....	Suffix A

(b) All sections not designated by a suffix are applicable to all categories, except as otherwise specified.

### Airworthiness and Type Certificates

4a.15 *Requirements for issuance.* The airworthiness requirements specified in this part shall be used as a basis for the certification of airplanes: *Provided, That* an airplane manufactured in accordance with, and conforming to, the currently effective aircraft specifications issued therefor will be eligible for an airworthiness certificate, if the Administrator determines such airplane is in condition for safe operation: *Provided, further, That* an airplane which has not demonstrated compliance with the airworthiness requirements specified in this part but which, in the opinion of the Administrator, is in condition for safe operation for experimental purposes or for particular activities will be eligible for an airworthiness certificate.

4a.16 *Data required for airworthiness certificate.* When an airworthiness certificate is sought and a type certificate is not involved, data which are adequate to establish compliance of the aircraft with the requirements listed in this part shall be submitted to the Administrator.

4a.17 *Data required for type certificate.* Data which are adequate to establish compliance of the aircraft with the airworthiness requirements listed in this part and which are adequate for the reproduction of other airplanes of the same type shall be submitted to the Administrator. The procedure for submitting the required data, the technical contents of such data, and the methods of testing aircraft with respect to the prescribed airworthiness requirements shall be in accordance with Civil Aeronautics Manual 4, Airplane Airworthiness.

**4a.18 Inspection and tests.** Authorized representatives of the Administrator shall have access to the airplane and may witness or conduct such inspections and tests as are deemed necessary by the Administrator.

**4a.19 Flight tests.** (Applicable to all airplanes certificated as a type on or after May 15, 1947.) After proof of compliance with the structural requirements contained in this part, and upon completion of all necessary inspection and testing on the ground, and proof of the conformity of the airplane with the type design, and upon receipt from the applicant of a report of flight tests conducted by him, there shall be conducted such official flight tests as the Administrator finds necessary to determine compliance with Subparts C-G. After the conclusion of these flight tests such additional flight tests shall be conducted as the Administrator finds necessary to ascertain whether there is reasonable assurance that the airplane, its components, and equipment are reliable and function properly. The extent of such additional flight tests shall depend upon the complexity of the airplane, the number and nature of new design features, and the record of previous tests and experience for the particular airplane model, its components, and equipment. If practicable, the flight tests performed for the purpose of ascertaining the reliability and proper functioning shall be conducted on the same airplane which was used in flight tests to show compliance with Subparts C-G.

**4a.20 Procedure for type certification.** Acceptable procedures for type certification are outlined in Civil Aeronautics Manual 4.

### Changes

**4a.25 Continued compliance.** Changes to certificated aircraft shall be substantiated to demonstrate continued compliance of the aircraft with the pertinent airworthiness requirements.

**4a.26 Minor changes.** Minor changes to airplanes being manufactured under the terms of a type certificate and which obviously do not impair the condition of the airplane for safe operation may be approved by

authorized representatives of the Administrator prior to submittal to the Administrator of any required revised drawings. The approval of such minor changes shall be based on the airworthiness requirements in effect when the particular airplane model was originally certificated, unless, in the opinion of the Administrator, compliance with current airworthiness requirements is necessary.

**4a.27 Major changes.** Major changes to airplanes being manufactured under the terms of a type certificate may require the issuance of a new type certificate and the Administrator may, in his discretion, require such changes to comply with current airworthiness requirements.

**4a.28 Changes required by the Administrator.**

(a) In the case of aircraft models approved under the airworthiness requirements in effect prior to the currently effective regulations, the Administrator may require that aircraft submitted for original airworthiness certification comply with such portions of the currently effective regulations as are considered necessary.

(b) All aircraft certificated under the transport category, the manufacture of which is completed after September 30, 1947, shall comply with the following sections of part 4b of this subchapter, as amended: sections 4b.58, 4b.442, 4b.445(a), 4b.447, 4b.478, 4b.848, 4b.503(c), 4b.516-4b.518, 4b.556, 4b.557, 4b.560, 4b.561, 4b.586, 4b.621-4b.624, 4b.651-4b.655 and 4b.661-4b.676.

### Approval of Materials, Parts, Processes, and Appliances

**4a.31 Specifications.**

(a) Materials, parts, processes, and appliances shall be approved upon a basis and in a manner found necessary by the Administrator to implement the pertinent provisions of the Civil Air Regulations. The Administrator may adopt and publish such specifications as he finds necessary to administer this section, and shall incorporate

therein such portions of the aviation industry, Federal, and military specifications respecting such materials, parts, processes, and appliances as he finds appropriate.

(b) Any material, part, process, or appliance shall be deemed to have met the requirements for approval when it meets the pertinent specifications adopted by the Administrator, and the manufacturer so cer-

tifies in a manner prescribed by the Administrator.

4a.31-1 *Approval of aircraft components (FAA rules which apply to sec. 4a.31)*. Aircraft components, made the subject of Technical Standard Orders, shall be approved upon the basis and in the manner provided in Part 514 of this title.

[Supp. 9, 16 F.R. 671, Jan. 25, 1951]

## Subpart B—Definitions

### 4a.37 *Weights.*

(a) *Weight, W.* The total weight of the airplane and its contents.

(b) *Designed weight.* The weight of the airplane assumed for purposes of showing compliance with the structural requirements specified in this part.

(c) *Minimum design weight.* Weight empty with standard equipment, plus crew, plus fuel of 0.25 pound per maximum (except take-off) horsepower, plus oil as per capacity.

(d) *Standard weight.* The maximum weight for which the airplane is certificated as complying with all the airworthiness requirements for normal operations.

(e) *Provisional weight.* The maximum weight for which the airplane is certificated as complying with the airworthiness requirements as modified for scheduled air carriers in section 4a.687.

### 4a.38 *Structural terms.*

(a) *Design wing area, S.* The area enclosed by the projection of the wing outline, including ailerons and flaps but ignoring fairings and fillets, on a surface containing the wing chords. The outline is assumed to extend through nacelles and through the fuselage to the plane of symmetry.

(b) *Design power, P.* The total engine horsepower chosen for use in determining the maneuvering load factors. The corresponding engine output will be incorporated in the aircraft certificate as a maximum operational limitation in all flight operations other than take-off or climbing flight (see sec. 4a.727).

(c) *Design wing loading, W/S.* The design weight (sec. 4a.37(b)) divided by the design wing area (sec. 4a.38(a)).

(d) *Design power loading, W/P.* The design weight (sec. 4a.37(b)) divided by the design power (see sec. 4a.38(b) and Fig. 4a-3).

4a.39 *Air density,  $\rho$ .* The mass density of the air through which the airplane is moving, in terms of the weight of a unit volume of air divided by the acceleration of gravity. The symbol  $\rho_0$  denotes the mass density of air at sea level under standard atmospheric conditions and has the value of 0.002378 slugs per cubic foot.

*Cross Reference:* For definition of standard atmosphere, see sec. 4a.45.

### 4a.40 *Speed.*

(a) *True air speed,  $V_t$ .* The velocity of the airplane, along its flight path, with respect to the body of air through which the airplane is moving.

(b) *Indicated air speed,  $V$ .* The true air speed multiplied by the term  $\sqrt{\rho/\rho_0}$ . (See sec. 4a.39.)

(c) *Design level speed,  $V_L$ .* The indicated air speed chosen for use in determining the pertinent structural loading conditions. This value will be incorporated in the aircraft certificate as a maximum operational limitation in level and climbing flight (see sec. 4a.726).

(d) *Design gliding speed,  $V_g$ .* The maximum indicated air speed to be used in determining the pertinent structural loading conditions (see secs. 4a.73 and 4a.726).

(e) *Design stalling speed,  $V_s$ .* The computed indicated air speed in unaccelerated flight based on the maximum lift coefficient of the wing and the design gross weight.



The effects of slipstreams and nacelles shall be neglected in computing  $V_s$ . When high-lift devices are in operation the corresponding stalling speed will be denoted by  $V_{st}$ .

(f) *Design flap speed,  $V_f$* . The indicated air speed at which maximum operation of high-lift devices is assumed (see secs. 4a.73 and 4a.726).

(g) *Maximum vertical speed,  $V_m$* . A fictitious value of indicated air speed computed for unaccelerated flight in a vertical dive with zero propeller thrust.

(h) *Design maneuvering speed,  $V_p$* . The indicated air speed at which maximum operation of the control surfaces is assumed (see sec. 4a.73).

4a.41 *Design gust velocity,  $U$* . A specific gust velocity assumed to act normal to the flight path. (See sec. 4a.76.)

4a.42 *Dynamic pressure,  $q$* . The kinetic energy of a unit volume of air.

$$\begin{aligned} q &= \frac{1}{2} \rho V^2 \text{ (in terms of true air speed).} \\ &= \frac{1}{2} \rho_0 V^2 \text{ (in terms of indicated air speed).} \\ &= V^2 / 391 \text{ pounds per square foot, when } V \text{ is miles} \\ &\quad \text{per hour indicated air speed.} \end{aligned}$$

*Cross Reference:* For definition of  $\rho$ , see sec. 4a.39.

4a.43 *Load factors*—(a) *Load factor or acceleration factor,  $n$* . The ratio of a load to the design weight. When the load in question represents the net external load acting on the airplane in a given direction,  $n$  represents the acceleration factor in that direction.

(b) *Limit load*. A load (or load factor, or pressure) which it is assumed or known may be safely experienced but will not be exceeded in operation.

(c) *Factor of safety,  $j$* . A factor by which the limit loads are multiplied for various design purposes.

(d) *Ultimate factor of safety,  $j_u$* . A specified factor of safety used in determining the maximum load which the airplane structure is required to support.

(e) *Yield factor of safety,  $j_y$* . A specified factor of safety used in connection with the prevention of permanent deformations.

(f) *Ultimate load*. A limit load multiplied by the specified ultimate factor (or factors) of safety. (See definition in this section and sec. 4a.61.)

(g) *Yield load*. A limit load multiplied by the specified yield factor (or factors) of safety. (See definitions in this section and sec. 4a.62.)

(h) *Strength test*. A static load test in which the ultimate loads are properly applied. (See secs. 4a.61 and 4a.230(b)).

(i) *Proof test*. A static load test in which the yield loads are properly applied for a period of at least 1 minute. (See sec. 4a.62.)

(j) *Balancing loads*. Loads by which the airplane is placed in a state of equilibrium under the action of external forces resulting from specified loading conditions. The state of equilibrium thus obtained may be either real or fictitious. Balancing loads may represent air loads, inertia loads, or both. (See sec. 4a.116.)

4a.44 *Aerodynamic coefficients,  $C_L$ ,  $C_M$ ,  $C_P$ , etc.* The coefficients hereinafter specified are those of the "absolute" (nondimensional) system adopted as standard in the United States. The subscripts  $N$  and  $C$  used hereinafter refer respectively to directions normal to and parallel with the basic chord of the airfoil section. Other subscripts have the usual significance. When applied to an entire wing or surface, the coefficients represent average values and shall be properly correlated with local conditions (load distribution) as required in section 4a.99.

4a.45 *Standard atmosphere (standard air)*. Standard atmosphere refers to that variation of air conditions with altitude which has been adopted as standard in the United States. (See any aeronautics text book or handbook, or NACA Technical Report No. 218.)

4a.46 *Primary structure*. Those portions of the airplane the failure of which would seriously endanger the safety of the airplane.

## Subpart C—Structural Loading Conditions

### General Structural Requirements

4a.61 *Strength.* The primary structure (see sec. 4a.46) shall be capable of supporting the ultimate loads (see sec. 4a.43(f)) determined by the loading conditions and ultimate factors of safety hereinafter specified, the loads being properly distributed and applied.

4a.62 *Deformations.* The primary structure shall be capable of supporting without detrimental permanent deformations, for a period of at least one minute, the yield loads (see sec. 4a.43(g)) determined by the loading conditions and yield factors of safety hereinafter specified, the loads being properly distributed and applied. Where no yield factor of safety is specified a factor of 1.0 shall be assumed. In addition, temporary deformations which occur before the yield load is reached shall be of such a nature that their repeated occurrence will not weaken or damage the primary structure.

4a.63 *Stiffness.* The primary structure shall be capable of supporting the limit loads (see sec. 4a.43(b)) determined by the loading conditions specified in this part without deflecting beyond whatever limits may be prescribed in this part or which may be deemed necessary by the Administrator for the case in question.

4a.64 *Proof of strength and rigidity.* No general requirements, but see Subpart D for specific requirements.

4a.65 *Materials, fabrication, protection, etc.* No general requirements, but see Subpart E for specific requirements.

### Flight Loads

4a.72 *General.* The airworthiness rating of an airplane with respect to its strength under flight loads will be based on the air speeds and accelerations (from maneuvering or gusts) which can safely be developed in combination. For certain classes of airplanes the acceleration factors and gust velocities are arbitrarily specified hereinafter and shall be used for those classes. The air speeds which can safely be developed

in combination with the specified acceleration factors and gusts shall be determined in accordance with the procedure specified in this part and shall serve as a basis for restricting the operation of the airplane in flight. (See sec. 4a.726.)

### Air Speeds

4a.73 *Air speeds.* (See sec. 4a.40 for definitions.) The design air speeds shall be determined as follows:

- (a)  $V_L$  (see sec. 4a.40(c)).
- (b)  $V_g$  shall not be less than

$$V_L + K_g (V_m - V_L),$$

except that it need not be greater than either  $V_L + 100$  miles per hour or  $1.5 V_L$ , whichever is lower.  $K_g$  is specified on Figure 4a-1.  $V_m$  is defined in section 4a.40(g). A special ruling may be obtained from the Administrator if the design gliding speed thus determined is greater than  $1.33 V_L$  and appears to be unnecessarily high for the type of airplane involved.

(c)  $V_f$  shall not be less than  $2V_{sf}$ .  $V_{sf}$  is defined in section 4a.40(e).

- (d)  $V_p$  shall not be less than

$$V_{sf} + K_p (V_L - V_{sf}),$$

except that it need not be greater than  $V_L$ .  $K_p$  is specified on Figure 4a-2.

(e) (See secs. 4a.120, 4a.123, and 4a.125 for exceptions for multiengine airplanes.)

### Load Factors

4a.74 *General.* The flight load factors specified in sections 4a.75-4a.99 shall represent wing load factors. The net load factor, or acceleration factor, shall be obtained by proper consideration of balancing loads acting on the airplane in the specific flight conditions.

4a.75 *Maneuvering load factors.* The limit maneuvering load factors specified in this part (see Fig. 4a-3) are derived largely from experience with conventional types of airplanes and shall be considered

as minimum values unless it can be proved, to the satisfaction of the Administrator, that the airplane embodies features of design which make it impossible to develop such values in flight, in which case lower values may be used subject to the approval of the Administrator.

**4a.76 Gust load factors.** The gust load factors shall be computed on the basis of a gust of the magnitude specified, acting normal to the flight path, and proper allowance shall be made for the effects of aspect ratio on the slope of the lift curve. The gust velocities specified shall be used in conjunction with the gust formulas specified in Civil Aeronautics Manual 4.2121.

**4a.77 Factors of safety.** The minimum factors of safety are specified for each loading condition.

*Cross Reference:* For multiplying factors of safety required in certain cases, see secs. 4a.207-4a.216.

#### Symmetrical Flight Conditions (Flaps Retracted)

**4a.78 General.** The flight conditions as set forth in sections 4a.79-4a.85, together with Table 4a-1, shall be considered as representing the minimum number of conditions required to cover a suitable range of symmetrical flight loadings.

**4a.79 Condition I (positive high angle of attack).** The factors given in Table 4a-1 and Figure 4a-3 for this condition shall be used. To provide for flight conditions critical for the front lift truss or its equivalent the aerodynamic characteristics  $C_N$ ,  $CP$  (or  $C_M$ ), and  $C_C$  shall be determined as follows:

$$(a) \quad C_{N_I} = \frac{n_I(W/S)}{q_L}$$

( $q_L$  is dynamic pressure corresponding to  $V_L$ ; see secs. 4a.40 (c) and 4a.42.)

(b)  $C_C'$  = value corresponding to  $C_{N_I}$ , or value equal to  $-.20 C_{N_I}$ , whichever is greater negatively.

(c)  $CP'$  = most forward position of the center of pressure between  $C_L = C_{N_I}$  and  $C_{L \max.}$ ; when  $C_{N_I}$  exceeds  $C_{L \max.}$ , the  $CP$  curve shall be extended accordingly.

(d) For biplane combinations the  $CP$  of the upper wing shall be assumed to be 2.5 percent of the chord forward of its nominal position.

(e)  $C_M'$  = moment coefficient necessary to give the required  $CP'$  in conjunction with  $C_{N_I}$ .

**4a.80 Condition I<sub>1</sub> (positive high angle of attack modified).** The smaller of the two values of  $C_C$  specified in section 4a.79 (b), and the most rearward  $CP$  position in the range specified in section 4a.79 (c) shall also be investigated when Condition I is critical for the rear spar (or its equivalent) or if any portion of the front spar (or its equivalent) is likely to be critical in tension. Only the wings and wing bracing need be investigated for this condition.

**4a.81 Condition II (negative high angle of attack).** The factors given in Table 4a-1 for this condition shall be used, with the following provisions:

$$(a) \quad C_{N_{II}} = \frac{n_{II}(W/S)}{q_L}$$

(b)  $C_C$  = actual value corresponding to

$C_{N_{II}}$ .

(c) When  $C_C$  is positive or has a negative value smaller than 0.02 it may be assumed to be zero.

(d)  $C_M$  = actual value corresponding to  $C_{N_{II}}$ .

**4a.82 Condition III.**

(a) **Positive low angle of attack.** The factors given in Table 4a-1 for this condition shall be used, with the following provisions:

$$(1) \quad C_{N_{III}} = \frac{n_{III}(W/S)}{q_g}$$

( $q_g$  is dynamic pressure corresponding to  $V_g$ ; sections 4a.40 (d) and 4a.42).

(2)  $C_C$  = actual value corresponding to  $C_{N_{III}}$ .

(3) When  $C_C$  is positive or has a negative value smaller than 0.02 it may be assumed to be zero.

(4)  $C_M$  = actual value corresponding to  $C_{N_{III}}$ .

(b) *Positive low angle of attack, modified.* If the moment coefficient of the airfoil section at zero lift has a positive value, or a negative value smaller than 0.06, the effects of displaced ailerons on the moment coefficient shall be accounted for in condition III for that portion of the span incorporating ailerons. To cover this point it will be satisfactory to combine 75 percent of the loads acting in condition III with the loads due to a moment coefficient of  $-0.08 - C_{MIII}$  acting over that portion only of the span incorporating ailerons. The design dynamic pressure for the additional moment forces shall be equal to  $0.75q_0$ . Only the wings and wing bracing need be investigated for this condition.

4a.83 *Condition IV (negative low angle of attack).* The factors given in Table 4a-1 for this condition shall be used, with the following provisions:

$$(a) \quad C_{NIV} = \frac{n_{IV}(W/S)}{q_s}$$

(b)  $C_C$  = actual value corresponding to  $C_{NIV}$ .

(c) When  $C_C$  is positive or has a negative value smaller than 0.02 it may be assumed to be zero.

(d)  $C_M$  = actual value corresponding to  $C_{NIV}$ .

4a.84 *Condition V (inverted flight).* The factors given in Table 4a-1 for this condition shall be used, with the following provisions:

$$(a) \quad C_{NV} = \frac{n_V(W/S)}{q_L}$$

$$(b) \quad C_C' = 0.$$

$$(c) \quad CP' = 25 \text{ percent.}$$

(d) Only the rear (or single) lift truss system of externally braced wing structures need be investigated for this condition.

4a.85 *Condition VI (gliding).* The factors given in Table 4a-1 shall be used for this condition, with the following provisions:

(a)  $C_{NVI}$  = value corresponding to  $C_{C \max}$  (positive).

$$(b) \quad C_C' = C_{C \max} \text{ (positive)} + 0.01.$$

(c)  $C_M$  = actual value corresponding to  $C_{NVI}$ .

(d) The drag of nacelles and other items attached to the wings shall be conservatively estimated and properly included in the investigation of this condition.

(e) Only the wings and wing bracing need be investigated for this condition.

### Symmetrical Flight Conditions (Flaps or Auxiliary Devices in Operation)

4a.86 *General.* When flaps or other auxiliary high-lift devices are installed on the wings, the design conditions shall be suitably modified to account for their use in flight. The modifications shall be based on the intended use of such devices and the aerodynamic characteristics of the wing. The conditions as set forth in sections 4a.87-4a.89, together with Table 4a-2, shall be considered as representing the minimum number of conditions required to cover a suitable range of symmetrical flight loadings in cases where the flaps are used only at relatively low air speeds.

4a.87 *Condition VII (positive gust, flaps deflected).* The factors given in Table 4a-2 for this condition shall be used, with the following provisions:

(a) The most critical deflection of the flap shall be investigated.

(b) The magnitude and distribution of normal, chord, and moment forces over the wing shall correspond to that which would be obtained in developing the specified limit gust load factor at the specified air speed.

4a.88 *Condition VIII (negative gust, flaps deflected).* The factors given in Table 4a-2 for this condition shall be used, with the following provisions:

(a) The most critical deflection of the flap shall be investigated.

(b) The magnitude and distribution of normal, chord, and moment forces over the wing shall correspond to that which would be obtained in encountering the specified limit gust load factor at the specified air speed.

4a.89 *Condition IX (dive, flaps deflected).* The factors given in Table 4a-2 for this con-

dition shall be used, with the following provisions:

(a) The most critical deflection of the flap shall be investigated.

(b) The load factor and the magnitude and distribution of normal, chord, and moment forces over the wing shall correspond to the angle of attack at which the greatest rearward chord loads are produced on the wing structure.

(c) Only the wings and wing bracing need be investigated for this condition.

#### Unsymmetrical Flight Conditions

4a.90 *General*. In the unsymmetrical flight conditions set forth in sections 4a.91–4a.93, the unbalanced rolling moment shall be assumed to be resisted by the angular inertia of the complete airplane. See Civil Aeronautics Manual 4.2150 for an acceptable alternative procedure.

4a.91 *Condition I<sub>w</sub>*. Condition I (sec. 4a.79) shall be modified by assuming 100 percent of the air load acting on one wing and 40 percent on the other. For airplanes over 1,000 pounds standard weight the latter factor may be increased linearly with standard weight up to 80 percent at 25,000 pounds.

4a.92 *Condition III<sub>w</sub>*. Condition III (sec. 4a.82) shall be modified as described for condition I<sub>w</sub> in section 4a.91.

4a.93 *Condition V<sub>w</sub>*. Condition V (sec. 4a.84) shall be modified as described for condition I<sub>w</sub> in section 4a.91.

#### Special Flight Conditions

4a.94 *Gust at reduced weight*. The requirements for gust conditions (excepting tail surface gust conditions) under any loading between minimum and maximum design weight shall be met by primary structure critically loaded thereby.

4a.95 *Lift-wire-cut*. For wings employing wire bracing in the lift truss, Conditions I and III shall be investigated, using load factors  $n_I$  and  $n_{III}$  of one-half the values specified for these conditions and assuming that any lift wire is out of action. This requirement does not apply to parallel double lift wires, for which case see section 4a.210.

4a.96 *Drag-wire-cut*. Drag struts in double-truss systems shall be designed to withstand the loads developed when the drag wire of the upper system in one bay and the drag wire of the lower system in the adjacent bay are each carrying their limit loads from any flight condition, the remaining wires in these two bays being assumed to be out of action. The minimum ultimate factor of safety shall be 1.5.

4a.97 *Unsymmetrical propeller thrust*. The structure shall incorporate an ultimate factor of safety of 1.5 against failure due to loads caused by maximum (except take-off) power applied on one side of the plane of symmetry only, when power on the other side is off and the airplane is in unaccelerated rectilinear flight.

4a.98 *Wing tanks empty*. If fuel tanks are supported by the wing structure such structure and its bracing shall also be investigated for conditions I, II, III, and IV with wing tanks empty. The design weight may be reduced by 0.9 pound per certified maximum (except take-off) horsepower.

#### Wing Load Distribution

4a.99 *Wing Load Distribution*. The limit air loads and inertia loads acting on the wing structure shall be distributed and applied in a manner closely approximating the actual distribution in flight.

#### Control Surface Loads

4a.115 *General*. In addition to the flight loads specified in sections 4a.72–4a.99 the primary structure shall meet the requirements specified in this part to account for the loads acting on the control surfaces. The following loading conditions include the application of balancing loads (sec. 4a.43(j)) derived from the symmetrical flight conditions and also cover the possibility of loading the control surfaces and systems in operating the airplane and by encountering gusts. See also sections 4a.207–4a.316 for multiplying factors of safety required in certain cases.

## Horizontal Tail Surfaces

**4a.116 *Balancing.*** The limit load acting on the horizontal tail surface shall not be less than the maximum balancing load obtained from conditions I, II, III, IV, VII, and VIII set forth in sections 4a.79, 4a.81, 4a.82, 4a.83, 4a.87 and 4a.88. In computing these loads for tail surface design the moments of fuselage and nacelles shall be suitably accounted for. The factors given in Table 4a-3 shall be used with the following provisions:

(a) For conditions I, II, III, and IV,  $P$  (in Fig. 4a-4) = 40 percent of net balancing load. (This means that the load on the fixed surface should be 140 percent of the net balancing load.) In any case  $P$  need not exceed that corresponding to a limit elevator control force of 150 pounds, applied by the pilot.

(b) For conditions VII and VIII,  $P$  may be assumed equal to zero.

**4a.117 *Maneuvering (horizontal surfaces).*** The factors and distributions specified in Table 4a-3 and Fig. 4a-5 for this condition shall be used, together with the following provisions:

(a) The limit unit loading in either direction need not exceed that corresponding to a 200-pound force on the elevator control (see Table 4a-6).

(b) The average limit unit loading shall not be less than 15 pounds per square foot (see Table 4a-3).

**4a.118 *Damping (horizontal surfaces).*** The total limit load acting down on the fixed surface (stabilizer) in the maneuvering condition (sec. 4a.117) shall be applied in accordance with the load distribution of Fig. 4a-6, acting in either direction. The load acting on the movable surface in the maneuvering condition may be neglected in determining the damping loads.

**4a.119 *Tab effects (horizontal surfaces).*** When a tab is installed so that it can be used by the pilot as a trimming or assisting device, a limit up load over the tab corre-

sponding to the dynamic pressure at  $V_L$  and the maximum tab deflection shall be assumed to act in conjunction with the limit down load specified in section 4a.117, disregarding the provisions of section 4a.117 (a), applied over the remaining area. If the control force necessary to balance the resulting loads on the elevator and tab exceeds 200 pounds (Table 4a-6), the loadings over the areas not covered by the tab may be reduced until the control force is equal to this maximum limit value.

## Vertical Tail Surfaces

**4a.120 *Maneuvering.*** The factors given in Table 4a-4 and Fig. 4a-5 for this condition shall be used, with the following provisions:

(a) If the propeller axes are not in the plane of symmetry, the design speed shall not be less than the maximum speed in level flight with any engine inoperative.

(b) The limit unit loading in either direction need not exceed that corresponding to the maximum limit control force (Table 4a-6) except as modified by paragraph (c) of this section.

(c) In any case the average limit unit loading shall not be less than the minimum pressure specified in Table 4a-4 for this condition.

**4a.121 *Damping (vertical surfaces).*** The total limit load acting on the fixed surface (fin) in the maneuvering condition shall be applied in accordance with the load distribution of fig. 4a-6, acting in either direction. The load acting on the movable surface in the maneuvering condition may be neglected in determining the damping loads.

**4a.122 *Gusts (vertical surfaces).*** The gust conditions specified in Table 4a-4 shall be applied, using the following formulas and provisions:

(a) The gust shall be assumed to be sharp-edged and to act normal to the plane of symmetry in either direction.

(b) The average limit unit pressure,  $w$ , developed in striking the gust shall be determined from the following formula:

$$\bar{w} = UVm/575,$$

where

$\bar{w}$  is in pounds per square foot,

$U$  is in feet per second,

$V$  is in miles per hour, and

$m$  = slope of lift curve,  $C_L$ , per radian, corrected for aspect ratio. The aspect ratio shall not be taken as less than 2.0 in any case.

(c) This condition applies only to that portion of the vertical surface which has a well-defined leading edge.

(d) The chord distribution extending over the fixed and movable surfaces shall simulate that for a symmetrical airfoil, except that the distribution in Fig. 4a-6 may be used where applicable.

#### 4a.123 *Tab effects (vertical surfaces).*

(a) When a tab is installed on the vertical movable tail surface so that it can be used by the pilot as a trimming device the limit unit loading over the entire vertical tail surfaces shall not be less than that corresponding to the maximum deflection of the tab together with simultaneous application of the following control force in a direction assisting the tab action:

(1) For airplanes with all propeller axes in the plane of symmetry, zero.

(2) For airplanes with propeller axes not in the plane of symmetry, 200 pounds.

(b) The factors specified in Table 4a-4 for this condition shall be used, with the following exception:

(1) If the propeller axes are not in the plane of symmetry, the design speed  $V_L$  specified in Table 4a-4 may be reduced to the maximum speed in level flight with any engine inoperative.

4a.124 *Special cases (vertical surfaces).* A special ruling shall be obtained from the Administrator when an automatic pilot is used on airplanes with propeller axes not in the plane of symmetry.

#### Ailerons

4a.125 *Maneuvering.* The factors given in Table 4a-5 and Fig. 4a-7 for this con-

dition shall be used, with the following provisions:

(a) If the propeller axes are not in the plane of symmetry, the design speed shall not be less than the maximum speed in level flight with any engine inoperative.

(b) The limit unit loading in either direction need not exceed that corresponding to the maximum control force (Table 4a-6) resisted by only one aileron, except as modified by paragraph (c) of this section.

(c) In any case the average limit unit loading shall not be less than the minimum pressure specified in Table 4a-5 for this condition.

4a.126 *Tab effects (aileron).* (Applies only to airplanes with propeller axes not in the plane of symmetry.) When a tab is installed on one or both ailerons so that it can be used by the pilot to assist in moving the ailerons, the limit unit loading over both ailerons shall be of sufficient magnitude and in such direction as to hold the ailerons in equilibrium with the tab or tabs deflected to the maximum position. The factors specified in Table 4a-5 for this condition shall be used.

4a.127 *Flying conditions (aileron).* The ailerons and their control system shall be capable of meeting all requirements specified in the basic symmetrical flying conditions so far as the latter produce symmetrical loads on the ailerons.

#### Wing Flaps and Tabs

4a.128 *Wing flaps.* Wing flaps shall be loaded in accordance with conditions VII and VIII (secs. 4a.87, 4a.88) and in addition shall be capable of developing an ultimate factor of safety of at least 1.5 with respect to any intermediate conditions which are more severe for any part of the flap or its operating mechanism.

4a.129 *Tabs.* The limit forces acting on control-surface tabs shall be determined from the most severe combination of airplane speed and tab normal force coefficient likely to be obtained for any usable loading condition of the airplane and at speeds up to the

design gliding speeds,  $V_g$ . An ultimate factor of safety of at least 1.5 shall be maintained.

### Special Devices

4a.130 *Special devices.* Special rulings shall be obtained from the Administrator in connection with the design and analysis of wing-slot structures, spoilers, unconventional ailerons, auxiliary airfoils, and similar devices. Requests for special rulings shall be accompanied by suitable drawings or sketches of the structure in question, together with general information and an outline of the method by which it is proposed to determine the structural loading.

### Control System Loads

4a.137 *General.* All control systems shall be designed for limit loads 25 percent greater than those corresponding to the limit loads specified for the control surfaces to which they are attached, assuming the movable surface to be in that position which produces the greatest load in the control system, except that the maximum and minimum control force limits in Table 4a-6 shall apply as specified in this part. The factors of safety specified in Table 4a-6 shall be used.

*Cross References:* For multiplying factors of safety required in certain cases, see secs. 4a.207-4a.216. For operation requirements for control systems, see sec. 4a.271.

4a.138 *Control wires or push rods.* The forces in the control wires or push rods operating the movable surfaces shall be computed and their effect on the rest of the structure shall be investigated and allowed for in the design of such structure.

4a.139 *Elevator systems.* In applying section 4a.137 the control force specified in Table 4a-6 and fig. 4a-8 shall be assumed to act in fore-and-aft direction and shall be applied at the grip of a control stick, or shall be equally divided between two diametrically opposite points on the rim of a control wheel.

4a.140 *Rudder systems.* In applying section 4a.137 the control force specified in Table 4a-6 shall be assumed to act in a direction which will produce the greatest load

in the control system and shall be applied at the point of contact of the pilot's foot.

4a.141 *Aileron systems.* In applying section 4a.137 it shall be assumed that the ailerons are loaded in opposite directions. The control force specified in Table 4a-6 and fig. 4a-9 shall be assumed to act in a lateral direction at the grip of a control stick, or shall be assumed to act as part of couple equal to the specified force multiplied by the diameter of a control wheel. The following assumptions shall be made:

(a) For nondifferential ailerons, 75 percent of the stick force or couple shall be assumed to be resisted by a down aileron, the remainder by the other aileron; also, as a separate condition, 50 percent shall be assumed to be resisted by an up aileron, the remainder by the other aileron.

(b) For differential ailerons, 75 percent of the stick force or couple shall be assumed to be resisted by each aileron in either the up or down position, or rational assumptions based on the geometry of the system shall be made.

4a.142 *Flap and tab control systems.* In applying section 4a.137 suitable minimum manual forces shall be assumed to act on flap and tab control systems and other similar controls.

### Ground Loads

4a.147 *General.* The conditions set forth in sections 4a.148-4a.156 represent the minimum amount of investigation required for conventional (tail down type) landing gear. For unconventional types it may be necessary to investigate other landing attitudes, depending on the arrangement and design of the landing gear members. Consideration will be given to a reduction of the specified limit load factors when it can be proved that the shock absorbing system will positively limit the acceleration factor to a definite lower value in the drop test specified in section 4a.148(b). The minimum factors of safety are specified for each loading condition. See also sections 4a.207 through 4a.216 for multiplying factors of safety required in certain cases.



4a.148 *Level landing.* The minimum limit load factor is specified in fig. 4a-10. The resultant of the ground reaction shall be assumed to be a force lying at the intersection of the plane of symmetry and a plane in which are located the axles and the center of gravity of the airplane less chassis. The propeller axis (or equivalent reference line) shall be assumed horizontal and the basic value of the vertical component of the resultant of the ground reaction shall be equal to the gross weight of the airplane minus chassis and wheels. The horizontal component shall be of the magnitude required to give the resultant force the specified direction except that it need not be greater than 25 percent of the vertical component. The resultant of the ground reaction shall be assumed to be divided equally between wheels and to be applied at the axle at the center of the wheel. The shock-absorber unit and tires shall be assumed to be deflected to half their total travel, unless it is apparent that a more critical arrangement could exist. The minimum ultimate factor of safety shall be 1.5.

(a) *Sliding element.* If a sliding element instead of a rolling element is used for the landing gear, a horizontal component of one-half of the vertical component shall be used to represent the effect of ground friction, except that ski gear which is designed and used only for landing on snow and ice may be designed for the same horizontal component as wheel gear.

(b) *Energy absorption.* The level landing condition specified in section 4a.148 shall be assumed to be produced by a free drop, in inches, equal to 0.36 times the calculated stalling speed ( $V_s$ ) in miles per hour, except that the height of free drop shall not be less than 18 inches for airplanes employing devices which increase the normal sinking speed, but need not exceed 18 inches when such devices are not employed. The height of free drop is measured from the bottom of the tire to the ground, with the landing gear extended to its extreme unloaded position. (See secs. 4a.278, 4a.475.)

4a.151 *Three-point landing.* The minimum limit load factor is specified in Fig. 4a-10. The value of the sum of the static ground reactions shall be the gross weight of the airplane less chassis. The total load shall be divided between the chassis and tail skid or wheel in inverse proportion to the distances, measured parallel to the ground line, from the center of gravity of the airplane less chassis to the points of contact with the ground. The load on the chassis shall be divided equally between wheels. Loads shall be assumed to be perpendicular to the ground line in the three-point landing attitude, with all shock absorbers and tires deflected to the same degree as in level landing. The tail wheel or skid installation shall also be investigated for this condition. The minimum ultimate factor of safety shall be 1.5.

4a.152 *Energy absorption.* The three-point landing condition specified in section 4a.151 shall be assumed to be produced by a free drop as specified under section 4a.148 (b). This requires shock absorption by both main wheels and tail wheel (or skid). (See secs. 4a.278, 4a.475.)

4a.153 *Side load.* The minimum limit load factor shall be 0.667. The weight of the airplane shall be assumed to act on one wheel in a direction perpendicular to the ground. In addition, a side component of equal magnitude shall be assumed to act inward and normal to the plane of symmetry at the point of contact of the wheel, and an aft component equal to 0.55 times the vertical component shall be assumed to act parallel to the ground at such point. The airplane shall be assumed to be in a three-point attitude with the shock absorbers deflected to their static position and the tires deflected one-quarter the nominal diameter of their cross section. The minimum ultimate factor of safety shall be 1.5.

4a.154 *One-wheel landing.* An investigation of the fuselage structure is required for a one-wheel landing, in which only those loads obtained on one side of the fuselage in the level landing condition are applied. The

resulting load factor is therefore one-half of the level landing load factor. (This condition is identical with the level landing condition insofar as the landing gear structure is concerned.) The minimum ultimate factor of safety shall be 1.5.

4a.155 *Braked landing.* The minimum limit load factor shall be 1.33. Airplanes equipped with brakes shall be investigated for the loads incurred when a landing is made with the wheels locked and the airplane is in an attitude such that the tail skid or wheel just clears the ground. The weight of the airplane less chassis shall be assumed to act on the wheels in a direction perpendicular to the ground line in this attitude. In addition, a component parallel to the ground line shall be assumed to act at the point of contact of the wheels and the ground, the magnitude of this component being equal to the weight of the airplane less chassis times a coefficient of friction of 0.55. The tire in all cases shall be assumed to have deflected not more than one-quarter the nominal diameter of its cross section, and the deflection of the shock absorbers shall be the same as in level landing. The minimum ultimate factor of safety shall be 1.5.

4a.156 *Side loads on tail wheel or skid.* Suitable assumptions shall be made to cover side loads acting on tail skids or tail wheels which are not free to swivel or which can be locked or steered by the pilot.

#### Water Loads

4a.161 *General.* The requirements set forth in sections 4a.162-4a.177 shall apply to the entire airplane, but have particular reference to hull structures, wings, nacelles, and float supporting structure. The requirements for certification of floats as individual items of equipment are specified in Part 15 of this subchapter. The minimum factors of safety are specified for each loading condition.

*Cross References:* For multiplying factors of safety required in certain cases, see secs. 4a.207-4a.216. For detail design requirements for hulls and floats, see secs. 4a.488-4a.492.

#### Float Seaplanes

4a.162 *Landing with inclined reactions (float seaplanes).* The vertical component of the limit load factor shall be 4.20 except that it need not exceed a value given by the following formula:

$$n=3.0+0.133 W/S.$$

The propeller axis (or equivalent reference line) shall be assumed to be horizontal and the resultant water reaction to be acting in the plane of symmetry and passing through the center of gravity of the airplane less floats and float bracing, but inclined so that its horizontal component is equal to one-quarter of its vertical component. The forces representing the weights of and in the airplane shall be assumed to act in a direction parallel to the water reaction. The weight of the floats and float bracing may be deducted from the gross weight of the airplane.

4a.163 *Float attachment members.* For the design of float attachment members, including the members necessary to complete a rigid brace truss through the fuselage, the minimum ultimate factor of safety shall be 1.85. For the remaining structural members the minimum ultimate factor of safety shall be 1.50.

4a.164 *Landing with vertical reactions (float seaplanes).* The limit load factor shall be 4.33, acting vertically, except that it need not exceed a value given by the following formula:

$$n=3.0+0.133W/S.$$

The propeller axis (or equivalent reference line) shall be assumed to be horizontal, and the resultant water reaction to be vertical and passing through the center of gravity of the airplane less floats and float bracing. The weight of the floats and float bracing may be deducted from the gross weight of the airplane.

4a.165. *Safety factors.* The minimum factors of safety shall be the same as those specified in section 4a.163.

4a.166 *Landing with side load (float seaplanes).* The vertical component of the

limit load factor shall be 4.0, to be applied to the gross weight of the airplane less floats and float bracing. The propeller axis (or equivalent reference line) shall be assumed to be horizontal and the resultant water reaction shall be assumed to be in the vertical plane which passes through the center of gravity of the airplane less floats and float bracing and is perpendicular to the propeller axis. The vertical load shall be applied through the keel or keels of the float or floats, and evenly divided between the floats when twin floats are used. A side load equal to one-fourth of the vertical load shall be applied along a line approximately halfway between the bottom of the keel and the level of the water line at rest. When built-in struts are used, check calculations shall be made for the built-in struts with the side load at the level of the water line at rest. When twin floats are used, the entire side load specified shall be applied to the float on the side from which the water reaction originates. The minimum ultimate factor of safety shall be 1.50.

### Boat Seaplanes

#### 4a.167 *Local bottom pressures.*

(a) *Maximum local pressure.* The maximum value of the limit local pressure shall be determined from the following equation:

$$p_{max} = 0.055 V_s^{1.4} \left( 1 + \frac{W}{50,000} \right)^{1/4}$$

where

$p$  = pressure, pounds per square inch

$V_s$  = stalling speed, flaps down, power on, in miles per hour. (To be calculated on the basis of wind tunnel data or flight tests on previous airplanes.)

$W$  = design weight

The minimum ultimate factor of safety shall be 1.5.

(b) *Variation in local pressure.* The local pressures to be applied to the hull bottom shall vary in accordance with Figure 4a-11. No variation from keel to chine (beamwise) shall be assumed, except when the chine flare indicates the advisability of higher pressures of the chine.

(c) *Application of local pressure.* The local pressure determined from section 4a.167 (a) and Figure 4a-11 shall be applied over a local area in such a manner as to cause the maximum local loads in the hull bottom structure.

#### 4a.168 *Distributed bottom pressures.*

(a) For the purpose of designing frames, keels, and chine structure, the limit pressures obtained from section 4a.167(a) and Figure 4a-11 shall be reduced to one-half the "local" values and simultaneously applied over the entire hull bottom. The loads so obtained shall be carried into the sidewall structure of the hull proper, but need not be transmitted in a fore-and-aft direction as shear and bending loads. The minimum ultimate factor of safety shall be 1.5.

(b) *Unsymmetrical loading.* Each floor member or frame shall be designed for a load on one side of the hull centerline equal to the most critical symmetrical loading, combined with a load on the other side of the hull centerline equal to one-half of the most critical symmetrical loading.

#### 4a.169 *Step loading condition.*

(a) *Application of load.* The resultant water load shall be applied vertically in the plane of symmetry so as to pass through the center of gravity of the airplane (in full load condition).

(b) *Acceleration.* The limit acceleration shall be 4.33.

(c) *Hull shear and bending loads.* The hull shear and bending loads shall be computed from the inertia loads produced by the vertical water load. To avoid excessive local shear loads and bending moments near the point of water load application, the water load may be distributed over the hull bottom, using pressures not less than those specified in section 4a.168(a). The minimum ultimate factor of safety shall be 1.5.

#### 4a.170 *Bow loading condition.*

(a) *Application of load.* The resultant water load shall be applied in the plane of symmetry at a point one-tenth of the distance from the bow to the step and shall be directed upward and rearward at an angle of 30° from the vertical.

(b) *Magnitude of load.* The magnitude of the limit resultant water load shall be determined from the following equation:

$$P_s = \frac{1}{2} n_s W_e,$$

where

$P_s$  = load in pounds,

$n_s$  = step landing load factor,

$W_e$  = effective weight which is assumed equal to one-half the design weight of the airplane.

(c) *Hull shear and bending loads.* The hull shear and bending loads shall be determined by proper consideration of the inertia loads which resist the linear and angular accelerations involved. To avoid excessive local shear loads, the water reaction may be distributed over the hull bottom, using pressures not less than those specified in section 4a.168(a). The minimum ultimate factor of safety shall be 1.5.

4a.171 *Stern loading condition.*

(a) *Application of load.* The resultant water load shall be applied vertically in the plane of symmetry and shall be distributed over the hull bottom from the second step forward with an intensity equal to the pressures specified in section 4a.168(a).

(b) *Magnitude of load.* The limit resultant load shall equal three-quarters of the design weight of the airplane.

(c) *Hull shear and bending loads.* The hull shear and bending loads shall be determined by assuming the hull structure to be supported at the wing attachment fittings and neglecting internal inertia loads. This condition need not be applied to the fittings or to the portion of the hull ahead of the rear attachment fittings. The minimum ultimate factor of safety shall be 1.5.

4a.172. *Side loading condition.*

(a) *Application of load.* The resultant water load shall be applied in a vertical plane through the center of gravity. The vertical component shall be assumed to act in the plane of symmetry and the horizontal component at a point half way between the bottom of the keel and the load water line at design weight (at rest).

(b) *Magnitude of load.* The limit vertical component of acceleration shall be 3.25 and

the side component shall be equal to 15 percent of the vertical component.

(c) *Hull shear and bending loads.* The hull shear and bending loads shall be determined by proper consideration of the inertia loads or by introducing couples at the wing attachment points. To avoid excessive local shear loads, the water reaction may be distributed over the hull bottom, using pressures not less than those specified in section 4a.168(a). The minimum ultimate factor of safety shall be 1.5.

### Seaplane Float Loads

4a.173 *Seaplane float loads.* Each main float of a float seaplane shall be capable of carrying the following loads when supported at the attachment fittings as installed on the airplane. The minimum ultimate factor of safety shall be 1.5.

(a) A limit load, acting upward, applied at the bow end of the float and of magnitude equal to one-half of that portion of the airplane gross weight normally supported by the particular float.

(b) The limit load specified in paragraph (a) of this section, acting upward at the stern.

(c) A limit load, acting upward, applied at the step and of magnitude equal to 1.33 times that portion of the airplane gross weight normally supported by the particular float.

4a.174 *Seaplane float bottom loads.* Main seaplane float bottoms shall be designed to withstand the following loads. The minimum ultimate factor of safety shall be 1.5.

(a) A limit load of at least 5.33 pounds per square inch over that portion of the bottom lying between the first step and a section at 25 percent of the distance from the step to the bow.

(b) A limit load of at least 2.67 pounds per square inch over that portion of the bottom lying between the section at 25 percent of the distance from the step to the bow and a section at 75 percent of the distance from the step to the bow.

(c) A limit load of at least 2.67 pounds per square inch over that portion of the bot-

tom lying between the first and second steps. If only one step is used, this load shall extend over that portion of the bottom lying between the step and a section at 50 percent of the distance from the step to the stern.

### Wing-Tip Float Loads

4a.175 *Wing-tip float loads.* Wing-tip floats and their attachment, including the wing structure, shall be analyzed for each of the following conditions, using a minimum ultimate factor of safety of 1.5:

(a) A limit load acting vertically up at the completely submerged center of buoyancy and equal to three times the completely submerged displacement.

(b) A limit load inclined upward at 45 degrees to the rear and acting through the completely submerged center of buoyancy and equal to three times the completely submerged displacement.

(c) A limit load acting parallel to the water surface (laterally) applied at the center of area of the side view and equal to one and one-half times the completely submerged displacement.

4a.176 *Wing structure.* The primary wing structure shall incorporate sufficient extra strength to insure that failure of wing-tip float attachment members occurs before the wing structure is damaged.

### Miscellaneous Water Loads

4a.177 *Sea wing loads.* Special rulings shall be obtained from the Administrator for the strength requirements for sea wings.

### Special Loading Conditions

4a.187 *Engine torque.* In the case of engines having five or more cylinders the stresses due to the torque load shall be multiplied by a limit load factor of 1.5. For 4-, 3-, and 2-cylinder engines the limit load factors shall be 2, 3, and 4, respectively. The torque acting on the airplane structure shall be computed for the take-off power desired and the propeller speed corresponding thereto (see sec. 4a.727). The engine mount and forward portion of the fuselage and nacelles

shall be designed for this condition. The minimum ultimate factor of safety shall be 1.5 unless higher factors are deemed necessary by the Administrator in order to make special provision for conditions such as vibration, stress concentration, and fatigue.

4a.188 *High angle of attack and torque.* The limit loads determined from section 4a.187 shall be considered as acting simultaneously with 75 percent of the limit loads determined from condition I (sec. 4a.79). The engine mount, nacelles, and forward portion of the fuselage (when a nose engine is installed) shall be designed for this condition. The minimum ultimate factor of safety shall be 1.5.

4a.189 *Engine mounts, nacelles, etc.* The engine mounts, nacelles, and forward portion of the fuselage (when a nose engine is installed) shall be investigated for the limit loads determined from condition I (see secs. 4a.79 and 4a.94) acting simultaneously with the limit loads due to the engine torque determined in accordance with section 4a.187, except that the engine power and the propeller speed shall correspond to the design power (sec. 4a.38 (b)) or the output specified for climbing flight (see sec. 4a.727), whichever is higher. The minimum ultimate factor of safety shall be 1.5.

4a.190 *Side load and engine mount.* The limit load factor for this condition shall be equal to one-third of the limit load factor for flight condition I (sec. 4a.79) but shall in no case be less than 1.33. The engine mount and forward section of the fuselage and nacelles shall be analyzed for this condition, considering the limit load to be produced by inertia forces. The minimum ultimate factor of safety shall be 1.5.

4a.191 *Up load on engine mount.* For engine mounts the limit load in each member shall be arbitrarily assumed as 50 percent of that in the level landing condition but of opposite sign. The minimum ultimate factor of safety will be 1.5.

4a.192 *Passenger loads.* Passenger loads in the accelerated flight conditions shall be computed for a standard passenger weight of 170 pounds and a minimum ultimate factor

of safety of 1.50 shall be used, except that seats and berths need not be designed for the reduced weight gust conditions specified in section 4a.94. This shall not exempt the primary structure from such gust conditions.

**4a.193 Safety belt loads.** On all airplanes manufactured on or after January 1, 1951, structures, including seats, berths, and their attachments, which carry safety belt loads shall be capable of withstanding the following ultimate accelerations assumed to act upon the occupants of the belt.

Upward .....	2.0g
Forward .....	6.0g
Sidward .....	1.5g

**4a.194 Local loads.** The primary structure shall be designed to withstand local loads caused by dead weights and control loads. Baggage compartments shall be designed to withstand loads corresponding to the maximum authorized capacity. The investigation of dead weight loads shall include a sufficient number of reduced weight gust conditions to insure that the most severe combinations have been investigated.

*Cross Reference:* For standard weights, see sec. 4a.771.

**4a.195 Rigging loads.** Structures braced by wires (or tie-rods) shall be capable of developing an ultimate factor of safety of 1.5 with respect to the limit loads due to rigging the wires to 20 percent of their rated strength (strength of wire, not terminal). When the structure is such that all wires cannot be simultaneously rigged to 20 percent of their rated loads, a rigging condition shall be assumed in which the average of the rigging loads, expressed in percent, equals 20. (See also sec. 4a.211.) The above condition need not be superimposed on other loading conditions, but the Administrator may require additional investigation for residual rigging loads when such investigation appears necessary. (See also sec. 4a.253.)

**4a.196 Air loads on struts.** External wing-brace struts which are at an angle of more than 45° with the plane of symmetry and which have a cross-sectional fineness ratio of more than 3 shall be assumed to act

as lifting air foils and shall be designed to carry the resultant transverse loads in combination with the specified axial loads. In computing the limit loads the strut sections shall be assumed to have a normal force coefficient equal to 1.0 and the total air load shall be based on the exposed area of the strut. The chord components and vertical reactions of such air load and the lift contributed by the strut shall not be considered in the analysis of the wing.

### Multiplying Factors of Safety

**4a.207 General.** In addition to the minimum factors of safety specified for each loading condition, the multiplying factors specified in Table 4a-7 and sections 4a.208-4a.216 shall be incorporated in the structure. The total factor of safety required for any structural component or part equals the minimum factor of safety specified for the loading condition in question multiplied by the factors of safety hereinafter specified, except that certain multiplying factors may be included in others, as indicated in Table 4a-7.

**4a.208 Fittings.** All fittings in the primary structure shall incorporate the multiplying factor of safety specified in Table 4a-7. For this purpose fittings are defined as parts used to connect one primary member to another and shall include the bearing of those parts on the members thus connected. Continuous joints in metal plating and welded joints between primary structural members are not classified as fittings. (See also secs. 4a.320, 4a.321.)

**4a.209 Castings.** All castings used in the primary structure shall incorporate a multiplying factor of safety not less than that specified in Table 4a-7.

**4a.210 Parallel double wires.** When parallel double wires are used in wing lift trusses each wire shall incorporate a multiplying factor of safety not less than that specified in Table 4a-7.

**4a.211 Wires at small angles.** Wire or tie-rod members of wing or tail surface ex-

ternal bracing shall incorporate a multiplying factor of safety computed as follows:

$$K=L/2R \text{ (except that } K \text{ shall not be less than 1.0)}$$

where

$K$ =the additional factor.

$R$ =the reaction resisted by the wire in a direction normal to the wing or tail surface plane, and

$L$ =the load required in the wire to balance the reaction  $R$ .

**4a.212 Double drag trusses.** Whenever double drag trussing is employed, all drag wires shall incorporate a multiplying factor of safety varying linearly from 3.0, when the ratio of overhang to root chord of overhang is 2.0 or greater, to 1.20 when such ratio is 1.0 or less, assuming an equal division of drag load between the two systems.

**4a.213 Torque tubes used as hinges.** When steel torque tubes are employed in direct bearing against strap-type hinges they shall incorporate a multiplying factor of safety at the hinge point not less than that specified in Table 4a-7. (See also sec. 4a.448.)

**4a.214 Control surface hinges and control system joints.** Control surface hinges and control system joints subjected to angular motion, excepting ball or roller bearings and Army-Navy standard parts used in cable control systems, shall incorporate multiplying factors of safety not less than those specified in Table 4a-7 with respect to the ultimate bearing strength of the softest material used as a bearing. For ball or roller bearings a yield factor of safety of 1.0 with respect to the manufacturer's non-Brinell rating is considered sufficient to provide an adequate ultimate factor of safety.

**4a.215 Wire sizes.** (See secs. 4a.319, 4a.322, 4a.335.)

**4a.216 Wing lift truss system.** All structural members of the wing lift truss system which transmit direct loads from the landing gear shall, in the landing conditions, incorporate a multiplying factor of safety not less than that specified in Table 4a-7.

## Subpart D—Proof of Structure

**4a.227 General.** Proof of compliance with the loading requirements outlined in Subpart C shall be made in a manner satisfactory to the Administrator and may consist of structural analyses, load tests, flight tests, references to previously approved structures, or combinations of the above. Any condition which can be shown to be noncritical need not be further investigated.

**4a.228 Proof of structural analysis.**

(a) Structural analyses will be accepted as complete proof of strength only in the case of structural arrangements for which experience has shown such analyses to be reliable. References shall be given for all methods of analysis, formulas, theories, and material properties which are not generally accepted as standard. The acceptability of a structural analysis will depend to some extent on the excess strength incorporated in the structure.

(b) The structural analysis shall be based on guaranteed minimum mechanical properties of the materials specified on the draw-

ings, except in cases where exact mechanical properties of the materials used are determined.

(c) The effects of welding, form factors, stress concentrations, discontinuities, cut-outs, instability, end fixity of columns and vibration shall be accounted for when such factors are present to such an extent as to influence the strength of the structure.

**4a.229 Combined structural analysis and tests.** In certain cases it will be satisfactory to combine structural analysis procedure with the results of load tests of portions of the structure not subject to accurate analysis. In such cases test results shall be reduced to correspond to the mechanical properties of the materials actually used in the airplane. When a unit other than the specific one tested is incorporated in the airplane presented for certification, test results shall be reduced to correspond to the minimum guaranteed mechanical properties of the materials specified on the drawings.

**4a.230 Load tests.** Proof of compliance with structural loading requirements by means of load tests only is acceptable: *Provided*, That strength and proof tests (see sec. 4a.43 (h) and (i)) are conducted to demonstrate compliance with sections 4a.61, 4a.62, respectively: *And further provided*, That the following paragraphs of this section are complied with:

(a) The tests shall be supplemented by special tests or analyses to prove compliance with multiplying factor of safety requirements. (See secs. 4a.207-4a.216.)

(b) When a unit other than the specific one tested is incorporated in the airplane presented for certification, the results of strength tests shall be reduced to correspond to the minimum guaranteed mechanical properties of the materials specified on the drawings, unless test loads are carried at least 15 percent beyond the required values.

(c) The determination of test loads, the apparatus used, and the methods of conducting the tests shall be satisfactory to the Administrator.

(d) The tests shall be conducted in the presence of a representative of the Administrator unless otherwise directed by the Administrator.

**4a.231 Flight load tests.** Proof of strength by means of flight load tests will not be accepted unless the necessity therefor is established and the test methods are proved suitable to the satisfaction of the Administrator.

**4a.232 Load tests required.** The following load tests are required in all cases and shall be made in the presence of a representative of the Administrator unless otherwise directed by the Administrator.

(a) Strength tests of wing ribs. (See sec. 4a.248.)

(b) Pressure tests of fuel and oil tanks. (See sec. 4a.608.)

(c) Proof tests of tail and control surfaces. (See secs. 4a.263, 4a.264.)

(d) Proof and operating tests of control systems. (See secs. 4a.269, 4a.271.)

## Wings

**4a.237 Proof of wings.** The strength of stressed-skin wings shall be substantiated by load tests (sec. 4a.230) or by combined structural analysis and tests (sec. 4a.229). The torsional rigidity of the wings shall be within a range of values satisfactory for the prevention of flutter. Compliance with such torsional rigidity requirement shall be demonstrated by static tests or other methods acceptable to the Administrator.

**4a.238 Redundancies.** Wing cellules in which the division of loading between lift trusses and drag trusses is indeterminate shall be analyzed either by an acceptable method for indeterminate structures or by making assumptions which result in conservative design loads for all members.

## Beams

**4a.239 Beams.** The points set forth in sections 4a.240-4a.246 shall be covered in the proof of strength of wing beams, in addition to any special types of possible failure peculiar to the structure.

**4a.240 Secondary bending.** When axial loads are present the required minimum ultimate factor of safety shall be introduced before the computation of the bending moments in order to insure that the required ultimate loads can be supported by the structure.

**4a.241 Lateral buckling.** The ability of beams to resist lateral buckling shall be proved.

**4a.242 Webs.** The strength of shear webs shall be proved.

**4a.243 Axial load.** When axial load is present tests are required to determine the effective "EI" in the case of truss-type beams and beams having unconventional web construction.

**4a.244 Joint slippage in wood beams.** When a joint in a wood beam is designed to transmit bending from one section of the beam to another or to the fuselage, the stresses in each part of the structure shall be calculated on the assumption that the



joint is 100 percent efficient (except in mid-bay for which see sec. 4a.334) and also under the assumption that the bending moment transmitted by the joint is 75 percent of that obtained under the assumption of perfect continuity. Each part of the structure shall be designed to carry the most severe loads determined from the above assumptions.

4a.245. *Bolt holes.* In computing the area, moment of inertia, etc., of wood beams pierced by bolts, the diameter of the bolt hole shall be assumed to be one-sixteenth inch greater than the diameter of the bolt.

4a.246 *Box beams.* In computing the ability of box beams to resist bending loads only that portion of the web with its grain parallel to the beam axis and one-half of that portion of the web with its grain at an angle of 45 degrees to the beam shall be considered. The more conservative method of neglecting the web entirely may be employed.

#### Drag Trusses

4a.247 *Drag trusses.* Drag struts shall be assumed to have an end fixity coefficient of 1.0 except in cases of unusually rigid restraint, in which a coefficient of 1.5 may be used.

#### Ribs

4a.248 *Ribs.* The strength of ribs shall be proved by tests to at least 125 percent of the ultimate loads for the most severe loading conditions, except that consideration will be given to structural analyses in conjunction with suitable specimen test data when it can be demonstrated to the satisfaction of the Administrator that it is impractical to simulate the actual loading conditions in a static test. Such analyses shall, on the basis of guaranteed minimum material properties, show proof of strength at 125 percent of the required ultimate loads. The following points shall also apply in proving the strength of ribs.

4a.249 *Load distribution.* The load shall be suitably distributed between upper and

lower wing surfaces unless a more severe distribution is used.

4a.250 *Ailerons and high-lift devices.* The effects of ailerons and high-lift devices shall be properly accounted for.

4a.251 *Rib tests.* Rib tests shall simulate conditions in the airplane with respect to torsional rigidity of spars, fixity conditions, lateral support, and attachment to spars.

#### Covering

4a.252 *Covering.* Proof of strength of fabric covering is not required when standard grades of cloth and methods of attaching and doping are employed: *Provided, however,* That the Administrator may require special tests when it appears necessary to account for the effects of unusually high design air speeds or slip-stream velocities, or similar factors. When metal covering is employed its ability to perform its structural function shall be demonstrated by tests of typical panels or by other means acceptable to the Administrator. In particular, compliance with section 4a.62 requires demonstration of the behavior of the covering under load in order to determine the effects of temporary deformations (wrinkles).

#### Nonparallel Wires

4a.253 *Nonparallel wires.* When two or more wires are attached to a common point on the wing, but are not parallel, proper allowance for redundancies and the effects of rigging shall be made.

#### Tail and Control Surfaces

4a.263 *Proof of tail and control surfaces.* Structural analyses of tail and control surfaces will be accepted as complete proof of compliance with ultimate load requirements only when the structure conforms with conventional types for which reliable analytical methods are available. Proof tests as defined in section 4a.43(i) are required to prove compliance with yield load requirements.

(a) Control surface tests shall include the horn or fitting to which the control system is attached.

(b) In the analysis of control surfaces proper allowance shall be made for rigging

loads in brace wires in cases where the counter wires do not go slack before the ultimate load is reached.

(c) Analyses or individual load tests shall be conducted to demonstrate compliance with the multiplying factor of safety requirements outlined in sections 4a.207-4a.216 for control surface hinges and brace wires.

**4a.264 *Vibration tests.*** The natural frequencies of vibration of the wings, fuselage, and control surfaces shall be within such ranges of values as are satisfactory for the prevention of flutter. Compliance with this requirement shall be demonstrated by vibration tests or other methods acceptable to the Administrator.

### Control Systems

**4a.269 *Proof of control systems.*** Structural analyses of control systems will be accepted as complete proof of compliance with ultimate load requirements only when the structure conforms with conventional types for which reliable analytical methods are available. Proof tests as defined in section 4a.43 (i) are required to prove compliance with yield load requirements.

**4a.270 *Control system tests.*** In control system tests, the direction of test loads shall be such as to produce the most severe loading of the control system structure. The tests shall include all fittings, pulleys, and brackets used to attach the control system to the primary structure.

**4a.271 *Operation test.*** An operation test shall be conducted by operating the controls from the pilot's compartment with the entire system so loaded as to correspond to the minimum limit control force specified in item 3 of Table 4a-6 for the control system in question. In this test there shall be no jamming, excessive friction, or excessive deflection.

**4a.272 *Control system joints.*** Analyses or individual load tests shall be conducted to demonstrate compliance with the multiplying factor of safety requirements specified in sections 4a.207 through 4a.216 for control system joints subjected to angular motion.

### Landing Gear

**4a.277 *Proof of landing gear.*** Structural analyses of landing gear will be accepted as complete proof of compliance with load requirements only when the structure conforms with conventional types for which reliable analytical methods are available. Analyses may be used to demonstrate compliance with the energy absorption requirements in certain cases. When such analyses are not applicable, dynamic tests shall be conducted to demonstrate compliance with energy absorption requirements.

**4a.278 *Energy absorption tests.*** When tests for energy absorption are required they shall be so conducted as to simulate the landing conditions for which energy absorption requirements are specified in section 4a.475, and test data shall be obtained from which the maximum acceleration developed at the center of gravity of the airplane can be determined. When drop tests of wheels, tires, and shock absorbers are conducted in a combination differing from that employed on the airplane, proper allowance and corrections shall be made for the errors thus introduced.

### Hulls and Floats

**4a.283 *Proof of hulls and floats.*** Structural analyses of hulls and auxiliary floats will be accepted as complete proof of compliance with load requirements only when the structure conforms with conventional types for which reliable analytical methods are available. The strength of the structure as a whole and its ability to distribute water loads from the bottom plating into the main structural members shall be demonstrated. See Part 15 of this subchapter for the requirements for main floats.

### Fuselages and Engine Mounts

**4a.289 *Proof of fuselages and engine mounts.*** Structural analyses of fuselages and engine mounts will be accepted as complete proof of compliance with load requirements only when the structure conforms with conventional types for which reliable analytical methods are available.

4a.290 *Critical column loads.* The end fixity coefficient used in determining critical column loads shall in no case exceed 2.0. A value of 1.0 shall be used for all members in the engine mount. In doubtful cases, tests are required to substantiate the degree of restraint assumed.

4a.291 *Baggage compartments.* The ability of baggage compartments to sustain the maximum authorized baggage loads under all required flight and landing conditions shall be demonstrated.

#### Fittings and Parts

4a.297 *Proof of fittings and parts.* Proof of strength of all fittings and joints of the primary structure is required. Where applicable, structural analysis methods may be used. When such methods are inadequate, a

load test is required. Compliance with the multiplying factor of safety requirements for fittings (secs. 4a.207-4a.216) shall be demonstrated.

4a.298 *Fittings and attaching members.* Since the system of forces which designs a fitting does not necessarily include the forces which design the attaching members, all the forces acting in all the specified conditions shall be considered for every fitting. The strength of each part of a built-up fitting shall be investigated and proper allowance shall be made for the effects of eccentric loading when initially present or when introduced by deflection of the structure under load.

4a.299 *Bolts.* The allowable bearing load assumed for the threaded portion of a bolt shall not exceed 25 percent of the rated shear strength of the bolt.

### Subpart E—Detail Design and Construction

4a.301 *General.* The primary structure and all mechanisms essential to the safe operation of the airplane shall not incorporate design details which experience has shown to be unreliable or otherwise unsatisfactory. The suitability of all design details shall be established to the satisfaction of the Administrator. Certain design features which have been found to be essential to the airworthiness of an airplane are specified in this subpart and shall be observed.

#### Materials, Workmanship, and Fabrication Methods

4a.302 *Materials and workmanship.* The primary structure shall be made from materials which experience or conclusive tests have proved to be uniform in quality and strength and to be otherwise suitable for airplane construction. Workmanship shall be of sufficiently high grade as to insure proper continued functioning of all parts.

4a.303 *Fabrication methods.* The methods of fabrication employed in constructing the primary structure shall be such as to produce a uniformly sound structure which shall also be reliable with respect to main-

tenance of the original strength under reasonable service conditions.

4a.304 *Gluing.* Gluing may be used except in cases where inferior joints might result or where proper protection from moisture cannot be shown.

4a.305 *Torch welding.* Torch welding of primary structural parts may be used only for ferrous materials and for such other materials shown to be suitable therefor.

4a.306 *Electric welding.* Electric arc, spot, or seam welding may be used in the primary structure when specifically approved by the Administrator for the application involved. Requests for approval of the use of electric welding shall be accompanied by information as to the extent to which such welding is to be used, drawings of the parts involved, apparatus employed, general methods of control and inspection, and references to test data substantiating the strength and suitability of the welds obtained.

4a.307 *Brazing and soldering.* The use of brazing and soldering in joining parts of the primary structure is prohibited except that brazing may be used in special cases when

the suitability of the method and application can be definitely established to the satisfaction of the Administrator.

**4a.308 Protection.** All members of the primary structure shall be suitably protected against deterioration or loss of strength in service due to corrosion, abrasion, vibration, or other causes. This applies particularly to design details and small parts. In seaplanes special precautions shall be taken against corrosion from salt water, particularly where parts made from different metals are in close proximity. All exposed wood structural members shall be given at least two protective coatings of varnish or approved equivalent. Built-up box spars and similar structures shall be protected on the interior by at least one coat of varnish or approved equivalent and adequate provisions for drainage shall be made. Due care shall be taken to prevent coating of the gluing surfaces.

**4a.309 Inspection.** Inspection openings of adequate size shall be provided for such vital parts of the aircraft as require periodic inspection.

### Joins, Fittings, and Connecting Parts

**4a.312 Joints, fittings, and connecting parts.** In each joint of the primary structure the design details shall be such as to minimize the possibility of loosening of the joint in service, progressive failure due to stress concentration, and damage caused by normal servicing and field operations.

*Cross Reference:* For multiplying factors of safety required, see sec. 4a.208.

**4a.313 Bolts, pins, and screws.** All bolts and screws in the structure shall be of uniform material of high quality and of first-class workmanship. Machine screws shall not be used in the primary structure unless specifically approved for such use by the Administrator. The use of an approved locking device or method is required for all bolts, pins, and screws.

**4a.314 Wood screws.** The use of wood screws in the primary structure is prohibited except in special cases when the suitability of

the particular application is proved to the satisfaction of the Administrator.

**4a.315 Eyebolts.** Special eyebolts and similar bolts shall have a fillet between the head and the shank of at least one-fourth the diameter of the bolt when used in control surfaces or at other locations where they might be subjected to bending or vibration.

**4a.316 Castings.** Castings used in the primary structure shall incorporate the multiplying factor of safety specified in section 4a.209 and shall be of such material and design as to insure the maximum degree of reliability and freedom from defects. The Administrator has the right to prohibit the use of castings where such use is deemed to be unairworthy.

### Tie-Rods and Wires

**4a.319 Tie-rods and wires.** The minimum size of tie-rod which may be used in primary structure is No. 6-40. The corresponding minimum allowable size of single-strand hard wire is No. 13 (0.072-inch diameter).

**4a.320 Wire terminals.** The assumed terminal efficiency of single-strand hard wire shall not be greater than 85 percent.

**4a.321 Wire anchorages.** A fitting attached to a wire or cable up to and including the 3,400-pound size shall have at least the rated strength of the wire or cable, and the multiplying factor of safety for fitting (sec. 4a.208) is not required in such cases. In the case of fittings to which several tie-rods or wires are attached, this requirement applies separately to each portion of the fitting to which a tie-rod or wire is attached, but does not require simultaneous application of rated wire loads. The end connections of brace wires shall be such as to minimize restraint against bending or vibration.

**4a.322 Counter wire sizes.** (See also secs. 4a.211, 4a.212). In a wire-braced structure the wire sizes shall be such that any wire can be rigged to at least 10 percent of its rated strength without causing any other wire to be loaded to more than 20 percent of its rated strength. As used here "rated strength" refers to the wire proper, not the terminal.

### Flutter Prevention

4a.326 *General flutter prevention measures.* When he deems it necessary in the interest of safety, the Administrator may require special provisions against flutter. For specific requirements see secs. 4a.264, 4a.336, 4a.449, 4a.450, 4a.451, 4a.452, 4a.465, 4a.466 and 4a.680.

### Detail Design of Wings

4a.329 *External bracing.* When streamline wires are used for external lift bracing they shall be double unless the design complies with the lift-wire-cut condition specified in section 4a.95. (See also sec. 4a.210.)

4a.330 *Wire-braced monoplanes.* If monoplane wings are externally braced by wires only, the right and left sides of the bracing shall be independent of each other so that an unsymmetrical load from one side will not be carried through the opposite wires before being counteracted, unless the design complies with the following conditions:

(a) The minimum true angle between any external brace wire and a spar is 14 degrees.

(b) The counter (landing) wires are designed to remain in tension at least up to the limit load.

(c) The landing and flying wires are double.

4a.331 *Lift trusses.* Multiple-strand cable shall not be used in lift trusses.

4a.332 *Jury struts.* When clamps are used for attachment of jury struts to lift struts, the design shall be such as to prevent misalignment or local crushing of the lift strut.

4a.333 *Wing beams.* Provisions shall be made to reinforce wing beams against torsional failure, especially at the point of attachment of lift struts, brace wires, and aileron hinge brackets.

4a.334 *Wing beam joints.* Joints in metal beams (except pinned joints) and joints in mid-bays of wood beams shall maintain 100 percent efficiency of the beam with respect to bending, shear, and torsion.

### 4a.335 *Drag truss.*

(a) Fabric-covered wing structures having a cantilever length of overhang such that the ratio of span of overhang to chord at root of overhang is greater than 1.75 shall have a double system of internal drag trussing spaced as far apart as possible, or other means of providing equivalent torsional stiffness. In the former case counter wires shall be of the same size as the drag wires. (See also sec. 4a.212.)

(b) Multiple-strand cable shall not be used in drag trusses unless such use is substantiated to the satisfaction of the Administrator.

4a.336 *Aileron and flap attachments.* Aileron and flap attachment ribs or brackets shall be rigidly constructed and firmly attached to the main wing structure in order to reduce wing flutter tendencies.

4a.337 *Internally-braced biplanes.* Internally braced biplanes shall be provided with N or I struts to equalize deflections, and the effect of such struts shall be considered in the stress analysis.

4a.338 *Fabric covering.* Fabric covering shall comply with the requirements of section 4a.302 and shall be attached in a manner which will develop the necessary strength, with due consideration for slipstream effects. (See sec. 4a.252.)

4a.339 *Metal-covered wings.* The detail design of such wings shall incorporate suitable provision against buckling or wrinkling of metal covering as specified in sections 4a.62, 4a.252.

### Detail Design of Tail and Control Surfaces

4a.445 *Installation.* Movable tail surfaces shall be so installed that there is no interference between the surfaces or their bracing when any one is held in its extreme position and any other is operated through its full angular movement.

4a.446 *Stops.* When an adjustable stabilizer is used, stops shall be provided at the stabilizer to limit its movement, in the event of failure of the adjusting mechanism, to a

range equal to the maximum required to balance the airplane.

**4a.447 Elevator trailing edge tab systems.** Elevator trailing edge tab systems shall be equipped with stops which limit the tab travel to values not in excess of those provided for in the structural report. This range of tab movement shall be sufficient to balance the airplane under the conditions specified in section 4a.677.

**4a.448 Hinges.**

(a) Hinges of the strap type bearing directly on torque tubes are permissible only in the case of steel torque tubes which have a multiplying factor of safety as specified in section 4a.213. In other cases sleeves of suitable material shall be provided for bearing surfaces.

(b) Clevis pins may be used as hinge pins provided that they are made of material conforming with, or the equivalent of, SAE Specification 2330.

**4a.449 Elevators.** When separate elevators are used they shall be rigidly interconnected.

**4a.450 Dynamic and static balance.** All control surfaces shall be dynamically and statically balanced to the degree necessary to prevent flutter at all speeds up to the design gliding speed.

**4a.451 Wing flaps.** Flaps shall be so installed as not to induce flutter or appreciable buffeting.

**4a.452 Tabs.** The installation of trim and balancing tabs shall be such as to prevent the development of any free motion of the tab. When trailing edge tabs are used to assist in moving the main surface (balancing tabs), the areas and relative movements shall be so proportioned that the main surface is not overbalanced at any time.

### Detail Design of Control Systems

**4a.459 Installation.** All control systems and operating devices shall be so designed and installed as to provide reasonable ease of operation by the crew and so as to preclude the probability of inadvertent operation, jamming, chafing, interference by cargo, passengers, or loose objects, and the slapping

of cables against parts of the airplane. All pulleys shall be provided with satisfactory guards.

**4a.460 Stops.** All control systems shall be provided with stops which positively limit the range of motion of the control surfaces. Stops shall be capable of withstanding the loads corresponding to the design conditions for the control system.

**4a.461 Joints.** Bolts with castellated nuts safetied with cotter pins or with an approved type of self-locking nut shall be used throughout the control system, except that the use of clevis pins in standard cable ends, thimbles, and shackles is satisfactory for light airplanes.

**4a.462 Welds.** Welds shall not be employed in control systems to carry tension without reinforcement from rivets or bolts.

**4a.463 Flap controls.** The flap operating mechanism shall be such as to prevent sudden, inadvertent, or automatic opening of the flap at speeds above the design speed for the extended flap conditions. The time required to fully extend or retract flaps shall not be less than 15 seconds, unless it can be demonstrated to the satisfaction of the Administrator that the operation of the flaps in a lesser time does not result in unsatisfactory flight characteristics. Means shall be provided to retain flaps in their fully retracted position and to indicate such position to the pilot.

**4a.464-T Flap controls.**

(a) For transport category airplanes, the flap control shall provide means for bringing the flaps from any position within the operating range to any one of three positions, designated as landing, approach, and takeoff positions, or to the fully retracted position, by placing the primary flap control in a single setting marked as corresponding to each such flap position, the flaps thereupon moving directly to the desired position without requiring further attention. If any extension of the flaps beyond the landing position is possible, the flap control shall be clearly marked to identify such range of extension.

(b) The landing position, approach position, and takeoff position, or any of them, may be made variable with altitude or weight by means of a secondary flap control provided for that purpose. Such a secondary control, if provided, shall operate independently of the primary control and in such manner that when it has been adjusted (for the effect of weight or altitude), the necessary flap position can thereafter be obtained by placing the primary flap control in the desired position. The secondary control shall be so designed and marked as to be readily operable by the crew.

(c) The rate of flap retraction shall be such as to permit compliance with 4a.752-T.

**4a.465 Tab controls.**

(a) Tab controls shall be irreversible and nonflexible, unless the tab is statically balanced about its hinge line. Proper precautions shall be taken against the possibility of inadvertent or abrupt tab operation and operation in the wrong direction.

(b) When adjustable elevator tabs are used for the purpose of trimming the airplane, a tab position indicator shall be installed, and means shall be provided for indicating to the pilot a range of adjustment suitable for safe takeoff and the directions of motion of the control for noseup and nose-down motions of the airplane.

**4a.466 Spring devices.** The use of springs in the control system either as a return mechanism or as an auxiliary mechanism for assisting the pilot (bungee device) is prohibited except under the following conditions:

(a) The airplane shall be satisfactorily maneuverable and controllable and free from flutter under all conditions with and without the use of the spring device.

(b) In all cases the spring mechanism shall be of a type and design satisfactory to the Administrator.

(c) Rubber cord shall not be used for this purpose.

**4a.467 Single-cable controls.** Single-cable controls are prohibited except in special cases in which their use can be proved to be satisfactory.

**4a.468 Control system locks.** When a device is provided for locking a control surface while the aircraft is on the ground or water, compliance with the following requirements shall be shown.

(a) The locking device shall be so installed as to positively prevent taxiing the aircraft faster than 20 miles per hour, either intentionally or inadvertently, while the lock is engaged.

(b) Means shall be provided to preclude the possibility of the lock becoming engaged during flights.

**4a.469-T Trim controls.** For transport category airplanes, the trimming devices shall be capable of continued normal operation in spite of the failure of any one connecting or transmitting element in the primary control system. Trim controls shall operate in the plane and with the sense of the motion of the airplane which their operation is intended to produce.

### Detail Design of Landing Gear

**4a.475 Shock absorption.** All landing gear (including tail gear installations) shall be provided with shock-absorbing systems which will permit the airplane to be landed under the conditions specified in sections 4a.148(b), 4a.152 without exceeding the ultimate load used in the analysis of any landing gear member. (See sec. 4a.278 for proof of absorption capacity.) If the design of the shock-absorbing system is such that the above method of specifying the required energy absorption capacity appears to give irrational results, an alternate method will be considered upon presentation of pertinent data.

**4a.476 Shock-absorbing systems.** The shock-absorbing systems employed shall incorporate suitable means for absorbing the shocks developed in taxiing or running over rough ground.

**4a.477 Wheels.** Main landing gear wheels shall be of a type or model certificated by the Administrator in accordance with the provisions of Part 15 of this subchapter and shall not be subjected to static loads in excess of those for which they are

certificated. Tail wheels may be of any type or model and are not certificated. Nose wheels are subject to special rulings to be made by the Administrator.

4a.478 *Main landing gear wheels.* For the purpose of the regulations in this part main landing gear wheels are considered as those nearest the airplane center of gravity with respect to fore-and-aft location.

4a.479 *Tail and nose wheels.* For the purpose of the regulations in this part, a tail wheel is considered as one which supports the tail of a conventional airplane in the three-point landing attitude. A nose wheel is considered to be a wheel supporting the nose of the airplane when the two main wheels are located behind the center of gravity.

4a.480 *Tires.* A landing gear wheel may be equipped with any make or type of tire, provided that the tire is a proper fit on the rim of the wheel and provided that the tire rating of the Airplane Tire Committee of the Tire and Rim Association is not exceeded.

4a.481 *Tire markings.* When specially constructed tires are used to support an airplane, the wheels shall be plainly and conspicuously marked to that effect. Such markings shall include the make, size, number of plies, and identification marking of the proper tire.

4a.482 *Retracting mechanism.*

(a) When retractable landing wheels are used visual means shall be provided for indicating to the pilot, at all times, the position of the wheels. Separate indicators for each wheel are required when each wheel is separately operated unless a single indicator is obviously satisfactory. In addition, landplanes shall be provided with an aural or equally effective indicator which shall function continuously after the throttle is closed until the gear is down and locked.

(b) A positive lock shall be provided for the wheels in the extended position, unless a rugged irreversible mechanism is used.

(c) Manual operation of retractable landing gears shall be provided for.

4a.483-T *Brakes.* Transport category airplanes shall be equipped with brakes cer-

tificated in accordance with the provisions of Part 15 of this subchapter, for the maximum certificated landing weight at sea level and the power-off stalling speed,  $V_{SO}$ , as defined in section 4a.739-T. The brake system for such airplanes shall be so designed and constructed that in the event of a single failure in any connecting or transmitting element in the brake system, or the loss of any single source of hydraulic or other brake operating energy supply, it shall be possible, as shown by suitable test or other data, to bring the airplane to rest under the conditions specified in section 4a.750-T with a mean negative acceleration during the land roll of at least 50 percent of that obtained in determining the landing distance under that section.

### Hulls and Floats

4a.488 *Hulls and floats.* (See also secs. 4a.497 through 4a.513.)

4a.489 *Buoyancy (main seaplane floats).*

(a) Main seaplane floats shall have a buoyancy in excess of that required to support the gross weight of the airplane in fresh water as follows:

(1) 80 percent in the case of single floats,

(2) 90 percent in the case of double

floats.

(b) Main seaplane floats for use on aircraft of 2,500 pounds or more maximum authorized weight shall contain at least five watertight compartments of approximately equal volume. Main seaplane floats for use on aircraft of less than 2,500 pounds maximum authorized weight shall contain at least four such compartments.

4a.490 *Buoyancy (boat seaplanes).* The hulls of boat seaplanes and amphibians shall be divided into watertight compartments in accordance with the following requirements:

(a) In aircraft of 5,000 pounds maximum authorized weight or more the compartments shall be so arranged that, with any two adjacent compartments flooded, the hull and auxiliary floats (and tires, if used) will retain sufficient buoyancy to support the gross weight of the aircraft in fresh water.



(b) In aircraft of 1,500 to 5,000 pounds maximum authorized weight the compartments shall be so arranged that, with any one compartment flooded, the hull and auxiliary floats (and tires, if used) will retain sufficient buoyancy to support the maximum authorized weight of the aircraft in fresh water.

(c) In aircraft of less than 1,500 pounds maximum authorized weight watertight subdivision of the hull is not required.

(d) Bulkheads may have watertight doors for the purpose of communication between compartments.

**4a.491 Water stability.** Auxiliary floats shall be so arranged that when completely submerged in fresh water, they will provide a righting moment which is at least 1.5 times the upsetting moment caused by the aircraft being tilted. A greater degree of stability may be required in the case of large flying boats, depending on the height of the center of gravity above the water level, the area and location of wings and tail surfaces, and other considerations.

**4a.492 Float design.** In designing the bow portion of floats and hulls suitable provision shall be made for the effects of striking floating objects.

### Fuselage and Cabins

**4a.497 Provision for turn-over.** The fuselage and cabins shall be designed to protect the passengers and crew in the event of a complete turn-over and adequate provision shall be made to permit egress of passengers and crew in such event. The requirements of this section may be suitably modified when the possibility of a complete turn-over in landing is remote.

**4a.498 External door.** Closed cabins on all aircraft carrying passengers shall be provided with at least one adequate and easily accessible external door.

**4a.499 Location of passenger door.** No passenger door shall be located in the plane of rotation of an inboard propeller, nor within 5 degrees thereof as measured from the propeller hub.

**4a.500 Exits.** Closed cabins on aircraft carrying more than 5 persons shall be provided with emergency exits, in addition to the one external door required by section 4a.498, consisting of movable windows or panels or of additional external doors which provide a clear and unobstructed opening, the minimum dimensions of which shall be such that a 19-inch by 26-inch ellipse may be completely inscribed therein. The location and the method of operation of emergency exits shall be approved by the Administrator. If the pilot is in a compartment separate from the cabin, passage through such compartment shall not be considered as an emergency exit for the passengers. The number of emergency exits required is as follows:

(a) Aircraft with a total seating capacity of more than 5 persons, but not in excess of 15, shall be provided with at least one emergency exit or one suitable door in addition to the main door specified in section 4a.498. This emergency exit, or second door, shall be on the opposite side of the cabin from the main door. If desired, an additional emergency exit may be provided in the top of the cabin, but such an installation shall not obviate the necessity for an exit on each side.

(b) Aircraft with a seating capacity of more than 15 persons shall be provided with an additional emergency exit or door either in the top or side of the cabin for every additional 7 persons or fraction thereof above 15, except that not more than 4 exits, including doors, will be required if the arrangement and dimensions are suitable for the purpose intended.

### Pilot Compartment

**4a.501 Construction.** The pilot compartment shall be so constructed as to afford suitable ventilation and adequate vision to the pilot under normal flying conditions. In cabin aircraft the windows shall be so arranged that they may be readily cleaned or easily opened in flight to provide forward vision for the pilot. The ventilation requirements of section 4a.510 shall also apply to the pilot compartment.

**4a.502 Location.** The pilot and the primary control units, excluding cables and control rods, shall be so located with respect to the propellers that no portion of the pilot or controls lies in the region between the plane of rotation of any propeller and the surface generated by a line passing through the center of the propeller hub and making an angle of 5 degrees forward or aft of the plane of rotation of the propeller.

**4a.504 Operation information and limitations.** Means shall be provided by which the operating personnel is suitably informed of all operation information and limitations deemed necessary by the Administrator.

**4a.505 Windows and windshields.** The windows and windshields of the pilot compartment in airplanes certificated for air transportation service shall be so arranged as to provide satisfactory forward vision and protection under all conditions and, to accomplish this, particular attention shall be paid to the following detail requirements:

(a) Sufficient data specifying the windshield material, number of laminations, binder if any, size and shape of panes, angle of panes to flight path, and method and rigidity of mounting, shall be forwarded to the Administrator for rulings as to the acceptability of the windshield from the standpoint of strength.

(b) Windshields shall be so installed that they can be easily opened in flight and shall be so arranged that the air stream and snow or rain are deflected across the opening, or to provide equivalent results.

(c) The pilot compartment shall be so constructed and arranged as to prevent glare or reflections which would interfere with the vision of either pilot, particularly while flying at night. The aircraft will be flown by a representative of the Administrator during hours of darkness to determine compliance with this provision.

**4a.506 Leakage.** The pilot compartment in airplanes certificated for air transportation service shall be so constructed as to pre-

vent any leakage into it when the airplane is flying in rain or snow.

**4a.507 Seats.** When a second pilot is required (sec. 61.121 of this subchapter) two seats shall be installed side by side in the pilot compartment of airplanes certificated for air transportation service from either of which the airplane shall be fully and readily controllable. If any difference exists as to convenience of the instruments and controls necessary for safe flight such difference should favor the left-hand seat. The left-hand seat shall be known as the first pilot's seat and the right-hand one as the second pilot's seat.

**4a.508 Navigation instruments.** The navigation instruments for use by the pilot in airplanes certificated for air transportation service shall be so installed as to be easily visible to him with the minimum practicable deviation from his normal position and line of vision when he is looking out and forward along the flight path and they shall also be visible to the second pilot.

**4a.509 Opening between pilot compartment and passengers' cabin.** All airplanes certificated for air transportation service shall be provided with a door or an adequate openable window between the pilot compartment and the passenger cabin. When a door is provided it shall be equipped with a locking means which shall prevent passengers from opening such door while in flight.

### Passenger and Baggage Compartments

**4a.510 Passenger compartments.** A suitable ventilation system shall be provided which will preclude the presence of fuel fumes and dangerous traces of carbon monoxide in each passenger compartment.

**4a.511 Passenger chairs.** Seats or chairs for passengers shall be securely fastened in place in both open and closed airplanes, whether or not the safety belt load is transmitted through the seat.

*Cross References:* For safety belt requirements, see secs. 4a.193, 4a.532(i), and 4a.565.

**4a.512 Baggage compartments.** Each baggage and mail compartment shall bear

a placard stating the maximum allowable weight of contents, as determined by the structural strength of the compartment (sec. 4a.194) and by flight test (sec. 4a.725). Suitable means shall be provided to prevent the contents of mail and baggage compartments from shifting.

## Subpart F—Equipment

**4a.523. General.** The equipment required shall be dependent upon the type of operation for which certification is to be made. The requirements specified in this subpart shall be the basic equipment requirements and such additional equipment as may be specified in other sections of the Civil Air Regulations for specific special cases shall be supplemental hereto unless otherwise specified.

**4a.524 Requirements.** Each item of equipment specified in the Civil Air Regulations shall be of a type and design satisfactory to the Administrator, shall be properly installed, and shall function to the satisfaction of the Administrator. Items of equipment for which certification is required shall have been certificated in accordance with the provisions of Part 15 of this subchapter or previous regulations.

**4a.524-1 Approval of aircraft components** (*FAA rules which apply to sec. 4a.524*). Aircraft components made the subject of Technical Standard Orders shall be approved upon the basis and in the manner provided in Part 514 of this title.

[Supp. 9, 16 F.R. 671, Jan. 25, 1951]

**4a.525 Life preserver or flotation device.** An approved life preserver or flotation device is one approved by the Administrator for such usage on sea-going vessels.

**4a.526 Fire extinguishing apparatus.** Fire extinguishing apparatus approved by the Underwriters Laboratories is considered to be of an approved type.

## Reinforcement Near Propellers

**4a.513 Reinforcement near propellers.** Surfaces near propeller tips shall be suitably stiffened against vibration and effects of ice thrown from the propeller.

*Cross Reference:* For clearance requirements, see sec. 4a.599.

## Non-Air Carrier (NAC) Airplanes

**4a.531 Non-air carrier (NAC) airplanes.** Airplanes which are certificated as non-air carriers shall have at least the equipment set forth in sections 4a.532-4a.537.

**4a.532 NAC landplanes; visual contact day flying (within 100 miles of a fixed base).**

(a) One air-speed indicator.

*Cross Reference:* For installation requirements, see sec. 4a.559.

(b) One altimeter.

(c) A tachometer for each engine.

(d) An oil-pressure gauge when an oil-pressure system is employed.

(e) A water thermometer for each water-cooled engine.

(f) An oil thermometer for each air-cooled engine.

(g) A manifold-pressure gauge, or equivalent, for each altitude engine.

(h) A fuel quantity gauge. (See sec. 4a.609 for requirements.)

(i) Approved safety belts for all occupants (see sec. 4a.565).

*Cross References:* For belt requirements, see Part 15 of this subchapter. For installation requirements, see sec. 4a.565.

(j) A portable fire extinguisher, which extinguisher shall be of an approved type, which shall have a minimum capacity, if carbon tetrachloride, of 1 quart, or, if carbon dioxide, of 2 pounds, or, if other, of equivalent effectiveness; except that any extinguisher of not less than half the above capacity may be used in an airplane equipped with an engine whose maximum rating is

40 horsepower or less. (See sec. 4a.566 for installation requirements.)

(k) Landing gear position indicator for retractable main landing gear. (See sec. 4a.482 for requirements.)

(l) A device for measuring or indicating the amount of oil in the tanks. (See sec. 4a.624 for requirements.)

(m) A first-aid kit.

(n) A logbook for the airplane and one for each engine. (See Part 1 of this subchapter for requirements.)

(o) Rigging information for airplanes with wire-braced wings, either in the form of a sketch or listed data, which shall include sufficient information to permit proper rigging.

**4a.533 NAC landplanes; visual contact day flying (unlimited distance).** Airplanes of this category shall have the equipment specified in section 4a.532 and, in addition, there shall be installed:

(a) A magnetic compass.

*Cross Reference:* For installation requirements, see sec. 4a.562.

**4a.534 NAC landplanes; visual contact night flying.** Airplanes of this category shall have the equipment specified in section 4a.533 and, in addition, there shall be installed:

(a) A set of certificated standard forward position lights in combination with a certificated tail light.

*Cross References:* For light requirements, see Part 15 of this subchapter. For installation requirements, see sec. 4a.578.

(b) Two electric landing lights if the aircraft is operated for hire: *Provided, however,* That only one such landing light shall be required for any airplane certificated for a weight of less than 1,500 pounds. (See sec. 4a.576 for installation requirements.)

(c) Certificated landing flares as follows, if the aircraft is operated for hire beyond an area within a circle with a radius of 3 miles drawn from the center of the airport of takeoff (see Part 15 of this subchapter for flare requirements and sec. 4a.568 for installation requirements):

(1) Airplanes of 3,500 pounds maximum authorized weight or less—five Class 3 flares or three Class 2 flares.

(2) Airplanes of between 3,500 pounds and 5,000 pounds maximum authorized weight—four Class 2 flares.

(3) Airplanes of 5,000 pounds maximum authorized weight or more—two Class 1 flares or three Class 2 flares and one Class 1 flare.

(4) If desired, airplanes of less than 5,000 pounds maximum authorized weight may carry the flare equipment specified for heavier airplanes.

(d) A storage battery suitable as a source of energy supply for such lights and radio as are installed.

*Cross Reference:* For installation requirements, see sec. 4a.571.

(e) Radio equipment, if the aircraft is operated in a control zone (sec. 60.13 of this subchapter), as follows: A radio range and weather broadcast receiver operating within the frequency range of 200 to 400 kilocycles. Under normal atmospheric conditions this receiver must be capable of receiving with a range of 100 miles intelligence emanated from a radio range or weather broadcast station the equivalent of a simultaneous Adcock range with scheduled broadcast installation.

(f) A set of spare fuses.

*Cross Reference:* For installation requirements, see sec. 4a.572.

**4a.535 NAC landplanes; instrument day flying.** Airplanes of this category shall have the equipment specified in section 4a.533 and, in addition, there shall be installed:

(a) Radio equipment: Same as section 4a.534(e), whether the aircraft is operated for hire or not, and, in addition, a radio transmitter operated on 3105 kilocycles with a power output sufficient to establish communication at a distance of at least 100 miles under normal atmospheric conditions. Additional frequencies may be employed subject to approval of the Federal Communications Commission.

(b) A gyroscopic rate-of-turn indicator.

(c) A bank indicator. (The rate-of-turn indicator may be combined in the bank indicator if desired.)

(d) A sensitive altimeter which shall be adjustable for changes in barometric pressure and compensated for changes in temperature.

(e) A clock with a sweep second hand.

(f) A storage battery suitable as a source of energy supply for the radio equipment installed.

*Cross Reference:* For installation requirements, see secs. 4a.571, 4a.573.

(g) A generator.

(h) A set of spare fuses.

*Cross Reference:* For installation requirements, see sec. 4a.572.

(i) A rate-of-climb indicator.

**4a.536 NAC landplanes; instrument night flying.** Airplanes of this category shall have the equipment specified in sections 4a.534, 4a.535 combined. The storage battery shall be suitable as a source of energy supply for both the radio equipment and the lights.

**4a.537 NAC seaplanes and amphibians.** The equipment requirements for seaplanes and amphibians shall be the same as specified for landplanes (secs. 4a.532-4a.536) except that seaplanes and amphibians shall not be certificated for operation over water out of sight of land unless they have at least the equipment specified in section 4a.533, and except that all certificated seaplanes and amphibians shall also have an approved life preserver or flotation device for each person for whom there is a seat, and except that all seaplanes and amphibians certificated for night operation shall also have a white anchor light.

#### **Air Carrier Airplanes; Passenger (ACP)**

**4a.547 Air carrier airplanes; passengers (ACP).** Airplanes certificated for use by an air carrier in passenger service shall have installed at least the equipment set forth in sections 4a.548-4a.552.

**4a.548 ACP landplanes; visual contact day flying.** The same as specified in section 4a.533 and, in addition, the following:

(a) An electrically heated pitot tube, or equivalent, for the air-speed indicator.

(b) One additional portable fire extinguisher of the type specified in section 4a.532 (j). (See sec. 4a.566 for installation requirements.)

(c) Fixed fire extinguishing apparatus of an approved type of each engine compartment.

(d) Type certificated radio equipment as specified in Part 40 of this subchapter.

(e) A set of spare fuses. (See sec. 4a.572 for installation requirements.)

(f) A rate-of-climb indicator.

(g) A storage battery: Same as section 4a.535(f).

(h) A means for providing, without continuous manual operation, vision through the windshield adequate for executing take-offs and landings in rain.

**4a.549 ACP landplanes; visual contact night flying.** The same as specified in section 4a.548 and, in addition, the following:

(a) A set of certificated air carrier airplane position lights. The forward lights may be air carrier forward position lights or a combination of standard forward position lights and a set of auxiliary forward position lights. (See Part 15 of this subchapter for light requirements and sec. 4a.578 for installation requirements.)

(b) A storage battery of sufficient capacity for such lights and radio as are installed.

*Cross Reference:* For installation requirements, see secs. 4a.571, 4a.573.

(c) Two electric landing lights.

*Cross Reference:* For installation requirements, see sec. 4a.576.

(d) Certificated landing flares as follows: two Class 1 flares or three Class 2 flares and one Class 1 flare.

*Cross References:* For flare requirements, see Part 15 of this subchapter. For installation requirements, see sec. 4a.568.

(e) Instrument lights.

*Cross Reference:* For installation requirements, see sec. 4a.577.

(f) Cabin lights in all passenger cabins and compartments.

(g) A generator. (See sec. 4a.573 for requirements.)

(h) Radio equipment same as section 40.61.

4a.550 *ACP landplanes; instrument day flying.* The same as specified in section 4a.548 except section 4a.548(b) and, in addition, the following:

(a) A gyroscopic rate-of-turn indicator combined with a bank indicator.

(b) A gyroscopic instrument showing bank and pitch.

(c) A gyroscopic direction finder.

(d) Two sensitive-type altimeters, both of which shall be adjustable for changes in barometric pressure and compensated for changes in temperatures: *Provided*, That aircraft in use on or before January 1, 1939, and thereafter replacements and additions of aircraft of the same make and model may, for purposes of standardization, be deemed to have met this requirement, if there are installed in each such aircraft one sensitive type altimeter and one standard type altimeter, provided each is adjustable for changes in barometric pressure and compensated for changes in temperature.

(e) A free air thermometer of the distance type with an indicating dial in the cockpit.

(f) A clock with a sweep second hand.

(g) A vacuum gauge, installed in the lines leading to instruments in paragraphs (a), (b) and (c) of this section.

(h) Type certificated radio equipment as specified in Part 40 of this subchapter.

(i) Means shall be provided to indicate icing conditions, or the probability thereof, in the carburetor if the de-icing device specified in section 4a.616 requires the manual manipulation of controls.

(j) A storage battery suitable as a source of energy supply for the radio equipment installed.

*Cross Reference:* For installation requirements, see secs. 4a.571, 4a.573.

(k) A generator.

*Cross Reference:* For installation requirements, see sec. 4a.573.

4a.551 *ACP landplanes; instrument night flying.* The same as specified in sections 4a.549 and 4a.550 combined. The storage

battery, in this case, shall be of sufficient capacity for all radio equipment and all lights installed.

4a.552 *ACP seaplanes and amphibians.* The same as specified for landplanes. (Secs. 4a.548-4a.551) and including the life preservers specified in section 4a.537, except that when certificated for night operation they shall also have installed the anchor light specified in section 4a.537.

### Installation Requirements

4a.557 *Installation requirements.* The regulations in sections 4a.558-4a.581 apply to the installation of specific items of equipment and are additional to the regulations of section 4a.523.

### Instrument Installation

4a.558 *Instruments.* The regulations in sections 4a.559-4a.564 shall apply to the installation of instruments when such instruments are required by the regulations in this part.

4a.559 *Air-speed indicator.* This instrument shall be so installed as to indicate true air speed at sea level with the maximum practicable accuracy, but the instrument error shall not be more than plus or minus 3 percent, except that it need not be less than plus or minus 5 miles per hour, at the level flight speed corresponding to the design power (sec. 4a.38(b)), at  $V_L$  (sec. 4a.40(c)), or at the maximum attainable level flight speed, whichever is lowest.

4a.560 *Powerplant instruments and controls.* See sections 4a.642, 4a.643.

4a.561 *Fuel quantity gauge.* See section 4a.609.

4a.562 *Magnetic compass.* This instrument shall be properly damped and compensated and shall be located where it is least affected by electrical disturbances and magnetic influences.

4a.563 *Navigational instruments.* Navigational instruments for use by the pilot shall be so installed as to be easily visible to him with the minimum practicable deviation from his normal position and line of

vision when he is looking out and forward along the flight path, and they shall also be visible to the second pilot.

**4a.564 Gyroscopic instruments.** All gyroscopic instruments shall derive their energy from engine-driven pumps or from auxiliary power units. Each source of energy supply and its attendant complete installation shall comply with the instrument manufacturer's recommendations for satisfactory instrument operation. On multiengine aircraft each instrument shall have two separate sources of energy, either one of which shall be capable of carrying the required load. Engine-driven pumps, when used, shall be on separate engines. The installation shall be such that failure of one source of energy or breakage of one line will not interfere with proper functioning of the instruments by means of the other source.

#### Safety Equipment Installation

**4a.565 Safety belts.** Airplanes manufactured on or after January 1, 1951 shall be equipped with safety belts approved in accordance with section 4a.31. In no case shall the rated strength of the safety belt be less than that corresponding with the ultimate load factors specified in section 4a.193, taking due account of the dimensional characteristics of the safety belt installation for the specific seat or berth arrangement. Safety belts shall be attached so that no part of the anchorage will fail at a load lower than that corresponding with the ultimate load factors specified in section 4a.193.

(Sec. 205(a), 52 Stat. 984; 49 U.S.C. 425(a). Interpret or apply secs. 601, 603, 52 Stat. 1007, 1009; 49 U.S.C. 551, 553.)

**4a.566 Fire extinguishers.** The portable fire extinguisher specified in section 4a.532 (j) shall be so installed as to be accessible to the passengers. The two portable fire extinguishers specified in section 4a.548 shall be so installed that one is readily available to the crew and the other is near the main external cabin door where it shall be readily available to passengers and ground personnel.

**4a.567 Safety belt signal.** When a signal or sign is used to indicate to passengers the times that seat belts should be fastened, such signal or sign shall be located in a conspicuous place and so arranged that it can be operated from the seat of either pilot.

**4a.568 Landing flares.** Landing flares shall be releasable from the pilot compartment. Structural provision shall be made for the recoil loads.

**4a.569 De-icers.** Positive means shall be provided for the deflation of all wing boots.

#### Electrical Equipment Installation

**4a.570 General.** Electrical equipment shall be installed in accordance with accepted practice and suitably protected from fuel, oil, water, and other detrimental substances. Adequate clearance shall be provided between wiring and fuel and oil tanks, fuel and oil lines, carburetors, exhaust piping, and moving parts.

**4a.571 Battery.** Batteries shall be easily accessible and adequately isolated from fuel, oil, and ignition systems. Adjacent parts of the aircraft structure shall be protected with a suitable acid-proof paint if the battery contains acid or other corrosive substance and is not completely enclosed. If the battery is completely enclosed, suitable ventilation shall be provided. All batteries shall be so installed that spilled liquid will be suitably drained or absorbed without coming in contact with the airplane structure.

**4a.572 Fuses.** Fuses shall be so located that they can readily be replaced in flight. They shall break the current in a generating system at a sufficiently small current flow adequately to protect the lights, radio equipment, and other parts of the circuit.

**4a.573 Generator.** When a generator is specified it shall have sufficient capacity to carry the entire running load. Such generator shall be engine driven unless an approved equivalent system is provided. Auxiliary power units will be approved in lieu of batteries and engine-driven generators, provided that they are at least two in number and that the supply system is capable of carrying

the entire running load with any one unit out of action.

**4a.574 *Running load.*** The running load shall be defined as the electric consumption of all lights, radio equipment, and other electrical devices, except those which are designed only for occasional intermittent use. Examples of devices regarded as intermittent are radio broadcasting equipment, landing lights, and electrically operated landing gears and wing flaps. Radio range signal receivers and all other lights are considered a part of the constant load.

**4a.575 *Anchor lights.*** The anchor light specified for seaplanes and amphibians shall be so mounted and installed that, when the airplane is moored or drifting on the water, it will show a white light visible for at least 2 miles at night under clear atmospheric conditions.

**4a.576 *Landing lights.*** Electric landing lights shall be so installed on multiengine aircraft that at least one shall be not less than 10 feet to the right or left of the first pilot's seat and beyond the swept disk of the outermost propeller. On single-engine aircraft such lights shall be so installed that no visible portion of the swept disk of the propeller, if of the tractor type, is illuminated thereby. Individual switches for each light shall be provided in the pilot compartment.

**4a.577 *Instrument lights.*** Instrument lights shall be so installed as to provide sufficient illumination to make all flight instruments easily readable and shall be equipped with rheostat control for dimming unless it can be shown that a nondimming light is satisfactory.

**4a.578. *Position lights.*** Position lights shall be installed so that, with the airplane in normal flying position, the forward red position light is displayed on the left side and

the forward green position light on the right side, each showing unbroken light between two vertical planes whose dihedral angle is 110 degrees when measured to the left and right, respectively, of the airplane from dead ahead. Such forward position lights shall be spaced laterally as far apart as practicable. One rear position light shall be installed on the airplane at the rear and as far aft as possible and shall show a light visible aft throughout a dihedral angle of 140 degrees bisected by a vertical plane through the longitudinal axis of the airplane. Such light shall emit (a) in the case of a non-air carrier airplane, either a continuous white light as specified in section 15.20(b)(5) of this subchapter, or alternate red and white flashes as specified in section 15.20(b)(6) of this subchapter, and (b) in the case of an air carrier airplane, alternate red and white flashes as specified in section 15.20(b)(6) of this subchapter. In lieu of such a single flashing rear position light, an airplane may carry two rear position lights, one red and one white, spaced as closely as possible to each other and in combination emitting the red and white flashes specified in section 15.20(b)(6) of this subchapter.

**4a.579 *Master switch.*** Electrical installations shall incorporate a master switch easily accessible to a member of the crew.

### Miscellaneous Equipment Installation

**4a.580 *Seats.*** Seats or chairs, even though adjustable, in open or closed airplanes, shall be securely fastened in place whether or not the safety belt load is transmitted through the seat.

**4a.581 *Accessories.*** Engine-driven accessories on multiengine aircraft shall be distributed among two or more engines.

## Subpart G—Powerplant Installation

### Engines

**4a.591 *Engines.*** Engines shall be of a type and design which has been type certifi-

cated, or found eligible for use in certified aircraft, in accordance with the re-



quirements of Part 13 of this subchapter or shall have been approved as airworthy in accordance with previous regulations.

### Propellers

**4a.597 Propellers.** Propellers shall be of a type and design which has been certificated as airworthy in accordance with the requirements of Part 14 of this subchapter or shall have been approved as airworthy in accordance with previous regulations, except that wood propellers of a conventional type for use in light airplanes need not be certificated. In certain cases maximum engine bore limitations are also assigned to propellers. Propellers may be used on any engine provided that the certified power ratings, speed ratings, and bore of the engine are not in excess of the limitations of the propeller as certificated, and further provided that the vibration characteristics of the combination are satisfactory to the Administrator.

**4a.598 Controllable pitch.** The control mechanism shall be designed and equipped with a positive stop which shall limit the minimum pitch so that the takeoff crankshaft speed for which the aircraft is certificated is not exceeded during takeoff with takeoff power unless it is necessary to so locate the stop that a higher crankshaft speed may be used in an emergency. The means provided for controlling the pitch shall be so arranged as to minimize the attention required from a pilot to prevent the engines from exceeding their crankshaft speed limitations under any flight condition.

**4a.599 Propeller clearance.** Propellers shall have a minimum ground clearance of 9 inches when the airplane is in a horizontal position with the landing gear deflected as it would be under the maximum authorized weight of the airplane. Propellers on seaplanes shall clear the water by at least 18 inches when the seaplane is at rest under the maximum authorized load condition. A clearance of at least 1 inch shall be provided between the tips of the propellers and any part of the structure.

### Fuel Systems

**4a.605 Capacity and feed.** The fuel capacity shall be at least 0.15 gallons per maximum (except takeoff) horsepower for which the airplane is certificated. Air-pressure fuel systems shall not be used. Only straight gravity feed or mechanical pumping of fuel is permitted. The system shall be so arranged that the entire fuel supply may be utilized in the steepest climb and at the best gliding angle and so that the feed ports will not be uncovered during normal maneuvers involving moderate rolling or side slipping. The system shall also feed fuel promptly after one tank has run dry and another tank is turned on. If a mechanical pump is used, an emergency hand pump of equal capacity shall be installed and available for immediate use in case of a pump failure during takeoff. Hand pumps of suitable capacity may also be used for pumping fuel from an auxiliary tank to a main fuel tank.

**4a.606 Tank installation.** No fuel tank shall be placed closer to an engine than the remote side of a fire wall. At least one-half inch clear air space shall be allowed between the tank and the fire wall. Spaces adjacent to the surfaces of the tank shall be ventilated so that fumes cannot accumulate or reach the crew or passengers in case of leakage. If two or more tanks have their outlets interconnected they shall be considered as one tank and the air space in the tanks shall also be interconnected to prevent differences in pressure at the air vents of each tank of sufficient magnitude to cause fuel flow between tanks. Mechanical pump systems shall not feed from more than one tank at a time except by special ruling from the Administrator.

**4a.607 Tank construction.** Each fuel tank shall be provided with either a sump and drain located at the point which is lowest when the airplane is in a normal position on the ground or outlets at the bottom of the tank provided with large mesh finger strainers. If a sump is provided, the main fuel supply shall not be drawn from the bot-

tom of this sump. If no sump is provided, the system drain shall be controllable from the pilot compartment and shall act as a tank drain. Each tank shall be suitably vented from the top portion of the air space. Such air vents shall be so arranged as to minimize the possibility of stoppage by dirt or ice formation. When large fuel tanks are used, the size of the vent tubes should be proportioned so as to permit rapid changes in internal air pressure to occur and thereby prevent collapse of the tanks in a steep glide or dive. Tanks of 10 gallons or more capacity shall be provided with internal baffles, unless suitable external support is provided to resist surging.

**4a.608 Tank strength.** Fuel tanks shall be capable of withstanding an internal test pressure of  $3\frac{1}{2}$  pounds per square inch without failure or leakage. Fuel tanks of large capacity which have a maximum fuel depth greater than 2 feet shall be investigated for the pressure developed during the maximum limit acceleration with full tanks. Tanks shall be so designed, and the rivets or welds so located, as to resist vibration failures or leakage.

**4a.609 Gauge.** A satisfactory gauge shall be so installed on all airplanes as to readily indicate to a pilot or flight mechanic the quantity of fuel in each tank while in flight. When two or more tanks are closely interconnected and vented, and it is impossible to feed from each one separately, only one fuel-level gauge need be installed. If a glass gauge is used, it shall be suitably protected against breakage.

**4a.610 Lines and fittings.** All fuel lines and fittings shall be of sufficient size so that under the pressure of normal operation the flow is not less than double the normal flow required for takeoff engine power. A test for proof of compliance with this requirement shall be made. All fuel lines shall be so supported as to prevent excessive vibration and should be located so no structural loads can be applied. Bends of small radius and vertical humps in the lines shall be avoided. Copper fuel lines which have been bent shall be annealed before installation.

Parts of the fuel system attached to the engine and to the primary structure of the airplane shall be flexibly connected thereto. Flexible hose connections and fuel lines shall have metal liners or the equivalent. Fittings shall be of a type satisfactory to the Administrator.

**4a.611 Strainers.** One or more strainers of adequate size and design, incorporating a suitable sediment trap and drain, shall be provided in the fuel line between the tank and the carburetor and shall be installed in an accessible position. The screen shall be easily removable for cleaning.

**4a.612 Valves.** One or more positive and quick-acting valves that will shut off all fuel to each engine shall be within easy reach of the first pilot and the second pilot or of the flight mechanic. In the case of airplanes employing more than one source of fuel supply, suitable provision shall be made for independent feeding from each source.

**4a.613 Dump valves.** When fuel tanks are equipped with dump valves, the operating mechanism for such valves shall be within convenient reach of the first pilot and the second pilot or of the flight mechanic. Dump valves shall be so installed as to provide for safe and rapid discharge of fuel.

**4a.614 Drains.** One or more accessible drains shall be provided at the lowest point on the fuel systems to completely drain all parts of each system when the airplane is in its normal position on level ground. Such drains shall discharge clear of all parts of the airplane and shall be equipped with suitable safety locks to prevent accidental opening.

#### Miscellaneous Fuel System Requirements

**4a.615 Filler openings.** All filler openings in the fuel system shall be plainly marked with the capacity and the word "fuel". Provision shall be made to prevent any overflow from entering the wing or fuselage.

**4a.616 Prevention of ice formation.** An adequate means shall be provided for preventing the formation of ice in the engine carburetors (see also sec. 4a.550(i)).

### Lubrication Systems

4a.621 *General.* Each engine shall have an independent oil supply. The oil capacity of the system shall be at least 1 gallon for every 25 gallons of fuel but shall not be less than 1 gallon for each 75 maximum (except takeoff) rated horsepower of the engine or engines. A special ruling concerning the capacity will be made by the Administrator when oil may be transferred between engines in flight or when a suitable reserve is provided. The suitability of the lubrication system shall be demonstrated in flight tests in which engine temperature measurements are obtained. The system shall provide the engine with an ample quantity of oil at a temperature suitable for satisfactory engine operation.

4a.622 *Tank installation.* Oil tanks shall be suitably vented and shall be provided with an expansion space which cannot be inadvertently filled with oil. Such expansion space shall be at least 10 percent of the total tank volume, except that it shall in no case be less than one-half gallon.

4a.623 *Tank strength.* Oil tanks shall be capable of withstanding an internal test pressure of 5 pounds per square inch without failure or leakage. Tanks shall be so designed and the rivets or welds so located as to resist vibration failures and leakage.

4a.624 *Gauge.* A suitable means shall be provided to determine the amount of oil in the system during the filling operation.

4a.625 *Piping.* Oil piping shall have an inside diameter not less than the inside diameter of the engine inlet or outlet and shall have no splices between connections. Connections in the oil system shall be of a type satisfactory to the Administrator.

4a.626 *Drains.* One or more accessible drains shall be provided at the lowest point on the lubricating systems to drain completely all parts of each system when the airplane is in its normal position on level ground. Such drains shall discharge clear of all parts of the airplane and shall be equipped with suitable safety locks to prevent accidental opening.

4a.627 *Oil temperature.* A suitable means shall be provided for measuring the oil temperature at the engine inlet.

4a.628 *Filler openings.* All filler openings in the oil system shall be plainly marked with the capacity and the word "oil".

### Cooling Systems

4a.633 *General.* The cooling system shall be of sufficient capacity to maintain engine temperatures within safe operating limits under all conditions of flight during a period at least equal to that established by the fuel capacity of the aircraft, assuming normal engine power and speeds. Compliance with this requirement shall be demonstrated in flight tests in which engine temperature measurements are obtained under critical flight conditions including flight with one or more engines inoperative.

4a.634 *Radiators.* Radiators shall be so mounted as to reduce vibration and eliminate strains causing distortion.

4a.635 *Piping.* Piping and connections shall conform to accepted standards and shall not transmit vibration to the radiator or the structure of the aircraft.

4a.636 *Drains.* One or more accessible drains shall be provided at the lowest points on the cooling system to drain completely all parts of such system when the airplane is in its normal position on level ground. Such drains shall discharge clear of all parts of the airplane and shall be equipped with suitable safety locks to prevent accidental opening.

4a.637 *Filler openings.* All filler openings in the cooling system shall be plainly marked with the capacity of the system and the name of the proper cooling liquid.

### Powerplant Instruments, Controls, and Accessories

4a.642 *Instruments.* The engine instruments required are specified in Subpart F. The installation requirements for navigational instruments in section 4a.563 shall apply to tachometers and manifold pressure gauges. All other instruments shall be visible in flight to the pilot and copilot or to

the flight mechanic. If the manifold pressure gauges and tachometers are not visible to the flight mechanic, he shall be provided with a duplicate set of these instruments.

4a.643 *Controls.* All powerplant controls, including those of the fuel system, shall be plainly marked to show their function and method of operation. The fuel capacity and the identity of each tank shall be inscribed on or adjacent to the fuel tank selector valve controls in the pilot compartment.

4a.644 *Throttle controls.* Throttle controls shall be easily accessible to both pilots and shall be so arranged as to afford a positive and immediately responsive means of controlling all engines separately or simultaneously. Flexible throttle control systems shall be of a certificated type. A forward movement shall open the throttle.

4a.645 *Ignition switches.* Ignition switches shall be easily accessible to both pilots. A positive means for quickly shutting off all ignition of multiengine aircraft, by grouping of switches or otherwise, shall be provided.

4a.646 *Propeller pitch controls.* Separate pitch controls shall be provided for each propeller.

4a.647 *Accessories (air carrier planes).* (See sec. 4a.581.)

### Manifolding, Cowling, and Fire Wall

4a.651 *General.* All manifolds, cowling, and fire walls shall be so designed and installed as to reduce to a minimum the possibility of fire either during flight or following an accident and shall therefore comply with accepted practice in all details of installation not specified in this part.

4a.652 *Manifolds.* Exhaust manifolds shall be constructed of suitable materials, shall provide for expansion, and shall be arranged and cooled so that local hot points do not form. Gases shall be discharged clear of the cowling, airplane structure, and fuel system parts of drains. They shall not blow back on the carburetor air intake or the pilot or passengers, nor cause a glare ahead of the pilot at night. No exhaust

manifolding shall be located immediately adjacent to or under the carburetor or fuel system parts liable to leakage.

4a.653. *Air intakes.* Carburetor air intakes shall be suitably drained and shall open completely outside the cowling, unless the emergence of back-fire flames is positively prevented. The drain shall not discharge fuel in the path of possible exhaust flames.

4a.654 *Engine cowling.* All cowling around the power plant and on the engine side of the fire wall shall be made of metal and shall be so arranged that any accumulations of dirt, waste, or fuel may be observed without complete removal of the cowling. It shall fit tightly to the fire wall, but openings may be provided if the airplane surface within 15 inches thereof is protected with metal or other suitable fireproofing material. The cowling shall be completely and suitably drained in all attitudes of flight and on the ground, with separate drains provided for the parts of the fuel system liable to leakage. All such drains shall be so located as to prevent fuel or oil from dripping onto the exhaust manifold or any parts of the aircraft and from permeating any material of a cellular nature.

#### 4a.655 *Fire wall.*

(a) A fire wall shall be provided unless the engine is mounted in an isolated nacelle with no fuel tanks. Such fire bulkhead shall be constructed in either of the following approved manners:

(1) A single sheet of terneplate not less than 0.028 inch thick.

(2) A single sheet of stainless steel not less than 0.015 inch thick.

(3) Two sheets of aluminum or aluminum alloy not less than 0.02 inch thick fastened together and having between them an asbestos paper or asbestos fabric sheet at least  $\frac{1}{8}$  inch thick.

(b) The fire wall shall completely isolate the engine compartment and shall have all necessary openings fitted with close-fitting grommets or bushings. Adjacent inflammable structural members shall be protected by asbestos or an equivalent insulating mater-

ial, and provision shall be made for preventing fuel and oil from permeating it.

4a.656 *Heating systems.* Heating systems involving the passage of cabin air over or in close proximity to engine exhaust manifolds shall not be used, unless adequate precautions are incorporated in the design to prevent the introduction of carbon monoxide into the cabin or pilot compartment. They shall be constructed of suitable materials,

be adequately cooled, and be susceptible to ready disassembly for inspection.

#### Miscellaneous Powerplant Requirements

4a.661 *Materials.* Fuel, oil, and cooling systems shall be made of materials which, including their normal or inherent impurities, will not react chemically with any fuels, oils, or liquids that are likely to be placed in them.

### Subpart H—Performance

4a.671 *Performance requirements.* All airplanes shall comply with the performance requirements set forth in sections 4a.680 and 4a.682. All airplanes except those certificated in the transport category shall comply with sections 4a.672 through 4a.679, inclusive. Compliance with such performance requirements shall be shown in standard atmosphere, at all weights up to and including the standard weight (sec. 4a.37(d)) and under all loading conditions within the center of gravity range certified (sec. 4a.725): *Provided*, That demonstration of compliance with landing-speed requirements, and with those relating to takeoff time and distance, may be limited to an intermediate range of center of gravity positions if it can be shown that it is possible for the airplane to continue flight with one engine inoperative, and that passengers or other load can be easily and rapidly shifted while in flight to permit the realization, at the pilot's discretion, of a center of gravity position within the range covered by this demonstration. There shall be no flight or handling characteristics which, in the opinion of the Administrator, render the airplane unairworthy.

4a.672 *Landing speeds.* The landing speed with power off, in standard calm air at sea level, shall not exceed a value determined as follows:

(a) Airplanes certificated for passenger carrying:

(1) 65 miles per hour for airplanes of 20,000 pounds standard weight or less,

(2) 70 miles per hour for airplanes of 30,000 pounds standard weight or more, and

a linear variation with standard weight shall apply for airplanes between 20,000 and 30,000 pounds.

(b) Airplanes which are certificated for the carriage of goods only:

The above landing speed values may be increased 5 miles per hour.

4a.673 *Takeoff.* Takeoff at sea level:

(a) Within 1,000 feet for land planes;

(b) Within 60 seconds in calm air for sea-planes.

4a.674 *Climb.* The average rate of climb for the first minute after the airplane leaves the takeoff surface in accordance with section 4a.673, and the rate of steady climb at sea level with not more than maximum-except-takeoff power, shall not be less in feet per minute than:

(a) *Land planes.* Eight times the measured power-off stalling speed in miles per hour with the flaps and landing gear retracted, or 300 feet per minute, whichever is greater;

(b) *Sea planes.* Six times the measured power-off stalling speed in miles per hour with the flaps retracted, or 250 feet per minute, whichever is greater.

4a.675 *Controllability and maneuverability.* All airplanes shall be controllable and maneuverable under all power conditions and at all flying speeds between minimum flying speed and the maximum certified speed. All airplanes shall have control adequate for an average landing at minimum landing speed with power off.

4a.676 *Controllability at the stall.* With power off and with 75 percent maximum-

except-takeoff power, with flaps and landing gear in any position, the airplane shall have sufficient directional and lateral control so that when the airplane is stalled, the downward pitching motion following the stall shall occur prior to any uncontrollable roll or yaw. Any such pitching motion shall not be excessive and recovery to normal flight shall be possible by normal use of the controls after the pitching motion is unmistakably developed, without excessive loss of altitude.

**4a.677 Balance.** As used in the regulations in this part the term "balanced" refers to steady flight in calm air without exertion of control force by the pilot or automatic pilot. Lateral and directional balance is required at cruising speed which for this purpose shall be taken as 90 percent of the high speed in level flight. Longitudinal balance is required under the following flight conditions:

(a) *Power on.* In level flight, at all speeds between cruising speed and a speed of 20 percent in excess of stalling speed. In a climb, at maximum (except takeoff) horsepower and a speed 20 percent in excess of stalling speed.

(b) *Power off.* In a glide, at a speed not in excess of 140 percent of the maximum permissible landing speed or the placard speed with flaps extended, whichever is lower, under the forward center of gravity position approved with maximum authorized load and under the most forward center of gravity position approved, regardless of weight.

**4a.678 Stability.** Under all power conditions all airplanes shall be longitudinally, laterally, and directionally stable. An airplane will be considered to be longitudinally stable if, in stability tests, the amplitude of the oscillations decreases.

**4a.679 Spinning.** (Not applicable to airplanes certificated in the transport category). At any permissible combination of weight and center of gravity position obtainable with all or part of the design useful load, there shall be no excessive reversal of

control forces during any possible spinning up to 6 turns. It shall be possible promptly to recover at any point in the spinning described above by using the controls in a normal manner for that purpose and without exceeding either the limiting air speed or the limit design normal acceleration for the airplane. It shall not be possible to obtain uncontrollable spins by means of any possible use of the controls: *Provided*, That compliance with the foregoing requirements with respect to spinning shall not be required for those airplanes:

(a) Permanently placarded "intentional spinning prohibited"; or

(b) Demonstrated to the satisfaction of the Administrator to be characteristically incapable of spinning.

**4a.680 Flutter and vibration.** Wings, tail surfaces, control surfaces, and primary structural parts shall be free from flutter or objectionable vibration in all normal attitudes or conditions of flight between the minimum flying speed and the maximum indicated air speed attained in official flight tests (see sec. 4a.708).

**4a.681-T Flutter and vibration.** All parts of transport category airplanes shall be free from flutter or excessive vibration under all speed and power conditions appropriate to the operation of the airplane during take-off, climb, level flight, and landing, and during glide at speeds up to the maximum indicated air speed attained during official flight tests (see sec. 4a.708). There shall be no appreciable buffeting for any flap position at any speed in excess of 10 miles per hour above stalling speed for such position nor shall buffeting at lower speeds be so violent as to interfere with the pilot's control of the airplane or cause discomfort to its occupants.

**4a.682 Ground and water characteristics.** Landplanes shall be maneuverable on the ground and shall be free from dangerous ground looping tendencies and objectionable taxiing characteristics. The seaworthiness and handling characteristics of seaplanes and amphibians shall be demonstrated by

tests deemed appropriate by the Administrator.

*Cross Reference:* For water stability requirements, see sec. 4a.491.

### Modified Performance Requirements

**4a.687 Modified performance requirements for multiengine airplanes not certificated in the transport category.** The weight of any multiengine airplane manufactured pursuant to a type certificate issued prior to January 1, 1941, may be increased beyond the values corresponding to the landing speed specified in section 4a.672 and takeoff requirements of section 4a.673, subject to the following conditions:

(a) The increased weight shall be known as the provisional weight (sec. 4a.37 (e)). The standard weight (sec. 4a.37(d)) shall be the maximum permissible weight for landing. The provisional weight shall be the maximum permissible weight for takeoff.

(b) Compliance with all the airworthiness requirements except landing speed and takeoff is required at the provisional weight, except that the provisional weight may exceed the design weight on which the structural loads for the landing conditions are based by an amount not greater than 15 percent: *Provided*, That the airplane is shown to be capable of safely withstanding the ground or water shock loads incident to taking off at the provisional weight.

(c) The airplane shall be provided with suitable means for the rapid and safe discharge of a quantity of fuel sufficient to reduce its weight from the provisional weight to the standard weight.

(d) In no case shall the provisional weight exceed a value corresponding to a landing speed of 5 miles per hour in excess of that specified in section 4a.672, a takeoff distance of 1,500 feet in the case of landplanes, or a takeoff time of 60 seconds in the case of seaplanes; nor shall any provisional weight authorized in respect to any type of airplane after January 1, 1945, exceed the value corresponding to a rate of climb of at least 180 feet per minute at an altitude of 5,000 feet

with the critical engine inoperative, its propeller windmilling with the propeller control in a position which would allow the engine (if operating normally and within approved limits) to develop at least 50 percent of maximum-except-takeoff engine speed, all other engines operating at the takeoff power available at such altitude, the landing gear retracted, center of gravity in the most unfavorable position permitted for takeoff, and the flaps in the takeoff position.

### Performance Tests

**4a.701 General.** Compliance with the performance requirements in sections 4a.671 through 4a.687 shall be demonstrated by means of suitable flight tests of the type airplane. Computations may be used to estimate the effects of minor changes. Additional information concerning the performance characteristics of air carrier airplanes is specified in section 4a.717. Such characteristics shall be determined by direct flight testing, or by methods combining basic flight tests and calculations. All performance characteristics shall be corrected to standard atmospheric conditions and zero wind. Methods of performance calculation and correction employed shall be subject to the approval of the Administrator.

#### 4a.702 Flight test pilot.

(a) The applicant shall provide a person holding an appropriate commercial pilot certificate to make the flight tests, but a designated inspector of the Administrator may pilot the airplane during such parts of the tests as he may deem advisable.

(b) In the event that the applicant's test pilot is unable or unwilling to conduct any of the required flight tests, the tests shall be discontinued until the applicant furnishes a competent pilot.

**4a.703 Parachutes.** Parachutes shall be worn by members of the crew during the flight tests.

**4a.704 Reports.** The applicant shall submit to the inspector of the Administrator a report covering all computations and tests required in connection with calibration of flight instruments and correction of tests re-

sults to standard atmospheric conditions. The inspector will conduct any flight tests which appear to him to be necessary in order to check the calibration and correction report or to determine the airworthiness of the airplane.

**4a.705 Loading conditions.** The loading conditions used in performance tests shall be such as to cover the range of loads and center of gravity positions for which the airplane is to be certificated.

**4a.706 Use of ballast.** Ballast may be used to enable airplanes to comply with the flight requirements as to longitudinal stability, balance, and landing in accordance with the following provisions:

(a) Ballast shall not be used for this purpose in airplanes having a gross weight of less than 5,000 pounds nor in airplanes with a total seating capacity of less than seven persons.

(b) The place or places for carrying ballast shall be properly designed and installed and plainly marked.

(c) The loading schedule which will accompany each certificate issued for an airplane requiring special loading of this type shall be conspicuously posted in either the pilot compartment or in or adjacent to the ballast compartments, and strict compliance therewith will be required of the airplane operator.

**4a.707 Fuel to be carried.** When low fuel adversely affects balance or stability, the airplane shall be so tested as to simulate the condition existing when the amount of fuel on board does not exceed 1 gallon for every 12 maximum (except takeoff) horsepower of the engine or engines installed thereon. When the engine is limited to a lower power, the latter shall be used in computing low fuel.

**4a.708 Maximum air speed.** The flight tests shall include steady flight in relatively smooth air at the design gliding speed ( $V_g$ ) for which compliance with the structural loading requirements (secs. 4a.72 through 4a.99) has been proved, except that they need not involve speeds in excess of  $1.33 V_L$  (sec. 4a.40(c)): *Provided*, That the operation lim-

its are correspondingly fixed (see sec. 4a.726). When high-lift devices having nonautomatic operation are employed, the tests shall also include steady flight at the design flap speed  $V_f$  (sec. 4a.40(f)), except that they need not involve speeds in excess of  $2 V_{sf}$  (see sec. 4a.40(e)). In cases where the high-lift devices are automatically operated, the tests shall cover the range of speeds within which the devices are operative.

**4a.709 One-engine-inoperative performance.** Multiengine airplanes shall be flight tested at such altitudes and weights as are necessary, in the opinion of the Administrator, to prepare accurate data to show climbing performance within the range of weight for which certification is sought, with the critical engine inoperative and each other engine operating at not more than maximum-except-takeoff power. Such data when approved by the Administrator shall be kept in the airplane at all times during flight in a place conveniently accessible to the pilot.

**4a.710 Air-speed indicator calibration.** In accordance with section 4a.559, the air-speed indicator of the type airplane shall be calibrated in flight. The method of calibration used shall be subject to the approval of the Administrator.

**4a.711 Check of fuel system.** The operation of the fuel system shall be checked in flight to determine its effectiveness under low fuel conditions and after changing from one supply tank to another. (See sec. 4a.605.) For such test low fuel is defined as approximately 15 minutes supply in each tank tested, at the maximum-except-take-off power certified.

### Air Carrier Aircraft Performance Characteristics

**4a.717 Performance characteristics of air carrier aircraft.** No air carrier shall operate aircraft in scheduled air transportation unless data shall have been submitted to and approved by the Administrator, covering the determination of such performance characteristics, in addition to those specified in sections 4a.671-4a.711, as are, in the opinion of the Administrator, necessary to determine



the ability of such aircraft to safely perform the type of operation which the air carrier proposes to conduct. The method used for the determination of such ability shall be subject to the approval of the Administrator.

### Operation Limitations

**4a.723 Weight.** Non-air carrier airplanes may be certificated at a maximum authorized weight which is not sufficient to permit carrying simultaneously the full fuel and full pay load, provided that such weight shall be sufficient to provide a gasoline load of at least 0.15 gallon per certified maximum (except takeoff) horsepower, with all seats occupied and with sufficient oil for this amount of fuel.

4a.723-1 *New production aircraft—empty weight and c.g. determination (FAA policies which apply to § 4a.723).* See § 4a.725-1.

[Supp. 5, 14 F.R. 6282, Oct. 15, 1949]

**4a.724 Provisional weight (air carrier airplanes).** (See sec. 4a.687.)

**4a.725 Center of gravity limitations.** The maximum variation in the location of the center of gravity for which the airplane is certificated to be airworthy shall be established. Means shall be provided, when necessary in the opinion of the Administrator, by which the operator is suitably informed of the permissible loading conditions which result in a center of gravity within the certified range.

4a.725-1 *New production aircraft—empty weight and c.g. determination (FAA policies which apply to sec. 4a.725)*

(a) *Purpose.* The purpose of this section is to provide a procedure which will permit manufacturers of new aircraft, as described in paragraph (b) of this section to establish an average empty weight and empty c.g. for such aircraft, thus avoiding the necessity of weighing each aircraft.

(b) *Coverage.* Aircraft to which the procedure outlined in this section may be applied are those which are newly manufactured in accordance with requirements contained in this part and Part 3 of this subchapter (except transport category aircraft), and which are produced under the terms of a production certificate.

(c) *Procedure.* Manufacturers producing aircraft in accordance with the requirements prescribed in paragraph (b) of this section who are interested in establishing an average empty weight and empty c.g. in lieu of actually weighing each aircraft, should prepare and forward through the local Aviation Safety Agent to the Chief, Manufacturing Inspection Branch, for coordination and approval, a detailed proposal regarding the procedure to be followed in establishing the system outlined in this section. Any proposal submitted by a manufacturer which can be shown to achieve the objective of the present requirements applying to weight and balance control; i.e., an accurate determination of average empty weight and empty c.g., will be considered acceptable.

(d) *Example.* The following example outlines an acceptable method for effecting this system:

(1) Actually weigh and determine empty c.g. of 5 to 10 aircraft of a particular model, which have comparatively identical equipment installed, to determine the average weight and c.g.

(2) Weigh an individual aircraft at regular intervals; e.g., each tenth aircraft, as circumstances and conditions may warrant, for the purpose of determining continued accuracy of the initial empty weight and c.g. established.

(3) When the spot checking, as prescribed in subparagraph (2) of this paragraph, indicates a variation in weight in excess of 1 percent of the initially established empty weight and/or a variation in the empty weight c.g. in excess of ½ percent of the MAC, a new average should be established in accordance with subparagraph (1) of this paragraph.

(4) Inasmuch as a weight and balance report is required in connection with each aircraft presented for certification, these reports may be computed for aircraft which are not actually weighed. Such reports should be marked "computed" for those aircraft which are not actually weighed, and other reports will be marked "actual."

[Supp. 5, 14 F.R. 6282, Oct. 15, 1949]

**4a.726 Air-speed limitations.** Maximum operation limitations will be incorporated in

the aircraft certificate and will specify the indicated air speeds which shall not be exceeded in level and climbing flight (sec. 4a.40(c)), in gliding and diving flight, and with flaps extended. The values in gliding flight and with flaps extended will be 10 percent less than the corresponding maximum air speeds attained in flight tests in accordance with sec. 4a.708.

**4a.727 Powerplant limitations.** Maximum operational limitations will be incorporated in the aircraft certificate and will specify powerplant outputs on takeoff (sec. 4a.187), in climbing flight, and for all operations other than takeoff and climbing flight (sec. 4a.38(b)). The output, except for takeoff, shall not exceed that corresponding to the maximum (except takeoff) rating of the engine installed. For the above purposes no specified output will be in excess of that corresponding to the limits imposed by either the pertinent engine or propeller certification (see secs. 4a.25 and 4a.26).

### Transport Category Airplane Performance Requirements

**4a.737-T Performance requirements for transport category airplanes.** The following requirements shall apply in place of sections 4a.672-4a.679.

**4a.738-T Minimum requirements for certification.**

(a) An airplane may be certificated under the provisions of section 4a.737-T upon there having been established, in accordance with the terms of that section:

(1) A maximum takeoff weight at sea level;

(2) A maximum landing weight at sea level;

(3) A maximum one-engine-inoperative operating altitude (as defined in sec. 4a.741-T), which shall be at least 5,000 feet at a weight equal to the maximum sea level takeoff weight;

(4) Takeoff characteristics at maximum sea level takeoff weight, and landing characteristics at maximum sea level landing

weight, in accordance with the provisions of sections 4a.747-T and 4a.750-T, and

(5) Compliance with the requirements of all other applicable parts of the regulations of this part.

(b) If a certificate is issued under these conditions, it may be amended from time to time to include landing and takeoff weights over an increased range of altitudes and other pertinent performance data, including additional landing and takeoff characteristics obtained in accordance with the provisions of sections 4a.747-T and 4a.750-T.

### Definitions

**4a.739-T Stalling speeds.** In sections 4a.737-T through 4a.760-T:

(a)  $V_{s_0}$  denotes the true indicated stalling speed of the airplane in miles per hour with engines idling, throttles closed, propellers in low pitch, landing gear extended, flaps in the "landing position", as defined in section 4a.740-T, cowl flaps closed, center of gravity in the most unfavorable position within the allowable landing range, and the weight of the airplane equal to the weight in connection with which  $V_{s_0}$  is being used as a factor to determine a required performance.

(b)  $V_{s_1}$  denotes the true indicated stalling speed in miles per hour with engines idling, throttles closed, propellers in low pitch, and with the airplane in all other respects (flaps, landing gear, etc.) in the condition existing in the particular test in connection with which  $V_{s_1}$  is being used.

**4a.740-T Flap positions.** The flap positions denoted respectively as the landing position, approach position, and takeoff position are those provided for in section 4a.464-T, and may be made variable with weight and altitude in accordance with that section.

**4a.741-T Maximum one-engine-inoperative operating altitude.** Maximum one-engine-inoperative operating altitude (to be determined in complying with section 4a.709) shall be the altitude in standard air at which the steady rate of climb in feet per minute is  $0.02 V_{s_0}^2$  with the critical engine inopera-

tive, its propeller stopped, all other engines operating at the maximum-except-takeoff power available at such altitude, the landing gear retracted, and the flaps in the most favorable position.

### Weights

**4a.742-T Weights.** The maximum takeoff weight and maximum landing weight shall be established by the applicant and may be made variable with altitude. The maximum takeoff weight for any altitude shall not exceed the maximum design weight used in the structural loading conditions for flight loads (secs. 4a.72-4a.99), and shall not exceed the design weight used in the structural loading conditions for ground or water loads (secs. 4a.147-4a.156 and secs. 4a.161-4a.177, respectively) by a ratio of more than 1.15. The maximum landing weight for any altitude shall not exceed the design weight used in the structural loading conditions for ground or water loads.

**4a.743-T Fuel dumping provisions.** (a) If the maximum take-off weight for any altitude exceeds the maximum landing weight for the same altitude, adequate provision shall be made, in accordance with Subpart G, for the rapid and safe dumping during flight of a quantity of fuel sufficient to reduce the weight of the airplane from such maximum take-off weight to such maximum landing weight. Compliance with this section shall be shown by dumping suitable colored fluids and fuel in flight tests in the following conditions:

- (1) Level flight at a speed of  $2.0 V_{s1}$ ,
- (2) Climb at a speed of  $1.4 V_{s1}$  with 75 percent of maximum-except-take-off power,
- (3) Glide with power off at a speed of  $1.4 V_{s1}$ .

(b) In conditions (a) (1) and (2), the time required to dump the necessary amount of fuel shall not exceed 10 minutes. During such tests, the dumped fluid shall not come in contact with any portion of the aircraft or adversely affect its control, nor shall any fumes from such fluid enter any portion of the aircraft.

### Performance Requirements and Determinations

**4a.744-T Required performance and performance determinations.** Performance data shall be corrected to standard atmosphere and still air where such corrections are applicable. Performance data may be determined by calculation from basic flight tests if the results of such calculation are substantially equal in accuracy to the results of direct tests.

**4a.745-T Stalling speed requirements.**

(a)  $V_{s0}$  at maximum landing weight shall not exceed 80 miles per hour.

(b)  $V_{s1}$  at maximum landing weight, flaps in the approach position, landing gear extended, and center of gravity in the most unfavorable position permitted for landing, shall not exceed 85 miles per hour.

**4a.746-T Climb requirements.** In the climb tests required by this section, the engine cowl flaps, or other means of controlling the engine cooling air supply, shall be in a position which will provide adequate cooling with maximum-except-take-off power at best climbing speed under standard atmospheric conditions.

(a) **Flaps in landing position.** The steady rate of climb in feet per minute, at any altitude within the range for which landing weight is to be specified in the certificate, with the weight equal to maximum landing weight for that altitude, all engines operating at the take-off power available at such altitude, landing gear extended, center of gravity in the most unfavorable position permitted for landing, and flaps in the landing position, shall be at least  $0.07 V_{s0}^2$ .

(b) **Flaps in approach position.** The steady rate of climb in feet per minute, at any altitude within the range for which landing weight is to be specified in the certificate, with the weight equal to maximum landing weight for that altitude, the critical engine inoperative, its propeller stopped, all other engines operating at the take-off power available at such altitude, the landing gear retracted, center of gravity in the most unfavorable position permitted for

landing, and the flaps in the approach position, shall be at least  $0.04 V_{s_0}^2$ .

(c) *Flaps in takeoff position.* The steady rate of climb in feet per minute, at any altitude within the range for which takeoff weight is to be specified in the certificate, with the weight equal to maximum takeoff weight for that altitude, the speed equal to the minimum takeoff climb speed permitted in section 4a.748-T(b), the critical engine inoperative, its propeller windmilling with the propeller control in a position which would allow the engine (if operating normally and within approved limits) to develop at least 50 percent of maximum-except-take-off engine speed, all other engines operating at the takeoff power available at such altitude, the landing gear retracted, center of gravity in the most unfavorable position permitted for takeoff, and the flaps in the takeoff position, shall be at least  $0.035 V_{s_1}^2$ .

4a.747-T *Takeoff determination.* The takeoff data set forth in sections 4a.748-T and 4a.749-T shall be determined over such range of weights and altitudes as the applicant may desire, with a constant takeoff flap position for a particular weight and altitude, and with the operating engines at not more than the takeoff power available at the particular altitude. These data shall be based on a level takeoff surface with zero wind.

4a.748-T *Speeds.*

(a) *Critical-engine-failure speed*, denoted by  $V_1$ , is a true indicated air speed, chosen by the applicant, but in any case not less than the minimum speed at which the controllability is adequate to proceed safely with the takeoff, using normal piloting skill, when the critical engine is suddenly made inoperative.

(b) *Minimum takeoff climb speed*, denoted by  $V_2$ , is a true indicated air speed chosen by the applicant, which shall permit the rate of climb required in section 4a.746-T(c) but which shall not be less than  $1.20 V_{s_1}$  for two-engined airplanes, or  $1.15 V_{s_1}$  for airplanes having more than two engines, or less than 1.10 times the minimum speed at which the airplane is fully controllable in flight using

normal piloting skill when the critical engine is suddenly made inoperative.

4a.749-T *Takeoff path.* The lengths and slopes of segments of the takeoff path, and the location of critical points on the complete path shall be determined in accordance with the following conditions and assumptions. The location of the points defined in paragraphs (a) to (e) of this section shall be expressed in terms of the horizontal and vertical distances from the starting point.

(a) *Starting point.* The point from which a standing start is made with all engines operating.

(b) *Critical-engine-failure point.* The point at which the airplane attains speed  $V_1$  (critical-engine-failure speed) when accelerated from point (a) with all engines operating.

(c) *Accelerate-and-stop point.* The point on the takeoff surface at which the airplane can be brought safely to a stop if all engines are cut at point (b).

(d) *Start-of-climb point.* The point on or just clear of the takeoff surface at which the airplane attains speed  $V_2$  (takeoff climb speed) when the critical engine is made inoperative with its propeller windmilling in low pitch at point (b).

The takeoff acceleration segment, (a) to (d) shall be determined by making a continuous run up to speed  $V_2$  with the critical engine cut at point (b).

(e) *Retraction-completion point.* The point at which landing gear retraction is completed when retraction is initiated not earlier than point (d).

(1) The initial climb segment, (d) to (e), shall be assumed to correspond to the rate of climb at speed  $V_2$  with landing gear extended and windmilling propeller in low pitch.

(2) The second climb segment, beginning at point (e), shall be assumed to correspond to the rate of climb at speed  $V_2$  with landing gear retracted and windmilling propeller in high pitch, as defined in section 4a.746-T (c). This segment may continue indefinitely or may end at point (g) in accordance with paragraph (g) of this section.

(f) *50-foot height point.* The point at which the airplane attains a height of 50 feet (above the takeoff surface) along the takeoff flight path defined herein.

(g) *Feathering-completion point.* The point where feathering or stopping of the inoperative propeller is completed, if the applicant desires to include this step in the takeoff determination. It shall be assumed that the decision to feather or stop is made not earlier than the instant of attaining point (f).

(1) In the event that it is desired to include propeller feathering or stopping in the takeoff path, the final climb segment, beginning at point (g), shall be assumed to correspond to the rate of climb at speed  $V_2$  with landing gear retracted and the propeller of the inoperative engine feathered or stopped.

4a.749a-T *Temperature accountability.* Operating correction factors for takeoff weight and takeoff distance shall be determined to account for temperatures above and below standard, and when approved by the Administrator shall be included in the *Airplane Flight Manual*. These factors shall be obtained as set forth in paragraphs (a) and (b) of this section.

(a) For any specific airplane type, the average full temperature accountability shall be computed for the range of weights of the airplane, altitudes above sea level, and ambient temperatures required by the expected operating conditions. Account shall be taken of the temperature effect on both the aerodynamic characteristics of the airplane and on the engine power. The full temperature accountability shall be expressed per degree of temperature in terms of a weight correction, a takeoff distance correction, and a change, if any, in the critical engine failure speed  $V_1$ .

(b) The operating correction factors for the airplane weight and takeoff distance shall be at least one-half of the full accountability values. The value of  $V_1$  shall be further corrected by the average amount necessary to assure that the airplane can stop within the runway length at the ambient

temperature; except that the corrected value of  $V_1$  shall not be less than a minimum at which the airplane can be controlled with the critical engine inoperative.

4a.750-T *Landing determination.* The horizontal distance required to land and come to a complete stop from a point at a height of 50 feet above the landing surface shall be determined for such range of weights and altitudes as the applicant may desire. In making this determination:

(a) Immediately prior to reaching the 50-foot altitude, a steady gliding approach shall have been maintained, with a true indicated air speed of at least  $1.3 V_{s_0}$ .

(b) The nose of the airplane shall not be depressed, nor the power increased, after reaching the 50-foot altitude. At all times during and immediately prior to the landing, the flaps shall be in the landing position, except that after the airplane is on the landing surface and the true indicated air speed has been reduced to not more than  $0.9 V_{s_0}$  the flap position may be changed.

(c) The operating pressures on the braking system shall not be in excess of those approved by the manufacturer of the brakes.

(d) The brakes shall not be used in such manner as to produce excessive wear of brakes or tires.

(e) The landing shall be made in such manner that there is no excessive vertical acceleration, no tendency to bounce, nose over, porpoise, ground loop, or water loop, and in such manner that its reproduction shall not require any exceptional degree of skill on the part of the pilot, or exceptionally favorable conditions. If this last condition (with respect to exceptional skill or favorable conditions) is not met, the distance to be determined shall be that considered to correspond to a piloting technique normally usable.

#### Flight Characteristics

4a.751-T *Flight characteristics.* There shall be no flight characteristic which makes the airplane unairworthy. The airplane shall also meet the following requirements under all critical loading conditions within

the range of center of gravity, and, except as provided in section 4a.753-T(d) at the maximum weight for which certification is sought.

**4a.752-T Controllability and maneuverability.** The airplane shall be controllable and maneuverable during takeoff, climb, level flight, glide, and landing, and it shall be possible to make a smooth transition from one flight condition to another, without requiring an exceptional degree of skill, alertness, or strength on the part of the pilot, under all conditions of operation probable for the type, including those conditions normally encountered in the event of sudden failure of any engine. It shall be possible, with power off, with flaps either retracted or in the landing position, with the center of gravity in the most unfavorable location within the certificated range, and with the airplane trimmed for a speed of  $1.4 V_{s1}$ , to change the flap position to the opposite extreme, to make a sudden application of take-off power on all engines, or to change the speed to any value between  $1.10 V_{s1}$ , and  $1.70 V_{s1}$ , without requiring a change in the trim control or the exertion of more control force than can readily be applied with one hand for a short period. It shall not be necessary to use exceptional piloting skill in order to prevent loss of altitude when flap retraction from any position is initiated during steady horizontal flight at  $1.1 V_{s1}$ , with simultaneous application of not more than maximum-except-takeoff power.

**4a.753-T Trim.** The means used for trimming the airplane shall be such that after being trimmed and without further pressure upon or movement of either the primary control or its corresponding trim control by the pilot or the automatic pilot, the airplane will maintain:

(a) Lateral and directional trim under all conditions of operation consistent with the intended use of the airplane, including operation at any speed from best rate-of-climb speed to high speed and operation in which there is greatest lateral variation in the distribution of the useful load;

(b) Longitudinal trim, under the following conditions:

(1) During climb at the best rate-of-climb speed with maximum-except-take-off power.

(2) During a glide with power off at a speed not in excess of  $1.4 V_{s1}$ , and

(3) During level flight at any speed from 90 percent of high speed to the sum of  $V_{s1}$  and 20 percent of the difference between high speed and  $V_{s1}$ ;

(c) Rectilinear climbing flight with the critical engine inoperative, each other engine operating at maximum-except-take-off power and the best rate-of-climb speed under such conditions;

(d) Rectilinear flight with any two engines inoperative and each other engine operating at maximum-except-take-off power under the following conditions:

(1) With the weight of the airplane not more than that at which there is a speed range in level flight of not less than 10 miles per hour;

(2) With the speed of the airplane not more than the high speed obtained under the conditions specified in subparagraph (1) of this paragraph less 10 miles per hour.

**4a.754-T Stability.** The airplane shall be longitudinally, directionally, and laterally stable in accordance with the following provisions. Suitable stability and control "feel" may be required in other conditions normally encountered in service if flight tests show such stability to be necessary for safe operation.

**4a.755-T Static longitudinal stability.** In the flight conditions described in section 4a.756-T,

(a) At any speed which can be obtained without excessive control force and which is more than 10 miles per hour above or below the specified trim speed, but not greater than the appropriate maximum permissible speed or less than the minimum speed in steady unstalled flight, the characteristics of the elevator control forces and friction shall be such that:

(1) A pull is required to maintain speeds below the specified trim speed and a push to

maintain speeds above the specified trim speed.

(2) The control will, when unrestrained by the pilot, move continuously toward its original trim position.

(b) Where a stable slope of the stick force versus speed curve is specified, any decrease in speed below trim speed shall require an increase in the steady pull on the elevator control and any increase in speed above trim speed shall require an increase in the steady push on the control. Such slope shall be between such limits that any substantial change in speed is clearly perceptible to the pilot through a resulting change in stick force, and that the stick force required to produce necessary changes in speed does not reach excessive values.

4a.756-T *Specific stability conditions—*

(a) *Landing.* With flaps in the sea level landing position, the landing gear extended, maximum sea level landing weight, the airplane trimmed at  $1.4 V_{s_1}$  and throttles closed, the stick force curve shall have a stable slope at all speeds between  $1.1 V_{s_1}$  and  $1.8 V_{s_1}$ .

(b) *Approach.* With flaps in sea level approach position, landing gear retracted, maximum sea level landing weight, the airplane trimmed at  $1.4 V_{s_1}$  and with power sufficient to maintain level flight at this speed, the stick force curve shall have a stable slope at all speeds between  $1.1 V_{s_1}$  and  $1.8 V_{s_1}$ .

(c) *Climb.* With flaps retracted, landing gear retracted, maximum sea level take-off weight, 75 percent of maximum-except-take-off power, and with the airplane trimmed at  $1.4 V_{s_1}$ , the stick force curve shall have a stable slope at all speeds between  $1.2 V_{s_1}$  and  $1.6 V_{s_1}$ .

(d) *Cruising.* With flaps retracted, maximum sea level takeoff weight, 75 percent of maximum-except-take-off power, and with the airplane trimmed for level flight, the stick force curve shall have a stable slope at all speeds obtainable with reasonable stick forces between:

(1)  $1.2 V_{s_1}$  and the maximum permissible speed, when the landing gear is retracted;

(2)  $1.2 V_{s_1}$  and the level flight speed, when the landing gear is extended.

4a.757-T *Dynamic longitudinal stability.* The airplane shall not be dynamically unstable longitudinally, as shown by the damping of the normal long period oscillation, under any flight condition that is likely to be maintained for more than 10 minutes in ordinary service. Compliance with this requirement shall be demonstrated under at least the following conditions:

(a) During level flight with 75 percent of maximum-except-take-off power.

(b) During a climb with 75 percent of maximum-except-take-off power at a speed equal to 75 percent of that obtained in paragraph (a) of this section.

Any short period oscillation occurring between stalling speed and maximum permissible speed shall be heavily damped with the primary controls in a fixed position.

4a.758-T *Directional and lateral static stability.* The static directional stability, as shown by the tendency to recover from a skid with rudder free, shall be positive for all flap positions and symmetrical power conditions, and for all speeds from  $1.2 V_{s_1}$  up to the maximum permissible speed. The static lateral stability as shown by the tendency to raise the low wing in a sideslip, shall be positive within the same limits.

4a.759-T *Stalling.* With power off, and with that power necessary to maintain level flight with flaps in approach position at a speed of  $1.6 V_{s_1}$ , maximum landing weight, flaps and landing gear in any position, and center of gravity in the least favorable position for recovery, it shall be possible to produce and to correct roll and yaw by unreversed use of the aileron and rudder controls up to the time when the airplane pitches in the maneuver described below. During the pitching and recovery portions of the maneuver it shall be possible to prevent appreciable rolling or yawing by normal use of the controls.

In demonstrating this quality, the order of events shall be:

(a) With trim controls adjusted for straight flight at a speed of  $1.4 V_{s_1}$ , reduce speed by means of the elevator control until the speed is steady at slightly above stalling speed; then

(b) Pull elevator control back at a normal rate until a stall is produced as evidenced by an uncontrollable downward pitching motion of the airplane, or until the control reaches the stop. Normal use of the elevator control for recovery may be made after such pitching motion is unmistakably developed.

In any case, the airplane shall not pitch excessively before recovery is completed.

The airplane shall be recoverable without difficulty or the use of power from the inoperative engine when it is stalled with the critical engine inoperative and the remaining engines operating at 75 percent of maximum-except-take-off power.

### Operating Manual

4a.760-T *Airplane operating manual.* There shall be furnished with each airplane a copy of a manual which shall contain such information regarding the operation of the airplane as the Administrator may require, including, but not limited, to the following:

(a) All performance data secured under sections 4a.741-T through 4a.750-T together with any pertinent descriptions of the conditions, air speeds, etc., under which such data were determined.

(b) Adequate instructions for the use and adjustment of the flap controls under section 4a.464-T.

(c) The indicated air speeds corresponding to those determined in section 4a.748-T, together with pertinent discussion of procedures to be followed if the critical engine becomes inoperative on takeoff.

(d) A discussion of any significant or unusual flying or ground-handling characteristics, knowledge of which would be useful to a pilot not previously having flown the airplane.

4a.760-T-1 *Airplane flight manuals (FAA policies which apply to sec. 4a.760-T).* See section 4b.911-1 of this subchapter (14 F.R. 4176, July 16, 1949).

[Supp. 2, 13 F.R. 4182, July 22, 1948; 14 F.R. 37, Jan. 5, 1959]

## Subpart I—Miscellaneous Requirements

4a.770 *Identification plate.* A fireproof identification plate shall be securely attached to the structure in an accessible location where it will not likely be defaced during normal service. The identification plate shall not be placed in a location where it might be expected to be destroyed or lost in the event of an accident. The identification plate shall contain the identification data required by section 2.36 of this subchapter.

4a.771 *Standard weights.* In computing weights the following standard values shall be used:

Gasoline .....	6 pounds per gallon.
Lubricating oil.....	7.5 pounds per gallon.
Crew and passengers...	170 pounds per person, unless otherwise specified by the Administrator.
Parachutes .....	20 pounds each.

4a.772 *Leveling means.* Adequate means shall be provided for easily determining when the aircraft is in a level position.



# Appendix A

## Tables and Figures

1. Condition.....	I	I	III	IV	V	VI
2. Reference.....	§ 4a.79	§ 4a.81	§ 4a.82	§ 4a.83	§ 4a.84	§ 4a.85
3. Design Speed (see § 4a.73).....	$V_L$	$V_L$	$V_L$	$V_L$	$V_L$	$V_L$
4. Gust Velocity, $U$ , f. p. s. (1) (2).....	+30	-30	+15	-15	-----	-----
5. $\Delta n$ (a) Gust (2).....	§ 4a.76	§ 4a.76	§ 4a.76	§ 4a.76	-0.5 $\Delta n_{IV}$	-----
6. $\Delta n$ (b) Maneuvering.....	Fig. 4a-3	-----	Fig. 4a-3	-----	-0.25 $\Delta n_{IV}$	-----
6. Limit Load Factor, $n$ . When item 5 gives two values of $\Delta n$ , use larger.....	1+ $\Delta n_I$	1+ $\Delta n_{II}$	1+ $\Delta n_{III}$	1+ $\Delta n_{IV}$	-1+ $\Delta n_V$	-----
7. Minimum Value of $n$ .....	2.50	None	2.50	None	-1.5	None
8. Minimum Yield Factor of Safety, $f_y$ .....	1.0	1.0	1.0	1.0	1.0	1.0
9. Minimum Ultimate Factor of Safety, $f_u$ .....	1.5	1.5	1.5	1.5	1.5	1.5

(1) Feet per second.

(2) + means upward, - means downward.

(\*) May be limited by maximum dynamic lift coefficient obtainable under sudden changes of angle of attack.

TABLE 4a-1. Symmetrical Flight Conditions (Flaps Retracted).

1. Condition.....	VII	VIII	IX
2. Reference.....	§ 4a.87	§ 4a.88	§ 4a.89
3. Design Speed (see § 4a.73).....	$V_f$	$V_f$	$V_f$
4. Gust Velocity, $U$ , f. p. s. (1) (2).....	+15	-15	-----
5. $\Delta n$ (2).....	§ 4a.76	§ 4a.76	-----
6. Limit Load Factor, $n$ .....	1+ $\Delta n_{VII}$	1+ $\Delta n_{VIII}$	-----
7. Minimum Value of $n$ .....	2.00	None	None
8. Minimum Yield Factor of Safety, $f_y$ .....	1.0	1.0	1.0
9. Minimum Ultimate Factor of Safety, $f_u$ .....	1.5	1.5	1.5

(1) Feet per second.

(2) + means upward, - means downward.

(\*) May be limited by maximum dynamic lift coefficient obtainable under sudden changes of angle of attack.

TABLE 4a-2. Symmetrical Flight Conditions (Flaps Extended).

1. Condition.....	Balancing	Maneuvering	Damping	Tab effects
2. Reference.....	§ 4a.116	§ 4a.117	§ 4a.118	§ 4a.119
3. Design Speed (see § 4a.73).....	-----	$V_p$	-----	$V_L$
4. Force Coefficient, $C_N$ .....	-----	{ -0.55 (down) +0.35 (up) }	-----	-----
5. Average Limit Pressure, p. s. f. (1).....	-----	$C_N q_p$ (2)	-----	-----
6. Chord Distribution.....	Fig. 4a-4	Fig. 4a-5	Fig. 4a-6	Fig. 4a-5 (3)
7. Span Distribution.....	Constant $C_N$	Constant $C_N$	Constant $C_N$	Constant $C_N$ (3)
8. Minimum Average Limit Pressure, p. s. f. (1).....	-----	15	-----	-----
9. Special Requirements.....	None	None	None	None
10. Minimum Yield Factor of Safety, $f_y$ .....	1.0	1.0	1.0	1.0
11. Minimum Ultimate Factor of Safety, $f_u$ .....	1.5	1.5	1.5	1.5

(1) Over entire horizontal tail.

(2)  $q_p$  is the dynamic pressure corresponding to  $V_p$ , see § 4a.42.

(3) Refers to main surface, disregarding tab; uniform pressure distribution may be assumed over tab.

TABLE 4a-3. Loading Conditions for Horizontal Tail Surfaces.

1. Condition.....	Maneuvering	Damping	Gust	Tab effects
2. Reference.....	§ 4a.120	§ 4a.121	§ 4a.122	§ 4a.123
3. Design Speed (see § 4a.73).....	$V_s^{(1)}$		$V_L$	$V_L^{(1)}$
4. $C_N$ or Gust.....	$C_N=0.45$		$U=30 f. p. s.$	
5. Average Limit Pressure, p. s. f. (1).....	$C_N q_p^{(2)}$		§ 4a.122(b)	
6. Chord Distribution.....	Fig. 4a-5	Fig. 4a-6	Fig. 4a-6 <sup>(3)</sup>	Fig. 4a-5 <sup>(4)</sup>
7. Span Distribution.....	Constant $C_N$	Constant $C_N$	Constant $C_N$	Constant $C_N^{(5)}$
8. Minimum Average Limit Pressure, p. s. f. (1).....	12			
9. Special Requirements.....	4a.120 (b)	None	None	None
10. Minimum Yield Factor of Safety, $J_y$ .....	1.0	1.0	1.0	1.0
11. Minimum Ultimate Factor of Safety, $J_u$ .....	1.5	1.5	1.5	1.5

- (1) Over entire vertical tail.
- (2)  $q_p$  is the dynamic pressure corresponding to  $V_s$  (see § 4a.42).
- (3) See § 4a.120 (a) for exception.
- (4) See § 4a.123 (b) for exception.
- (5) See § 4a.122 (c).
- (6) Refers to main surface, disregarding tab; uniform pressure distribution may be assumed over tab

TABLE 4a-4. Loading Conditions for Vertical Tail Surfaces.

1. Condition.....	Maneuvering	Tab effects
2. Reference.....	§ 4a.125	§ 4a.126
3. Design Speed (see § 4a.73).....	$V_s^{(1)}$	$V_L^{(1)}$
4. $C_N$ or Gust.....	$C_N=0.45$	
5. Average Limit Pressure, p. s. f. (1).....	$C_N q_p^{(2)}$	
6. Chord Distribution.....	Fig. 4a-7	Fig. 4a-7 <sup>(4)</sup>
7. Span Distribution.....	Constant $C_N$	Constant $C_N^{(5)}$
8. Minimum Average Limit Pressure, p. s. f. (1).....	12	
9. Special Requirements.....	§ 4a.125(b)	None
10. Minimum Yield Factor of Safety, $J_y$ .....	1.0	1.0
11. Minimum Ultimate Factor of Safety, $J_u$ .....	1.5	1.5

- (1) See § 4a.125 (a) for exception.
- (2)  $q_p$  is the dynamic pressure corresponding to  $V_s$  (see § 4a.42).
- (3)  $V_L$  is the maximum level flight air speed with any engine inoperative.
- (4) Refers to main surface, disregarding tab; uniform pressure distribution may be assumed over tab.

TABLE 4a-5. Loading Conditions for Ailerons.

(See § 4a.137)

	Elevator	Rudder		Aileron	Flaps, tabs, etc.
		Symmetrical thrust (1)	Unsymmetrical thrust (2)		
1. Reference.....	.139	.140	140	.141	.142
2. Maximum Limit Control Force, pounds.....	200	200	200	80	None
3. Minimum Limit Control Force, pounds.....	Fig. 4a-8	130	200	Fig. 4a-9	See Ref.
4. Minimum Yield Factor of Safety, $J_y$ .....	1.0	1.0	1.0	1.0	1.0
5. Minimum Ultimate Factor of Safety, $J_u$ .....	1.5	1.5	1.5	1.5	1.5

- (1) Propeller axes all in plane of symmetry.
- (2) Propeller axes not all in plane of symmetry.

TABLE 4a-6. Loading Conditions for Control Systems.

(See §§ 4a.207-4a.216)

Item	Component	Reference	Additional yield factor of safety, $j_y$	Additional ultimate factor of safety, $j_u$	May be covered by Item No.
1.	Fittings (except control system fittings)	§ 4a.208	None	1.20	2, 4, 5, 6, 7, 8, 9
2.	Castings	§ 4a.209	None	2.00	7, 8
3.	Parallel double wires in wing lift truss	§ 4a.210	None	1.05	1
4.	Wires at small angles	§ 4a.211	None	See Ref.	
5.	Double drag truss wires	§ 4a.212	None	See Ref.	
6.	Torque tubes used as hinges	§ 4a.213	None	1.5	
7.	Control surface hinges (1)	§ 4a.214	None	6.67	
8.	Control system joints (1)	§ 4a.214	None	3.33	
9.	Wire sizes	§ 4a.215	None	See Ref.	
10.	Wing lift truss (landing conditions only)	§ 4a.216	None	1.10	

(1) For bearing stresses only.

TABLE 4a-7. Additional (Multiplying) Factors of Safety.

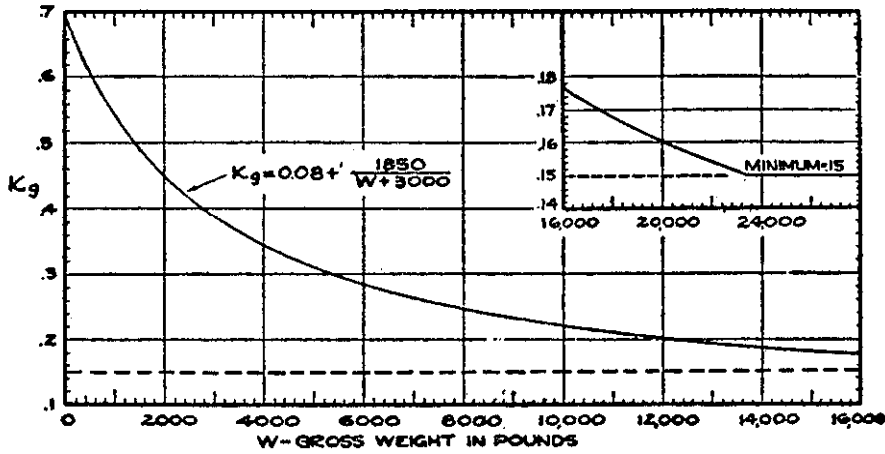


FIGURE 4a-1. Gliding Speed Factor.

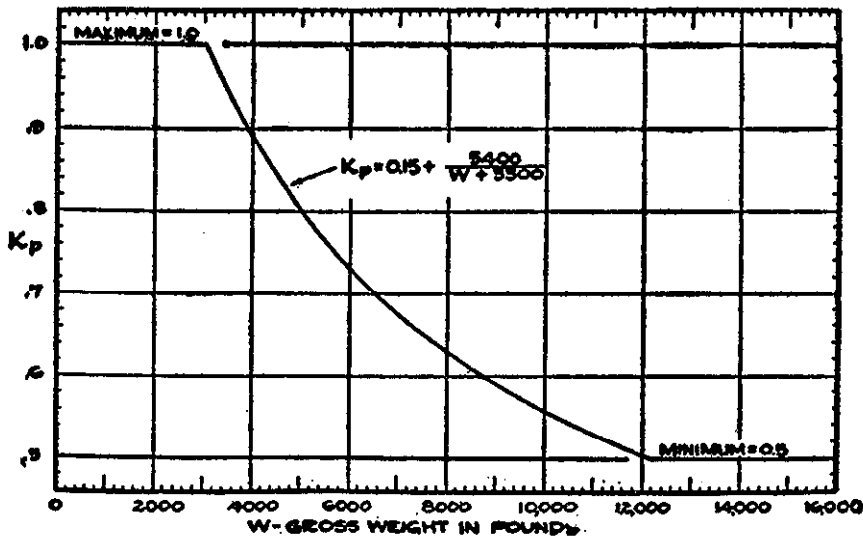


FIGURE 4a-2. Pull-Up Speed Factor.

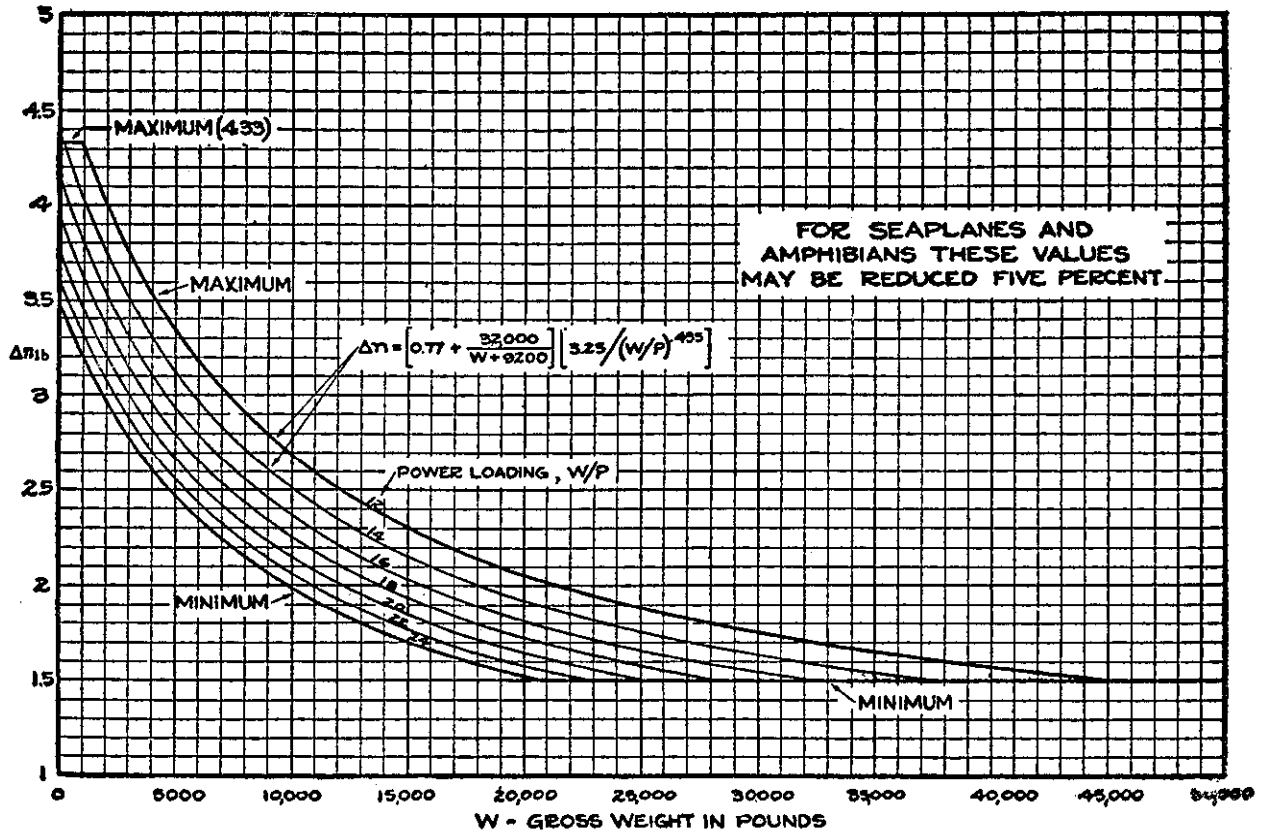


FIGURE 4a-3. Maneuvering Load Factor Increment, Conditions I and III.

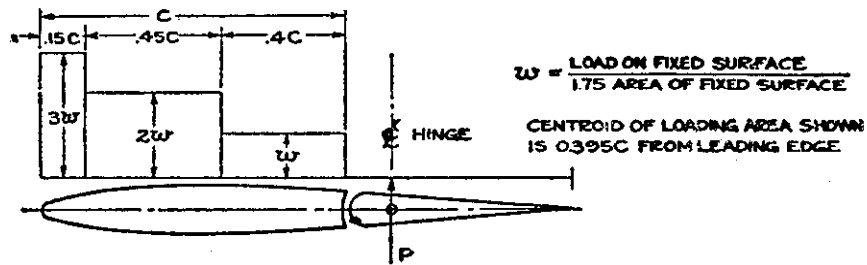


FIGURE 4a-4. "Balancing" Distribution—Horizontal Tail.

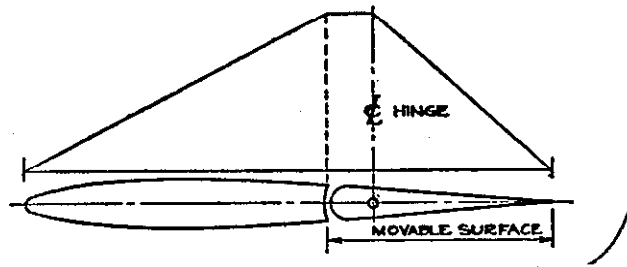


FIGURE 4a-5. "Maneuvering" Tail Load Distribution.

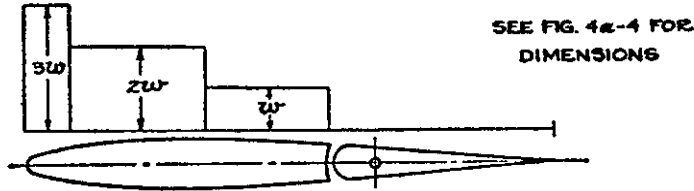


FIGURE 4a-6. "Damping" Tail Load Distribution.

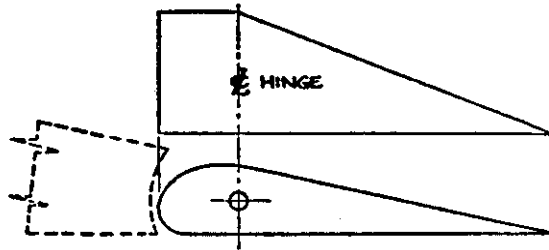


FIGURE 4a-7. Aileron Load Distribution.

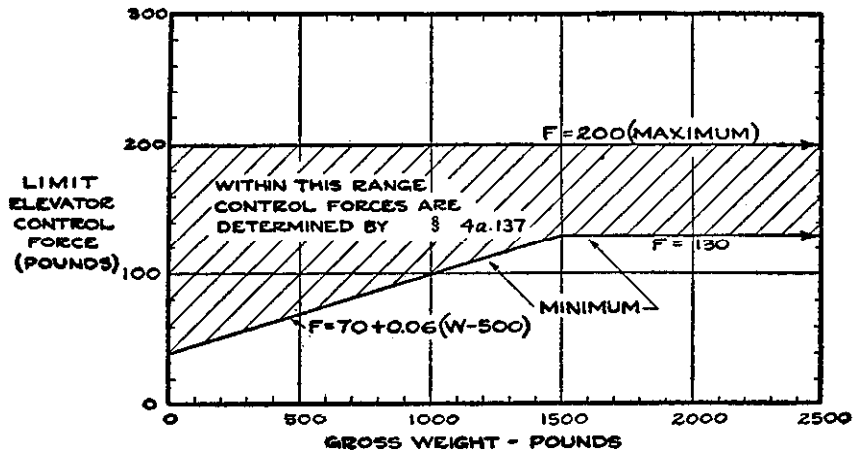


FIGURE 4a-8. Elevator Control Force Limits.

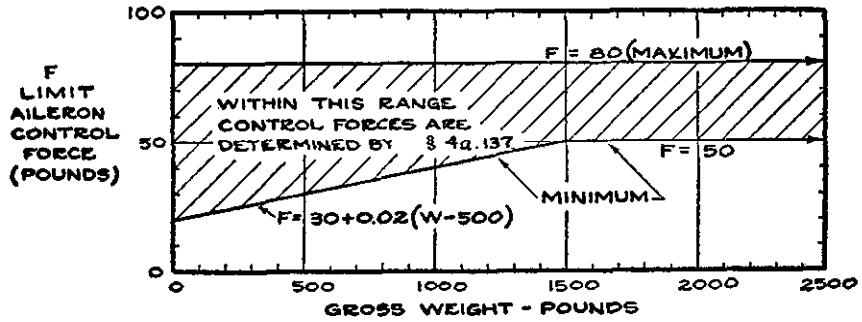
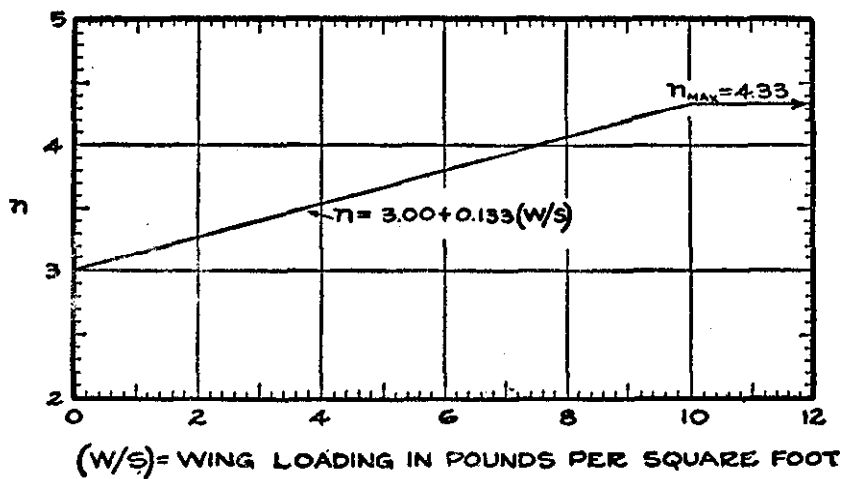
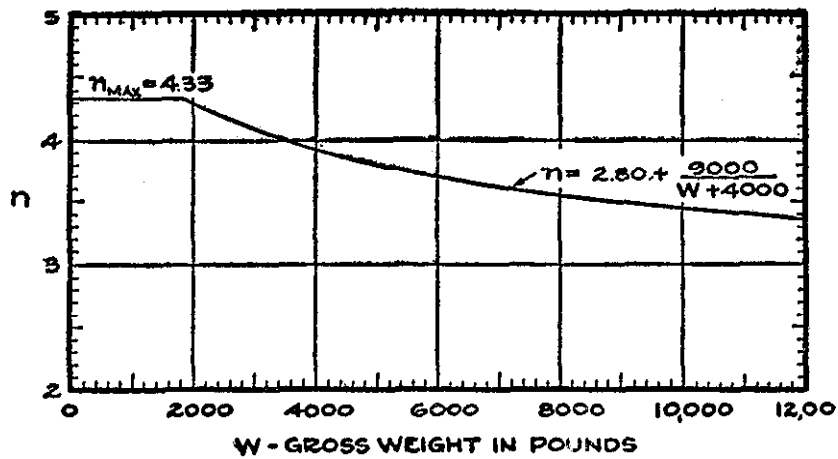


FIGURE 4a-9. Aileron Control Force Limits.



NOTE: USE THE CHART INDICATING THE LOWER VALUE

FIGURE 4a-10. Limit Load Factors for Level and 3-Point Landing Conditions.

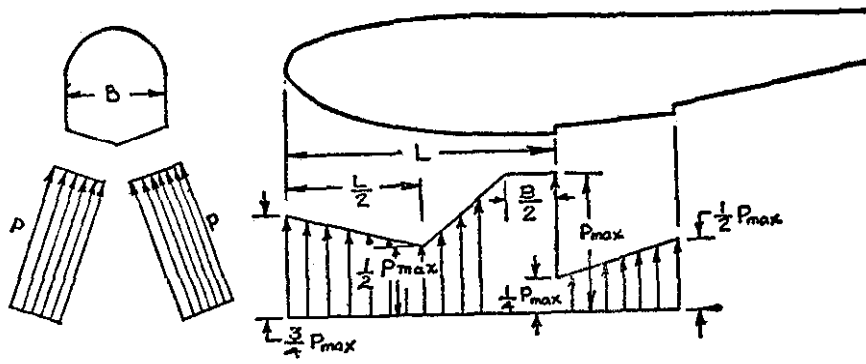


FIGURE 4a-11. Distribution of Local Pressures—Boat Seaplanes.

# **Appendix B**

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**Special Civil Air Regulations**  
Which affect Part 4a



SPECIAL CIVIL AIR REGULATION NO. SR-399B

Effective: May 28, 1959\*  
Issued: May 21, 1959

Provisional Maximum Certificated Weights for Certain Airplanes Operated by Alaskan Air Carriers, Alaskan Air Taxi Operators, and the Department of the Interior

Special Civil Air Regulations Nos. SR-399 and SR-399A, (18 F.R. 6799 and 20 F.R. 8091) authorized the Administrator to establish increased maximum authorized weights for certain airplanes of 12,500 pounds or less operated entirely within Alaska by Alaskan air carriers as designated by Part 292 of the Board's Economic Regulations or by the United States Department of the Interior. Effective February 3, 1959, the Civil Aeronautics Board amended Part 292 to eliminate Alaskan pilot-owners from the provisions of that Part (24 F.R. 437) and concurrently adopted new Part 293 (24 F.R. 127) redesignating such air carriers as Alaskan air taxi operators. Accordingly, in order to permit this new class of air carriers to continue operating airplanes under the increased maximum weights authorized by SR-399A, such regulations are revised to specifically include Alaskan air taxi operators. However, Alaskan air taxi operators will remain subject to the 7,900 pound weight limitation imposed by Part 293.

This special regulation does not impose any additional burden upon any person and is purely technical in nature. For these reasons, the Administrator finds that compliance with the notice, public participation and effective date provisions of section 4 of the Administrative Procedure Act is unnecessary. In consideration of the foregoing, I hereby adopt a Special Civil Air Regulation, effective immediately on the date of its publication in the Federal Register, to read as follows:

1. Notwithstanding any contrary provisions of the Civil Air Regulations, the Director, Bureau of Flight Standards and any employee of such administrative unit as he shall designate may increase the maximum certificated weight for airplanes which are:

(a) Operated entirely within the State of Alaska by an Alaskan air carrier or an Alaskan air taxi operator pursuant to Parts 292 and 293, respectively, of the Civil Aeronautics Board's Economic Regulations, or by the United States Department of the Interior in the conduct of its game and fish law enforcement activities and its management, fire detection, and fire suppression activities concerning public lands; and

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\*Date of publication in the Federal Register.

(b) Type certificated under the provisions of Aeronautical Bulletin No. 7 of the Aeronautics Branch of the United States Department of Commerce dated January 1, 1939, as amended, or under the normal category of Part 4a of the Civil Air Regulations.

2. The maximum certificated weight herein referred to shall not exceed any of the following:

(a) 12,500 pounds,

(b) 115 percent of the maximum weight listed in the FAA Aircraft Specification,

(c) The weight at which the airplane meets the positive maneuvering load factor requirement for the normal category specified in sec. 3.186 of the Civil Air Regulations, or

(d) The weight at which the airplane meets the climb performance requirements under which it was type certificated.

3. In determining the maximum certificated weight the structural soundness of the airplane and the terrain to be traversed in the operation will be considered.

4. The maximum certificated weight so determined will be added to the airplane's operation limitations and identified as the maximum weight authorized for operations within the State of Alaska.

This regulation supersedes Special Civil Air Regulation No. SR-399A, and shall terminate on October 25, 1960, unless sooner superseded or rescinded.

SPECIAL CIVIL AIR REGULATION NO. SR-407

Effective: September 10, 1954

Adopted: August 6, 1954

Basis for Approval of Modification of Airplane Types Douglas DC-3 and Lockheed L-18

On October 7, 1953, the Civil Aeronautics Board adopted Special Civil Air Regulation SR-398, effective November 11, 1953, (18 F.R. 6448) which provided a basis for approval of modifications of DC-3 and L-18 airplanes, but limited the scope of that regulation to increases in engine takeoff power limitation up to 1,350 horsepower per engine, to installation of engines of not more than 1,830 cubic inches displacement, and to the establishment of new maximum certificated weights not in excess of 26,900 pounds for the DC-3 and 19,500 pounds for the L-18. This regulation expands the scope of SR-398 to cover approval of additional modifications and maximum certificated weights for DC-3 and L-18 airplanes. In order to have all the regulations applicable to modifications and weight increases of the DC-3 and L-18 appear in one document, this regulation embodies herein the provisions of SR-398 and the additional regulations which are made applicable to these airplanes. SR-398 is thus superseded by this regulation.

The Douglas DC-3 and the Lockheed L-18 airplane types were originally designed and certificated on the basis of airworthiness standards in effect prior to 1940. These were contained in Bulletin 7-A promulgated by the former Bureau of Air Commerce and in the initial Part 04 of the Civil Air Regulations which was basically a recodification of Bulletin 7-A. Subsequently the Civil Aeronautics Board promulgated and kept up to date newer airworthiness requirements contained in Parts 3, 4a, and 4b of the Civil Air Regulations, the latter two parts containing rules for transport category airplanes. The DC-3 and L-18 are the only large airplane types now in general use in the United States which retain the old standards as a basis for their certification.

For a number of years after the initial certification of these two airplane types relatively few important design changes were introduced, and the airplane specifications regarding maximum certificated weights remained practically unchanged. More recently, however, several operators have made significant design changes in DC-3 and L-18 airplanes and it has become apparent that with the continued use of these airplanes, more operators are considering design changes, such as the installation of higher powered engines and increases in maximum certificated weights. Although the basis for certification of these airplanes remains unchanged, the Administrator of Civil Aeronautics in some instances has made changes to the pertinent aircraft specifications by applying certain provisions of Part 4a of the Civil Air Regulations. The Board does not consider Bulletin 7-A and the early versions of Part 04 adequate regulatory bases upon which further modifications to these airplane types can be approved by the Administrator.

The provisions of this Special Civil Air Regulation provide the basis for approval by the Administrator of future modifications of individual DC-3 and L-18 airplanes. The provisions in section 1 of SR-398 permit-

ting the Administrator to waive the requirements of the regulation when it was shown that a particular airplane was in the process of modification at the time that regulation became effective, has been omitted from this regulation because SR-398 has been in effect for sufficient time to eliminate the need for waiver authority by the Administrator.

This regulation continues the general provisions of SR-398 which required that modifications of DC-3 and L-18 airplanes be accomplished in accordance with the provisions of either Part 4a or Part 4b of the Civil Air Regulations applicable to the modification being made, and which were in effect on September 1, 1953, unless the applicant elects to make the modification in accordance with Part 4b as in effect on the date of modification. This regulation requires, also, that in electing to perform a modification under either Part 4a or Part 4b, each specific modification must be accomplished in accordance with all of the provisions of either Part 4a or Part 4b related to the particular modification. It does not permit selection of certain provisions of Part 4a and other provisions of Part 4b. For example, if it were desired to make a modification of the landing gear, it could be made under either Part 4a or Part 4b, but not partially under Part 4a and partially under Part 4b. This also applies when the applicant elects to make a modification in accordance with the provisions of Part 4b in effect on the date of modification in lieu of Part 4a or Part 4b as in effect on September 1, 1953.

In addition to the general provisions for modification, this regulation contains the specific requirements of SR-398 with respect to approval of increases in takeoff power limitation and the installation of new type engines in DC-3 and L-18 airplanes and provides some additional requirements. The intent of these specific requirements is to ensure that such changes will not result in a decrease in safety. This is in general consistent with the policy followed by the Administrator in approving changes made prior to the adoption of SR-398.

In the case of an increase in the takeoff power limitation beyond 1,200 horsepower per engine, but not to exceed 1,350 horsepower per engine, this regulation continues the provisions of SR-398 which required that the increase in power shall not adversely affect the flight characteristics of the airplane. The intent of this provision is to permit increases in takeoff power only if the applicant can show that the use of the increased power does not result in deterioration of the flight characteristics of the airplane when compared to its previous characteristics. It is believed, however, that increases in takeoff power limitation above 1,350 horsepower per engine may result in changes in the engine installation and the airplane's flight characteristics to such an extent that the basis for approval of increases in horsepower provided in SR-398 is no longer appropriate. Since Part 4b represents the most recent design practices, this regulation provides that the takeoff power limitation may be increased beyond 1,350 horsepower if compliance is shown with Part 4b at the increased power with respect to the engine installation provisions, the flight characteristics, and the ground handling requirements.

This regulation contains the provisions of SR-398 which established either Part 4a or Part 4b as the basis for approving the installation of engines not exceeding 1,830 cubic inches displacement which necessitate major modification or redesign, on the condition that there is no de-

crease in engine fire prevention and protection when compared to the prior engine installation. Since the use of engines with more than 1,830 cubic inches displacement will necessitate extensive modification of the entire engine installation, it appears necessary that such an installation be accomplished in accordance with the most recent airworthiness provisions of the regulations. This regulation, therefore, requires that the installation of these larger engines be accomplished in accordance with the requirements of Part 4b of the Civil Air Regulations.

This regulation also continues the provisions of SR-398 which permitted establishment of new maximum certificated weights not to exceed 26,900 pounds for the DC-3 and 19,500 pounds for the L-18. Where such new maximum weights are desired, the airworthiness certificates may be amended and the maximum weights established in accordance with the transport category performance requirements of either Part 4a or Part 4b, subject to the structural limitations of the airplane. With respect to maximum weights in excess of 26,900 pounds for the DC-3 and 19,500 pounds for the L-18, however, it is considered that such increases in weight may seriously affect not only the structural limitations, but also the flight and ground handling characteristics of the airplane. This regulation requires that weights in excess of 26,900 pounds for the DC-3 and 19,500 pounds for the L-18 shall be established in accordance with the performance, structural, flight characteristics, and ground handling requirements of Part 4b.

It should be noted that in certain cases, showing of compliance is required on the basis of Part 4b of the Civil Air Regulations. In some instances, literal compliance with the provisions of Part 4b may be extremely difficult to accomplish and would not contribute materially to the objective sought. In such cases this regulation provides that the Administrator may take into account the experience gained with the DC-3 and L-18 airplanes. Where such experience justifies it, he is authorized to accept in lieu of the literal provisions of Part 4b such measures of compliance as he finds will effectively accomplish the basic objectives.

SR-398 also required, as a basis for approval of new maximum certificated weights, that the applicant provide flight manual material containing information which will permit the application of the transport category performance operating limitations in the operation of the airplane. This Special Civil Air Regulation continues this requirement. In view of the fact that an applicant for new maximum certificated weights has such weights established in accordance with the transport category performance requirements of either Part 4a or Part 4b, this regulation also provides that an airplane having such newly established weights shall be considered a transport category airplane in applying the operating rules of the Civil Air Regulations. As a result, therefore, such airplanes when used in air carrier passenger service must be operated in accordance with the transport category performance operating limitations, but when used in other types of services the transport category performance operating limitations are not applicable as at present. Consideration is being given, however, to the problem of making them applicable to all types of operations conducted in transport category airplanes. If such rules are established, then all DC-3 and L-18 airplanes, having

certificated weights increased in accordance with SR-398 or this regulation, will have to comply with these rules also. In the meantime, it is recommended that operators, whose airplanes are approved for increased weights by reason of SR-398 or by this regulation, make use of this information to aid in assuring safety in their operations.

Interested persons have been afforded an opportunity to participate in the making of this regulation, and due consideration has been given to all relevant matter presented.

In consideration of the foregoing, the Civil Aeronautics Board hereby makes and promulgates the following Special Civil Air Regulation effective September 10, 1954:

1. *Applicability.* Contrary provisions of the Civil Air Regulations regarding certification notwithstanding,<sup>1</sup> this regulation shall provide the basis for approval by the Administrator of modifications of individual Douglas DC-3 and Lockheed L-18 airplanes subsequent to the effective date of this regulation.

2. *General Modifications.* Except as modified in sections 3 and 4 of this regulation, an applicant for approval of modifications to a DC-3 or L-18 airplane which result in changes in design or in changes to approved limitations shall show that the modifications were accomplished in accordance with the rules of either Part 4a or Part 4b in effect on September 1, 1953, which are applicable to the modification being made: *Provided*, That an applicant may elect to accomplish a modification in accordance with the rules of Part 4b in effect on the date of application for the modification in lieu of Part 4a or Part 4b as in effect on September 1, 1953: *And provided further*, That each specific modification must be accomplished in accordance with all of the provisions contained in the elected rules relating to the particular modification.

3. *Specific conditions for approval.* An applicant for any approval of the following specific changes shall comply with section 2 of this regulation as modified by the applicable provisions of this section:

(a) *Increase in takeoff power limitation—1,200 to 1,350 horsepower.* The engine take-off power limitation for the airplane may be increased to more than 1,200 horsepower but not to more than 1,350 horsepower per engine if the increase in power does not adversely affect the flight characteristics of the airplane.

(b) *Increase in takeoff power limitation to more than 1,350 horsepower.* The engine takeoff power limitation for the airplane may be increased to more than 1,350 horsepower per engine if compliance is shown with the flight characteristics and ground handling requirements of Part 4b.

(c) *Installation of engines of not more than 1,830 cubic inches displacement and not having a certificated take-off rating of more than 1,350 horsepower.* Engines of not more than 1,830 cubic inches displacement and not having a certificated takeoff rating of more than 1,350 horsepower which necessitate a major modification or redesign of the engine installation may be installed, if the engine fire prevention and fire protection are equivalent to that on the prior engine installation.

<sup>1</sup> It is not intended to waive compliance with such airworthiness requirements as are included in the operating parts of the Civil Air Regulations for specific types of operation.

(d) *Installation of engines of more than 1,830 cubic inches displacement or having certificated takeoff rating of more than 1,350 horsepower.* Engines of more than 1,830 cubic inches displacement or having certificated takeoff rating of more than 1,350 horsepower may be installed if compliance is shown with the engine installation requirements of Part 4b: *Provided*, That where literal compliance with the engine installation requirements of Part 4b is extremely difficult to accomplish and would not contribute materially to the objective sought, and the Administrator finds that the experience with the DC-3 or L-18 airplanes justifies it, he is authorized to accept such measures of compliance as he finds will effectively accomplish the basic objective.

4. *Establishment of new maximum certificated weights.* An applicant for approval of new maximum certificated weights shall apply for an amendment of the airworthiness certificate of the airplane and shall show that the weights sought have been established, and the appropriate manual material obtained, as provided in this section. (Note: Transport category performance requirements result in the establishment of maximum certificated weights for various altitudes.)

(a) *Weights—25,200 to 26,900 for the DC-3 and 18,500 to 19,500 for the L-18.* New maximum certificated weights of more than 25,200 but not more than 26,900 pounds for DC-3 and more than 18,500 but not more than 19,500 pounds for L-18 airplanes may be established in accordance with the transport category performance requirements of either Part 4a or Part 4b, if the airplane at the new maximum weights can meet the structural requirements of the elected part.

(b) *Weights of more than 26,900 for the DC-3 and 19,500 for the L-18.* New maximum certificated weights of more than 26,900 pounds for DC-3 and 19,500 pounds for L-18 airplanes shall be established in accordance with the structural, performance, flight characteristics, and ground handling requirements of Part 4b: *Provided*, That where literal compliance with the structural requirements of Part 4b is extremely difficult to accomplish and would not contribute materially to the objective sought, and the Administrator finds that the experience with the DC-3 or L-18 airplanes justifies it, he is authorized to accept such measures of compliance as he finds will effectively accomplish the basic objective.

(c) *Airplane flight manual—performance operating information.* An approved airplane flight manual shall be provided for each DC-3 and L-18 airplane which has had new maximum certificated weights established under this section. The airplane flight manual shall contain the applicable performance information prescribed in that part of the regulations under which the new certificated weights were established and such additional information as may be necessary to enable the application of the takeoff, en route, and landing limitations prescribed for transport category airplanes in the operating parts of the Civil Air Regulations.

(d) *Performance operating limitations.* Each airplane for which new maximum certificated weights are established in accordance with paragraphs (a) or (b) of this section shall be considered a transport category airplane for the purpose of complying with the performance operating limitations applicable to the operations in which it is utilized.

**5. Reference.** Unless otherwise provided, all references in this regulation to Part 4a and Part 4b are those parts of the Civil Air Regulations in effect on September 1, 1953.

*Note:* Parts 4a and 4b as amended and in effect on September 1, 1953, were published in the Federal Register at the following citations:

*Part 4a*, 14 F.R. 4072, 14 F.R. 3742, 14 F.R. 6769, 15 F.R. 28, 17 F.R. 11631.

*Part 4b*, 15 F.R. 3543, 15 F.R. 8903, 15 F.R. 9184, 16 F.R. 314, 16 F.R. 11759, 16 F.R. 12220, 17 F.R. 1087, 17 F.R. 11631, 18 F.R. 2213.

This regulation supersedes Special Civil Air Regulation SR-398 and shall remain effective until superseded or rescinded by the Board.



SPECIAL CIVIL AIR REGULATION NO. SR-411A

Effective: July 1, 1957

Adopted: June 28, 1957

Trial Operation of Transport Category Airplanes in Cargo Service at Increased Zero Fuel and Landing Weights

On June 30, 1955, the Civil Aeronautics Board adopted Special Civil Air Regulation No. SR-411 authorizing for a period of 2 years' trial operations of transport category airplanes in cargo service at weights in excess of those permitted in passenger service. The weights involved are the zero fuel weight (maximum weight of the airplane with no disposable fuel and oil) and the structural landing weight. The use of the higher weights is made contingent upon certain findings by the Administrator and upon certain conditions of operation and inspection.

Prior to the adoption of SR-411, trial operations of Douglas DC-6A airplanes at higher weights were authorized in orders issued by the Board to individual carriers. The first of such orders was issued on July 21, 1954. Authorization of the trial operations was predicated on the premise that such operations could eventually lead to the establishment of a sound basis for differentiating between standards for passenger and cargo air carrier operations. The success of the trial operations under the Board's orders led to the promulgation of SR-411 which permits any number of any type of transport category airplane to be operated at increased weights in cargo service. The basic intent of SR-411 was to obtain a more extensive background of operating experience to assure that the conditions governing the trial operations do in fact provide a sound basis for establishing possible future standards for airplanes in cargo operations.

Prior to the effective date of SR-411, there were a total of 6 DC-6A airplanes being used by 3 operators under waivers at higher weights. As of December 30, 1956, such operations encompassed 23 DC-6A airplanes and 5 operators. More recently Lockheed L1049-H airplanes have been introduced in cargo service at increased weights. Results of inspections submitted thus far to the Board have revealed no serious structural defects which could be attributed to operation at the increased weights.

In the preamble to SR-411, the Board indicated that it expected to have interested persons submit their evaluation of the trial operations and recommendations with respect to future regulatory action. In this regard, industry segments directly related to the trial operations under SR-411 recommended that the authorization for trial operations be continued on the basis that such operations have not resulted in any indications of structural distress and on the basis of the economic importance of such trial operations to the promotion and success of cargo service. Based on the foregoing and on the evaluation of the inspection and operational reports submitted by the air carriers, the Board finds that there is sufficient justification to permit the continuation of trial operations beyond June 30, 1957, the termination date of Special Civil Air Regulation No. SR-411. From the operational data submitted by the operators, the scope of operations under SR-411 has been such that substantiation of the conditions for these operations for inclusion in the regulations

on a permanent basis will entail a long range program. Consequently, the termination date of this regulation is being established to permit trial operations for an additional five-year period.

In connection with the rulemaking procedures effected on this matter, it was suggested that the Board extend the trial operations for a period no longer than one or two years on the premise that the shorter time extension would permit closer control of these operations. The Board fails to find any relationship between the extension period and the quality of control exerted over the trial operations inasmuch as the Board intends to continue its evaluation of the inspection reports submitted by the operators at regular intervals and envisions that the industry and the Administrator will do likewise.

Suggestions were also received to the effect that the trial operations should not be limited only to the older type airplanes, as proposed in the notice of proposed rulemaking; i.e., to airplanes certificated prior to March 13, 1956. In this regard, the Board considers that in view of the new concept of structural design requirements as well as other related changes in these requirements which were introduced on the aforementioned date into Part 4b of the Civil Air Regulations along with the new concept of structural design, it is advisable that some experience be gained with airplanes certificated under those provisions at the normal transport category weights before permitting such airplanes to operate at the higher weight. Accordingly, this special regulation extends the authorization to operate at higher weights only to airplanes certificated in accordance with the transport category requirements in effect prior to March 13, 1956.

It should be noted that, as indicated in the preamble to SR-411, the Board envisions that most of the airplanes taking advantage of this regulation will continue to be used in cargo service solely. However, if any of the airplanes operated under the provisions of this regulation are to be made eligible at any later time for passenger service, the provisions of this regulation require accomplishment of a special inspection and an evaluation of the fatigue limitations. It is suggested that, if it is contemplated to return the airplane to passenger service, the establishment of special inspections and the evaluation of the fatigue limitations be accomplished prior to increasing the airplane weight for cargo service. It is not the intent of this regulation to permit intermittent operations in passenger and cargo service.

The provisions of this regulation differ from the provisions of Special Civil Air Regulation No. SR-411 in that they are made applicable only to airplanes certificated in accordance with the transport category structural requirements effective prior to March 13, 1956; they require inspection and flight data records to be kept available for no longer than one year; and they require flight records (as distinguished from inspection records) to be made only with respect to those flights involving increased weights, rather than for all flights regardless of whether or not they are made at the increased weights.

Interested persons have been afforded an opportunity to participate in the making of this regulation (22 F.R. 3416), and due consideration has been given to all relevant matter presented. Since this regulation

imposes no additional burden on any persons, it may be made effective on less than 30 days' notice.

In consideration of the foregoing, the Civil Aeronautics Board hereby makes and promulgates the following Special Civil Air Regulation, effective July 1, 1957.

Notwithstanding the applicable structural provisions of the Civil Air Regulations, any air carrier may operate in cargo service transport category airplanes which were certificated under the provisions of Part 4a or Part 4b, effective prior to March 13, 1956, subject to the conditions hereinafter set forth:

(1) The air carrier shall submit an application to the Administrator indicating its desire to operate its airplane(s) under the provisions of this special regulation, and indicating which airplane(s) would be involved.

(2) The air carrier shall furnish a statement from each manufacturer for each type of airplane involved indicating in each case that the airplane manufacturer approves the operation of such type of airplane under the provisions of this regulation and that the airplane manufacturer will establish the inspection procedure prescribed in paragraph (4) of this Special Civil Air Regulation and will supervise such inspection as necessary.

(3) The zero fuel weight (maximum weight of the airplane with no disposable fuel and oil) and the structural landing weight may be increased beyond the maxima approved in full compliance with the applicable Civil Air Regulations: *Provided*, That the Administrator of Civil Aeronautics finds that the increase in either such weight is not likely to reduce seriously the structural strength, that the probability of sudden fatigue failure is not noticeably increased, and that the flutter, deformation, and vibration characteristics do not fall below those required by the Civil Air Regulations: *And provided further*, That any increase in the zero fuel weight approved shall not exceed 5 percent and that the increase in the structural landing weight shall not exceed the amount, in pounds, of the increase in the zero fuel weight.

(4) Airplanes for which the increased weights become effective shall be subject to inspections in addition to those normally performed, such inspections to be established by the manufacturer of the type airplane concerned, subject to the approval of the Administrator of Civil Aeronautics and to be supervised as found necessary by that manufacturer, to safeguard against possible structural distress resulting from the higher operating stress levels. The air carrier shall keep for a period of at least one year, and make available upon request to the Civil Aeronautics Board, the Administrator of Civil Aeronautics, or the manufacturer of the type of airplane concerned, the records of such inspections.

(5) Airplanes for which the increased weights become effective shall be operated in accordance with the transport category performance operating limitations prescribed in Part 40, 41, or 42 of the Civil Air Regulations, whichever is applicable.

(6) The air carrier shall keep for a period of at least one year and make available upon request to the Civil Aeronautics Board, the

Administrator of Civil Aeronautics, or the manufacturer of the type airplane concerned, records of all flights conducted at increased weights with airplanes for which the increased weights become effective, such records to include the actual takeoff, zero fuel, and landing weights.

(7) The Airplane Flight Manual of each airplane operating under the provisions of this special regulation shall be appropriately revised so as to include the necessary operating limitations and operating information.

(8) An airplane which has been operated at increased weights under the provisions of this regulation shall not be used for the carriage of passengers, except under the following conditions:

(a) Special inspections established by the manufacturer and approved by the Administrator of Civil Aeronautics shall have been accomplished.

(b) The effects of the operations at increased weights on structural fatigue shall have been evaluated by the airplane manufacturer and taken into account in any fatigue limitations established for the airplane.

This regulation shall terminate on June 30, 1962, unless sooner superseded or rescinded by the Board.

SPECIAL CIVIL AIR REGULATION NO. SR-426

Effective: October 27, 1958

Adopted: September 22, 1958

Performance Credit for Transport Category Airplanes Equipped with Standby Power

Standby power is power and/or thrust obtained from rocket engines and is separate from the power obtained from the airplane's main engines. Such power and/or thrust is available for a relatively short period for use in cases of emergency. The standby power system may be capable of producing more than a single thrust period. This special regulation authorizes the Administrator to grant performance credit to transport category airplanes when standby power is used in one or more of the following regimes of flight: Takeoff with one engine inoperative, approach climb with one engine inoperative, and the balked landing climb. By "performance credit" is meant the taking into account the increased performance of the airplane with standby power and, because thereof, the approval of higher maximum weights for the airplane than the weights approved on the basis of the airplane's performance without standby power. In granting performance credit, this regulation prescribes the applicable conditions and limitations.

Rocket assist takeoff units were developed initially for the military services to provide additional takeoff power for heavily loaded flying boats and carrier-based aircraft. The additional thrust improved the climb performance in the takeoff regime and permitted the airplane to attain a safe altitude and air speed in the event of engine failure. The reliability of such units has increased to the point where some civil operators have adopted them for use as emergency standby power in the event of engine failure. Other operators have been reluctant to adopt standby power installations and the attendant weight penalties without reasonable performance credit being given the airplane. With the performance credit granted by this regulation for the use of standby power, it is anticipated that such credit might be considered a compensating economic factor justifying the installation of standby power on such airplanes.

The currently effective regulations did not contemplate the use of standby power; however, the Administrator of Civil Aeronautics has established for an interim period a conservative policy permitting transport category airplanes equipped with standby power to operate at an increase in the normally approved weights by an amount equal to the weight of the standby power units.

In this regulation, the criteria for granting performance credit were formulated with the intent of providing an overall level of performance equivalent to that intended by the currently effective regulations. To this end, appropriate criteria are established with respect to the amount of performance credit which may be applied in determining the new takeoff paths, the extent to which the maximum certificated takeoff and landing weights may be increased, and operating procedures to be followed in service for the use of standby power and for the associated changes in the airplane's configuration and speed.

The basic element of the various criteria established in this regulation entails a comparison of flight paths. In Civil Air Regulations

Draft Release No. 57-28, all of the flight paths were based on procedures involving attainment of the en route configuration and the acceleration to a safe en route speed. Further consideration of this proposal indicates that an undue amount of flight testing and computation would be required for those flight conditions which presently do not involve the establishment of such flight paths. In addition, it appears that comparison of flight paths involving a particular procedure is not essential for the purpose of establishing an equivalent level of safety. In view of the foregoing, this regulation requires comparison of flight paths with respect to the takeoff regime consistent only with that required by the currently effective airworthiness performance requirements; i.e., Part 4b and Special Civil Air Regulations Nos. SR-422 and SR-422A. Further, the establishment of maximum landing weights is based on a comparison of flight paths obtained with a fixed airplane configuration and at the speed and power condition appropriate to the all-engines-operating landing climb or the one-engine-inoperative approach climb, as applicable.

Performance credit for standby power with respect to the takeoff path is limited by the specification that the all-engines-operating takeoff path reflect a conservatively greater margin of overall performance than the one-engine-inoperative takeoff path with standby power in use. It is intended that this margin exist throughout the takeoff path prescribed by the applicable performance requirements. This regulation provides for the establishment of the margin by the Administrator; however, in no case can this margin be less than 15 percent. In view of the many different types of airplanes to which this regulation is applicable, a higher margin might be necessary in certain cases to insure safe day-to-day operations.

For reciprocating-engine-powered airplanes, the provisions of this regulation require that the applicant establish a procedure for the use of standby power for attaining the en route configuration and a safe en route speed in the event of an engine failure during the takeoff. Provisions for such procedures are contained in the regulations for turbine-powered airplanes and will therefore be applicable to such airplanes. The establishment of the procedures made applicable to reciprocating-engine-powered airplanes is left to the discretion of the applicant to select the altitude at which the acceleration is to take place, the basic limitation being that the slope along all points of the airborne portion of the takeoff path must be positive. This will provide for the critical operation of cleanup and acceleration during the takeoff which are not covered by the performance requirements of Part 4b and which have been cited in the past in connection with several accidents. In addition, with respect to the clearing of obstacles, the takeoff path scheduled in the Airplane Flight Manual for airplanes for which the performance requirements of Part 4b are applicable must be appropriately modified to reflect the effect of the aforementioned procedure.

The criteria for the establishment of maximum certificated takeoff weights involve two separate comparisons of takeoff paths. These comparisons must be carried out to a height of 400 feet above the takeoff surface. This minimum height is intended to be associated with a procedure where standby power is actuated within the neighborhood of

the critical engine failure speed  $V_1$ . These provisions are intended to limit the increase in maximum takeoff weight consistent with the overall level of performance currently attained with respect to the one-engine-inoperative takeoff and to limit the amount of standby power for which performance credit can be granted to insure a reasonable margin of performance for the all-engine day-to-day operations.

The provisions for the establishment of maximum certificated landing weights require only a comparison between two flight paths based on a steady climb and fixed configuration. One represents a climb path at the maximum weight previously certificated without standby power and the other a climb path at the increased weight with standby power. In addition, there is a provision which requires the establishment of procedures for the execution of balked landings and missed approaches in conjunction with the use of standby power.

A question has been raised as to whether duplicate sets of standby power units are required to obtain performance credit for both the takeoff and the approach stages of flight. It is intended by this regulation to require duplicate sets of standby power for a flight for which the use of standby power is necessary to comply with both the maximum certificated takeoff and the maximum certificated approach weight limitations. Where the use of standby power is necessary for compliance with the maximum certificated weight limitations for only one regime of flight, i.e., takeoff or approach, one set of standby power units is required.

There are also included safety criteria for the installation and operation of the standby power system.

The Board presently has under consideration a special regulation which would be applicable to nontransport category airplanes equipped with standby power.

Interested persons have been afforded an opportunity to participate in the making of this regulation (22 F.R. 10464), and due consideration has been given to all relevant matter presented.

In consideration of the foregoing, the Civil Aeronautics Board hereby makes and promulgates the following Special Civil Air Regulation, effective October 27, 1958.

Contrary performance provisions of the Civil Air Regulations notwithstanding, the Administrator may grant performance credit for the use of standby power on transport category airplanes. Such credit shall be applicable only to the maximum certificated takeoff and landing weights, the takeoff distance, and the takeoff paths, and shall not exceed that found by the Administrator to result in an overall level of safety in the takeoff, approach, and landing regimes of flight equivalent to that prescribed in the regulations under which the airplane was originally certificated without standby power. (Note: Standby power is power and/or thrust obtained from rocket engines for a relatively short period and actuated only in cases of emergency.) The following provisions shall apply:

(1) *Takeoff; general.* The takeoff data prescribed in sections (2) and (3) shall be determined at all weights and altitudes, and at ambient temperatures if applicable, at which performance credit is to be applied.

(2) *Takeoff path.*

(a) The one-engine-inoperative takeoff path with standby power in use shall be determined in accordance with the performance requirements of the applicable airworthiness regulations.

(b) The one-engine-inoperative takeoff path (excluding that portion where the airplane is on or just above the takeoff surface) determined in accordance with paragraph (a) of this section shall lie above the one-engine-inoperative takeoff path without standby power at the maximum takeoff weight at which all of the applicable airworthiness requirements are met. For the purpose of this comparison, the flight path shall be considered to extend to at least a height of 400 feet above the takeoff surface.

(c) The takeoff path with all engines operating, but without the use of standby power, shall reflect a conservatively greater overall level of performance than the one-engine-inoperative takeoff path established in accordance with paragraph (a) of this section. The aforementioned margin shall be established by the Administrator to insure safe day-to-day operations, but in no case shall it be less than 15 percent. The all-engines-operating takeoff path shall be determined by a procedure consistent with that established in complying with paragraph (a) of this section.

(d) For reciprocating-engine-powered airplanes, the takeoff path to be scheduled in the Airplane Flight Manual shall represent the one-engine-inoperative takeoff path determined in accordance with paragraph (a) of this section and modified to reflect the procedure (see sec. (6)) established by the applicant for flap retraction and attainment of the en route speed. The scheduled takeoff path shall have a positive slope at all points of the airborne portion and at no point shall it lie above the takeoff path specified in paragraph (a) of this section.

(3) *Takeoff distance.* The takeoff distance shall be the horizontal distance along the one-engine-inoperative takeoff path determined in accordance with section (2)(a) from the start of the takeoff to the point where the airplane attains a height of 50 feet above the takeoff surface for reciprocating-engine-powered airplanes and a height of 35 feet above the takeoff surface for turbine-powered airplanes.

(4) *Maximum certificated takeoff weights.* The maximum certificated takeoff weights shall be determined at all altitudes, and at ambient temperatures if applicable, at which performance credit is to be applied and shall not exceed the weights established in compliance with paragraphs (a) and (b) of this section.

(a) The conditions of section (2) (b) through (d) shall be met at the maximum certificated takeoff weight.

(b) Without the use of standby power, the airplane shall meet all of the en route requirements of the applicable airworthiness regulations under which the airplane was originally certificated. In addition, turbine-powered airplanes without the use of standby power shall meet the final takeoff climb requirements prescribed in the applicable airworthiness regulations.

(5) *Maximum certificated landing weights.*

(a) The maximum certificated landing weights (one-engine-inoperative approach and all-engines-operating landing climb) shall be



determined at all altitudes, and at ambient temperatures if applicable, at which performance credit is to be applied and shall not exceed that established in compliance with the provisions of paragraph (b) of this section.

(b) The flight path, with the engines operating at the power and/or thrust appropriate to the airplane configuration and with standby power in use, shall lie above the flight path without standby power in use at the maximum weight at which all of the applicable airworthiness requirements are met. In addition, the flight paths shall comply with the provisions of subparagraphs (i) and (ii) of this paragraph.

(i) The flight paths shall be established without changing the appropriate airplane configuration.

(ii) The flight paths shall be carried out for a minimum height of 400 feet above the point where standby power is actuated.

(6) *Airplane configuration, speed, and power and/or thrust; general.* Any change in the airplane's configuration, speed, and power and/or thrust shall be made in accordance with the procedures established by the applicant for the operation of the airplane in service and shall comply with the provisions of paragraphs (a) through (c) of this section. In addition, procedures shall be established for the execution of balked landings and missed approaches.

(a) The Administrator shall find that the procedure can be consistently executed in service by crews of average skill.

(b) The procedure shall not involve methods or the use of devices which have not been proven to be safe and reliable.

(c) Allowances shall be made for such time delays in the execution of the procedures as may be reasonably expected to occur during service.

(7) *Installation and operation; standby power.* The standby power unit and its installation shall comply with the provisions of paragraphs (a) and (b) of this section.

(a) The standby power unit and its installation shall not adversely affect the safety of the airplane.

(b) The operation of the standby power unit and its control shall have proven to be safe and reliable.